

## **Revision 1 Transmittal**

December 29, 2015

**TO:**

Holders of Cirrus Design SR22T Pilot's Operating Handbook for Aircraft Serials SR22T-0442 and Subsequent with Teledyne Continental Motors Turbocharged Engine, 3600 Pound Takeoff Weight, P/N 13772-005.

**SUBJECT:**

Revision 1 dated 29 Dec 2015.

Revision 1 to the Model SR22T Pilot's Operating Handbook revises Sections 1, 2, 3, 3A, 4, 5, 6, 7, 8, 9, and 10.

Revise sections by inserting revised pages and removing superseded pages in accordance with the List of Effective Pages. After incorporating revision pages, discard superseded pages and this transmittal.

## Revision 1 Transmittal (Cont.)

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## **Revision 1 Highlights**

<b>Page</b>	<b>Revision Description</b>
Front Matter .....	Revised Front Matter.
Section 1 .....	Revised Introduction. Revised The Airplane section.
Section 2 .....	Incorporated TPOH 15-02: MD302. Incorporated TPOH 15-05: Baro-VNAV. Added door placard.
Section 3 .....	Revised Engine Partial Power Loss Checklist. Revised Starter Engaged Annunciation Checklist. Revised CAPS Deployment Checklist.
Section 3A .....	Incorporated TPOH 15-02: MD302. Revised Starter Engaged Annunciation Checklist.
Section 4 .....	Incorporated TPOH 13-04: Takeoff Airspeeds. Incorporated TPOH 15-25: Cruise Procedure. Revised Preflight Inspection Checklist. Revised Cold Weather Starting Checklist. Added Extended Ground Operation Checklist.
Section 5 .....	Revised Associated Conditions Affecting Performance section. Revised Cruise Performance section. Revised Landing Distance section. Added Landing Distance Table - Flaps 50%. Added Landing Distance Table - Flaps 0%.
Section 6 .....	Revised Introduction. Removed Airplane Weighing Form. Removed Airplane Weighing Procedures. Removed Airplane Leveling section.
Section 7 .....	Incorporated TPOH 14-03: Brakes. Incorporated TPOH 14-06: Electrical System. Incorporated TPOH 15-03: MD302. Incorporated TPOH 15-06: Baro-VNAV. Incorporated TPOH 15-09R1: USB-A Ports & Fire Extinguisher. Added Key fob to Cabin Doors section. Revised Magnetic Compass section. Revised Pitot-Static System illustration. Added GTX 33 ES Transponder. Added MY2016 Convenience Lighting option.
Section 8 .....	Incorporated TPOH 14-03: Brakes. Revised Operator's Publications section. Removed Brake Inspection. Added Keyfob Battery Replacement section. Added Care of Graphics section.
Section 9 .....	Revised Log of Supplements.
Section 10 .....	Removed Door Position table from Landing Considerations. Revised Taxiing, Steering, and Braking Practices section.

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## List of Effective Pages

Use this page to determine the current effective date for each page in the POH. Supplements are issued individually and are controlled by the Log of Supplements Page in Section 9.

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Date \_\_\_\_\_  
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## Foreword

This Pilot's Operating Handbook (POH) has been prepared by Cirrus Design Corporation to familiarize operators with the aircraft. Read this POH carefully. It provides operational procedures that will assure the operator obtains the performance published in the manual, data designed to allow the most efficient use of the airplane, and basic information for maintaining the airplane in a "like new" condition.

• Note •

All limitations, procedures, maintenance & servicing requirements, and performance data contained in this POH are mandatory for compliance with FAA operating rules and for continued airworthiness of the airplane.

This POH includes the material required to be furnished to the pilot by the Federal Aviation Regulations (FARs) and additional information provided by Cirrus Design Corporation and constitutes the FAA Approved Airplane Flight Manual for the aircraft.

## The Pilot's Operating Handbook

This Pilot's Operating Handbook has been prepared using GAMA Specification #1 for Pilot's Operating Handbook, Revision 2, dated 18 October 1996 as the content model and format guide. However, some deviations from this specification were made for clarity. The POH is presented in loose-leaf form for ease in inserting revisions and is sized for convenient storage. Tabbed dividers throughout the POH allow quick reference to each section. Logical and convenient Tables of Contents are located at the beginning of each section to aid in locating specific data within that section. The POH is divided into ten sections as follows:

Section 1.....	General
Section 2.....	Limitations
Section 3.....	Emergency Procedures
Section 3A.....	Abnormal Procedures
Section 4.....	Normal Procedures
Section 5.....	Performance Data
Section 6.....	Weight & Balance/Equipment List
Section 7.....	Systems Description
Section 8.....	Handling, Servicing & Maintenance
Section 9.....	Supplements
Section 10.....	Safety Information

The data presented in this POH is the result of extensive flight tests and is approved by the Federal Aviation Administration. However, as new procedures or performance data are developed, the POH will be revised.

• Note •

It is the responsibility of the owner to ensure that the Pilot's Operating Handbook is current at all times. Therefore, it is very important that all revisions be properly incorporated into this POH as soon as they are available.

## Revising the Pilot's Operating Handbook

Two types of revisions may be issued for this POH: Temporary and Numbered.

Temporary revisions are printed on yellow paper, normally cover only one topic or procedure, and are issued to provide safety related information or other time sensitive information where the rigor of providing a numbered revision is not possible in the time allowed. All the information needed to properly file a temporary revision is included on the revision itself. Typically, a temporary revision is superseded and replaced by the next numbered revision.

Numbered revisions are printed on white paper, normally cover several subjects, and are issued as general updates to the POH. Each numbered revision includes an "Instruction Sheet," a "List of Effective Pages", and a "Revision Highlights" page. The "Instruction Sheet" is intended to assist the manual holder in removing superseded pages and inserting new or superseding pages. The "List of Effective Pages" shows the issue or revision status of all pages in the POH. The "Revision Highlights" page gives a brief description of changes made to each page in the current revision.

## Identifying Revised Material

Each page in the POH has revision identification at the lower inside corner opposite the page number. Original issue pages will be identified by the words "Original Issue" at this location. In the event that the majority of pages in the POH are revised, Cirrus may determine that it is more effective to reissue the POH. Reissued pages will be identified by the word "Reissue" followed by a letter indicating the reissue level; for example, "Reissue A" Revised pages will be identified by the word "Revision" followed by the revision number at this location; for example, "Revision 2" (Original Issue, Revision 2) or "Revision B1" (Reissue B, Revision 1).

Revised material on a page can be identified by a change bar located at the outside page margin. Revision bars are not used at reissues of the POH.

## Revisions to the Pilot's Operating Handbook

POH revisions, temporary revisions, and supplements can be downloaded from Cirrus Design at [www.cirrusaircraft.com](http://www.cirrusaircraft.com), or from the Authorized Service Center website.

Paper copies of POH revisions and supplements can be purchased from Cirrus Connection at [www.cirrusconnection.com](http://www.cirrusconnection.com).

### • Note •

If at any time it is found that the POH is not current, temporary revisions are missing, or applicable supplements are not included, contact Cirrus Design.

## Supplements

The Supplements section (Section 9) of this POH contains FAA Approved Supplements necessary to safely and efficiently operate the airplane when equipped with optional equipment not provided with the standard airplane or not included in the POH. Supplements are essentially “mini-handbooks” and may contain data corresponding to most sections of the POH. Data in a supplement either adds to, supersedes, or replaces similar data in the basic POH.

Section 9 includes a “Log of Supplements” page preceding all Cirrus Design Supplements produced for this airplane. The “Log of Supplements” page can be utilized as a “Table of Contents” for Section 9. If the airplane is modified at a non Cirrus Design facility through an STC or other approval method, it is the owner's responsibility to ensure that the proper supplement, if applicable, is installed in the POH and that the supplement is properly recorded on the “Log of Supplements” page.

FAA Approved POH Supplements must be in the airplane for flight operations when the subject optional equipment is installed or the special operations are to be performed.

## Retention of Data

In the event a new title page is issued, the weight and balance data changes, the equipment list changes, or the “Log of Supplements” is replaced, the owner must ensure that all information applicable to the airplane is transferred to the new pages and the aircraft records are current. It is not a requirement that owners retain information, such as supplements, that is not applicable to their airplane.

In the event a new POH is purchased, the owner must ensure that all information applicable to the airplane is transferred to the new POH and the aircraft records are current.

## **Warnings, Cautions, and Notes**

Warnings, Cautions, and Notes are used throughout this POH to focus attention on special conditions or procedures as follows:

### **• WARNING •**

Warnings are used to call attention to operating procedures which, if not strictly observed, may result in personal injury or loss of life.

### **• Caution •**

Cautions are used to call attention to operating procedures which, if not strictly observed, may result in damage to equipment.

### **• Note •**

Notes are used to highlight specific operating conditions or steps of a procedure.

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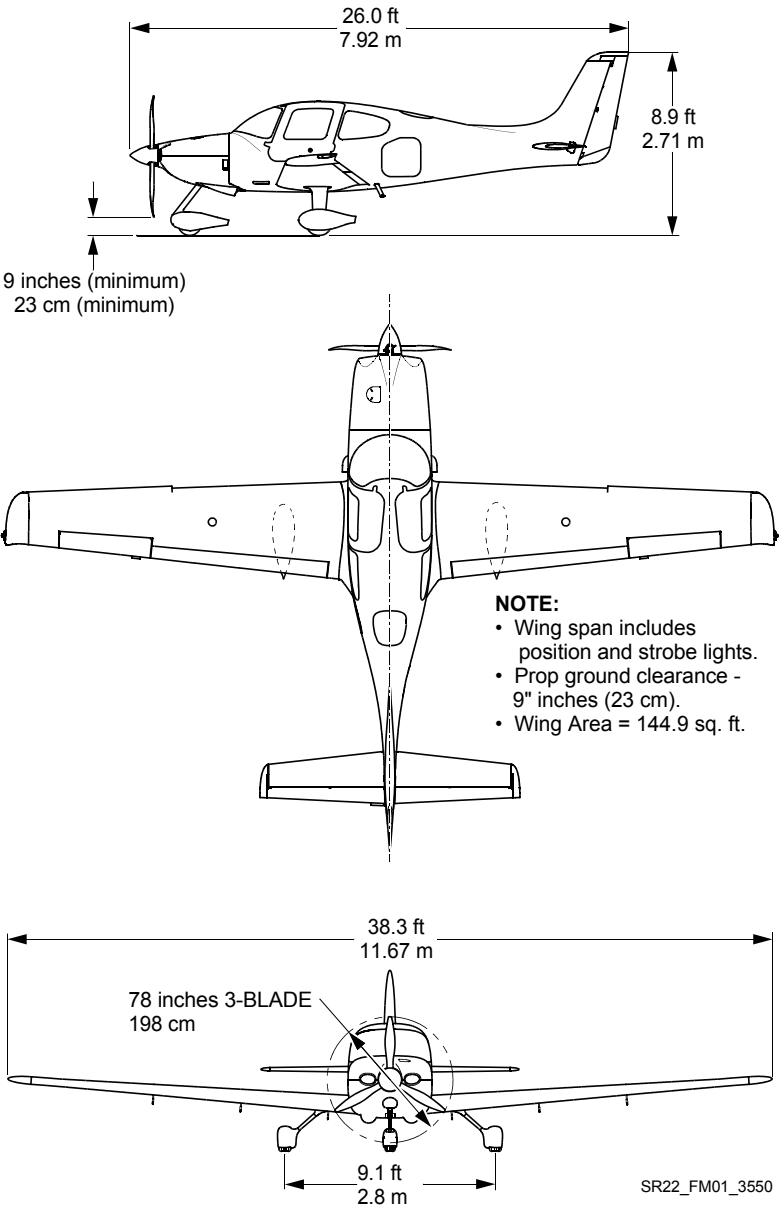
## Introduction

This section contains information of general interest to pilots and owners. You will find the information useful in acquainting yourself with the airplane, as well as in loading, fueling, sheltering, and handling the airplane during ground operations. Additionally, this section contains definitions or explanations of symbols, abbreviations, and terminology used throughout this handbook.

• Note •

For specific information regarding the organization of this Handbook, revisions, supplements, and procedures to be used to obtain publications, see the "Foreword" section.

All liquid volumes referenced in this publication are expressed in United States Customary Units, e.g., U.S. Gallons.



**Figure 1-1**  
**Airplane Three View**



## The Airplane

### Engine

Number of Engines..... 1  
Engine Manufacturer .....Teledyne Continental  
Engine Model ..... TSIO-550-K  
Engine Type ..... Turbocharged, direct drive, fuel injected, air cooled,  
horizontally opposed 6 cylinder engine with 550 cubic inch  
displacement.  
Horsepower Rating..... 315 bhp @ 2500 rpm

### Propeller

#### ***Hartzell Compact Series Lightweight Hub with Composite Blades***

Propeller Type ..... Constant Speed, Three Blade  
Model Number.....PHC-J3Y1F-1N/N7605(B)  
Diameter.....78.0"  
Model Number ..... PHC-J3Y1F-1N/N7605C(B)  
Diameter.....78.0"

## Fuel

Total Capacity ..... 94.5 U.S. Gallons (358.0 L)

Total Usable ..... 92.0 U.S. Gallons (348.0 L)

Approved Fuel Grades:

100 LL Grade Aviation Fuel (Blue)

100 (Formerly 100/130) Grade Aviation Fuel (Green)

## Oil

Oil Capacity (Sump) ..... 8 U.S. Quarts (7.6 L)

Oil Grades:

All Temperatures ..... 15W-50, 20W-50 or 20W-60

Above 40°F (4°C) ..... SAE 50, 20W50, or 20W60

Below 40°F (4°C) ..... SAE 30, 10W30, 15W50, or 20W50

## Maximum Certificated Weights

Maximum Gross for Takeoff ..... 3600 lb (1633 Kg)

Maximum Zero Fuel Weight ..... 3400 lb (1542 Kg)

Maximum Baggage Compartment Loading ..... 130 lb (59 Kg)

## Cabin and Entry Dimensions

Refer to the preceding figures for dimensions of the cabin interior and entry door openings.

## Baggage Spaces and Entry Dimensions

Refer to the preceding figures for dimensions of the baggage area and baggage door opening.

## Specific Loadings

Wing Loading ..... 24.8 lb per square foot

Power Loading ..... 11.4 lb per hp

## ***Section 2: Limitations***

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## Maneuver Limits

Aerobatic maneuvers are prohibited.

Spins are prohibited.

This airplane is certified in the normal category and is not designed for aerobatic operations. Only those operations incidental to normal flight are approved. These operations include normal stalls, chandelles, lazy eights, and turns in which the angle of bank is limited to 60°.

• Note •

Because the aircraft has not been certified for spin recovery, the Cirrus Airframe Parachute System (CAPS) must be deployed if the airplane departs controlled flight. Refer to Section 3, [Inadvertent Spin Entry](#).

## Flight Load Factor Limits

Flaps UP (0%), 3600 lb. .... +3.8g, -1.9g

Flaps 50%, 3600 lb..... +1.9g, 0g

Flaps 100%, 3600 lb..... +1.9g, 0g

## Minimum Flight Crew

The minimum flight crew is one pilot.

## Kinds of Operation

The aircraft is equipped and approved for the following type operations:

- VFR day and night.
- IFR day and night.

## Kinds of Operation Equipment List

The following listing summarizes the equipment required under Federal Aviation Regulations (FAR) Part 23 for airworthiness under the listed kind of operation. Those minimum items of equipment necessary under the operating rules are defined in FAR Part 91 and FAR Part 135 as applicable.

• Note •

All references to types of flight operations on the operating limitations placards are based upon equipment installed at the time of Airworthiness Certificate issuance.

System, Instrument, and/ or Equipment	Kinds of Operation				Remarks, Notes, and/or Exceptions
	VFR Day	VFR Nt.	IFR Day	IFR Nt.	
<b>Placards and Markings</b>					
Airplane Flight Manual (Included w/ POH)	1	1	1	1	
<b>Communications</b>					
VHF COM	—	—	1	1	
<b>Electrical Power</b>					
Battery 1	1	1	1	1	
Battery 2	—	—	1	1	
Alternator 1	1	1	1	1	
Alternator 2	—	—	1	1	
Amp Meter/Indication	1	1	1	1	

- n. *Serials w/ system software load 0764-21 or later:* Barometric vertical navigation (Baro-VNAV) operations may be conducted if SBAS is unavailable or disabled. The Perspective Integrated Avionics System will provide automatic, temperature-compensated glidepath vertical guidance and has been shown to meet the accuracy requirements of VFR/IFR enroute, terminal, and approach Baro-VNAV operations within the conterminous US and Alaska in accordance with the criteria in AC 20-138D.
5. Navigation using the Perspective Integrated Avionics System is not authorized in the following geographic areas:
    - a. north of 70°North latitude (northern polar region),
    - b. south of 70°South latitude (southern polar region),
    - c. north of the 65°North latitude between longitude 75°W and 120°W (Northern Canada),
    - d. south of 55°south latitude between longitude 120°E and 165°E (region south of Australia and New Zealand).
  6. The MFD checklist display supplements the Pilot Operating Handbook checklists and is advisory only. Use of the MFD checklists as the primary set of on-board airplane checklists is prohibited.
  7. The NAVIGATION MAP is intended only to enhance situational awareness. Use of the NAVIGATION MAP page for pilotage navigation is prohibited.
  8. Do not use SAFETAXI or CHARTVIEW functions as the basis for ground maneuvering. SAFETAXI and CHARTVIEW functions have not been qualified to be used as an Airport Moving Map Display (AMMD). SAFETAXI and CHARTVIEW are to be used by the flight crew to orient themselves on the airport surface to improve pilot situational awareness during ground operations.
  9. The TERRAIN PROXIMITY MAP is intended only to enhance situational awareness. Use of the TERRAIN PROXIMITY information for primary terrain avoidance is prohibited.
  10. LTNG information on the NAVIGATION MAP or WEATHER MAP is approved only as an aid to hazardous weather avoidance. Use of the WEATHER MAP for hazardous weather penetration is prohibited.

11. The SYNTHETIC VISION SYSTEM (SVS) cannot be used for flight guidance, navigation, traffic avoidance, or terrain avoidance. Maneuvering the airplane in any phase of flight such as taxi, takeoff, approach, landing, or roll out shall not be predicated on SVS imagery. The synthetic vision system is not intended to be used independently of traditional attitude instrumentation. Consequently, SVS is disabled when traditional attitude instrumentation is not available. Otherwise, the traditional attitude instrumentation will always be visible in the foreground with SVS features in the background.
12. Use of the TRAFFIC ADVISORY SYSTEM (TAS) to maneuver the airplane to avoid traffic is prohibited. The TAS is intended for advisory use only. TAS is intended only to help the pilot to visually locate traffic. It is the responsibility of the pilot to see and maneuver to avoid traffic.
13. Use of use of portable electronic devices during takeoff and landing is prohibited.

### **L-3 Skywatch Traffic Advisory System (Optional)**

1. Traffic information shown on the Perspective Integrated Avionics System displays is provided as an aid in visually acquiring traffic. Pilots must maneuver the aircraft based only upon ATC guidance or positive visual acquisition of conflicting traffic.
2. If the pilot is advised by ATC to disable transponder altitude reporting, Traffic Advisory System must be turned OFF.
3. When option installed, the appropriate revision of the L-3 Avionics Systems SkyWatch Traffic Advisory System Model SKY497 Pilot's Guide (p/n 009-10801-001) must be available to the pilot during flight.

### **L-3 Stormscope Weather Information System (Optional)**

1. Use of the Weather Information System is not intended for hazardous weather penetration (thunderstorm penetration). Weather information, as displayed on the Perspective Integrated Avionics System, is to be used only for weather avoidance, not penetration.
2. When option installed, the appropriate revision of the L-3 Avionics Systems WX500 Stormscope Series II Weather Mapping Sensor



User's Guide, (p/n 009-11501-001) must be available to the pilot during flight.

## Max Viz Enhanced Vision System (Optional)

1. The Enhanced Vision System (EVS) cannot be used for flight guidance, navigation, traffic avoidance, or terrain avoidance. Maneuvering the airplane in any phase of flight such as taxi, takeoff, approach, landing, or roll out shall not be predicated on EVS imagery. The EVS shall only be used as an aide to assist the flight crew to visually acquire objects normally viewed through the cockpit windows.
2. The appropriate revision of the Max Viz Enhanced Vision System Information Manual, (p/n 309100024) must be available to the pilot during flight.

## MD302 Standby Attitude Module (Optional)

1. Selection of the option menu of the MD302 is limited to ground or visual meteorological conditions.
2. The display has an operational lower temperature limit of -22°F (-30°C). Visibility of the display may be reduced between -4°F (-20°C) and -22°F (-30°C).
3. The appropriate revision of the Mid-Continent Instruments and Avionics MD302 Standby Attitude Module Pilot's Guide (p/n 9017846) must be available to the pilot whenever the system is in use.

## Air Conditioning System (Optional)

The use of Recirculation Mode during flight is prohibited.

## Inflatable Restraint System

Use of a child safety seat with the inflatable restraint system is prohibited.

## Flap Limitations

Approved Takeoff Settings ..... UP (0%) or 50%

Approved Landing Settings ..... 0%, 50%, or 100%

Do not use flaps above 17,500 feet MSL.

## Paint

To ensure that the temperature of the composite structure does not exceed 150°F (66°C), the outer surface of the airplane must be painted in accordance with the paint colors and schemes as specified in the Airplane Maintenance Manual. Refer to Airplane Maintenance Manual (AMM), Chapter 51, for specific paint requirements.

## Cirrus Airframe Parachute System (CAPS)

VPD Maximum Demonstrated Deployment Speed.....140 KIAS

• Note •

Refer to Section 10, *Cirrus Airframe Parachute System (CAPS)* for additional CAPS guidance.

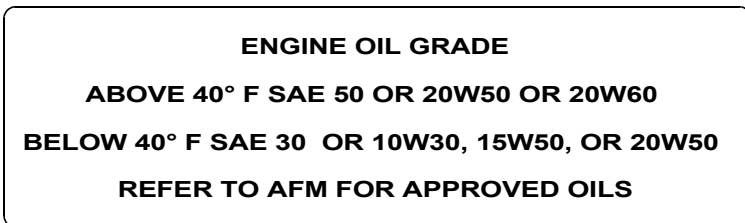
## Other Limitations

### Smoking

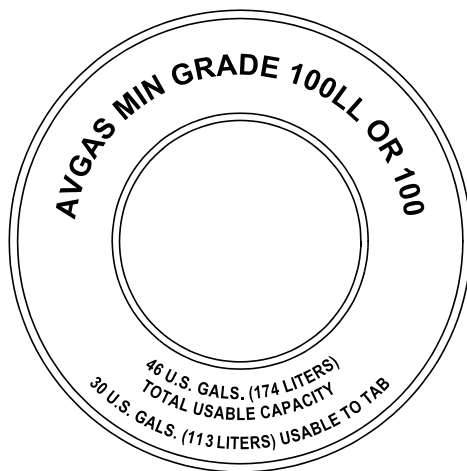
Smoking is prohibited in this airplane.

## Placards

*Engine compartment, inside oil filler access:*



*Wing, adjacent to fuel filler caps:*



*Upper fuselage, either side of CAPS rocket cover:*



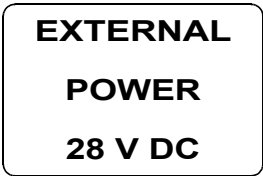
SR22\_FM02\_2680

**Figure 2-3**  
**Placards (Sheet 1 of 6)**

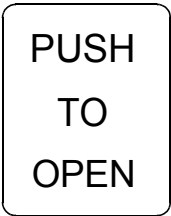
*Elevator and Rudder, both sides:*



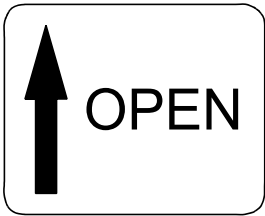
*Left fuselage, on external power supply door:*



*Doors, adjacent to latch:*

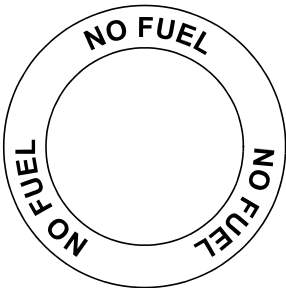


*Serials 0442 thru 1232*

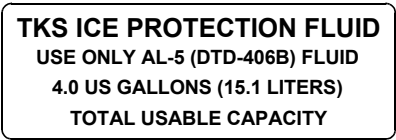


*Serials 1233 & subs*

*Wing, adjacent to fluid filler cap:*



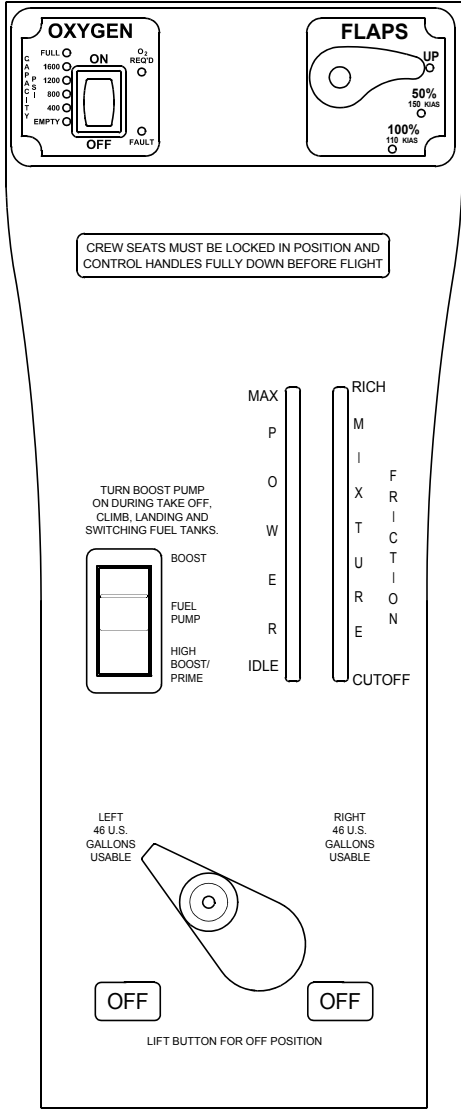
*Serials w/ Ice Protection.*



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**Figure 2-4**  
**Placards (Sheet 2 of 6)**

Engine control panel:



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Figure 2-5  
(Sheet 3 of 6)

*Wing, flap aft edge and fuselage vortex generator:*

**NO STEP**

*Cabin Door Window, lower edge, centered, applied upside down:*

**RESCUE: FRACTURE AND REMOVE WINDOW**

*Bolster Switch Panel, left edge:*

**THIS AIRCRAFT IS CERTIFIED FOR THE  
FOLLOWING FLIGHT OPERATIONS:  
DAY - NIGHT - VFR - IFR  
(WITH REQUIRED EQUIPMENT)  
FLIGHT INTO KNOWN ICING IS PROHIBITED  
OPERATE PER AIRPLANE FLIGHT MANUAL**

*Instrument Panel, left :*

**NORMAL CATEGORY AIRPLANE  
NO ACROBATIC MANEUVERS,  
INCLUDING SPINS, APPROVED**

---

**ABOVE 17,500 V<sub>NE</sub> AND V<sub>NO</sub>  
REDUCE LINEARLY WITH ALTITUDE:**

	V <sub>NE</sub>	V <sub>NO</sub>	
17,500	205	176	KIAS
25,000	175	150	KIAS

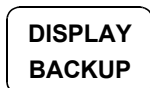
---

**MANEUVERING SPEED: V<sub>o</sub> 140 KIAS**

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**Figure 2-6  
(Sheet 4 of 6)**

*Instrument Panel, center:*



*Bolster Panel, both sides:*



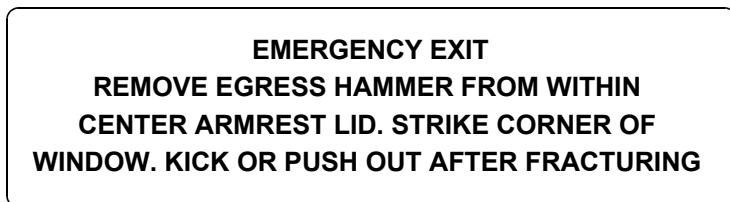
*Baggage Compartment, aft edge:*



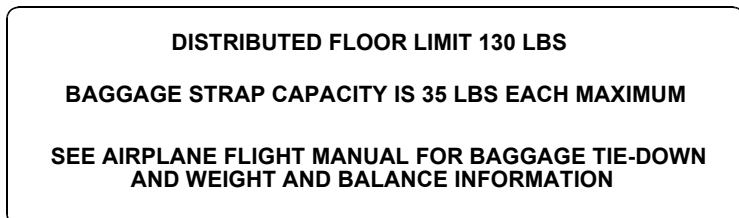
*Instrument Panel:*



*Cabin Window, above door latch:*



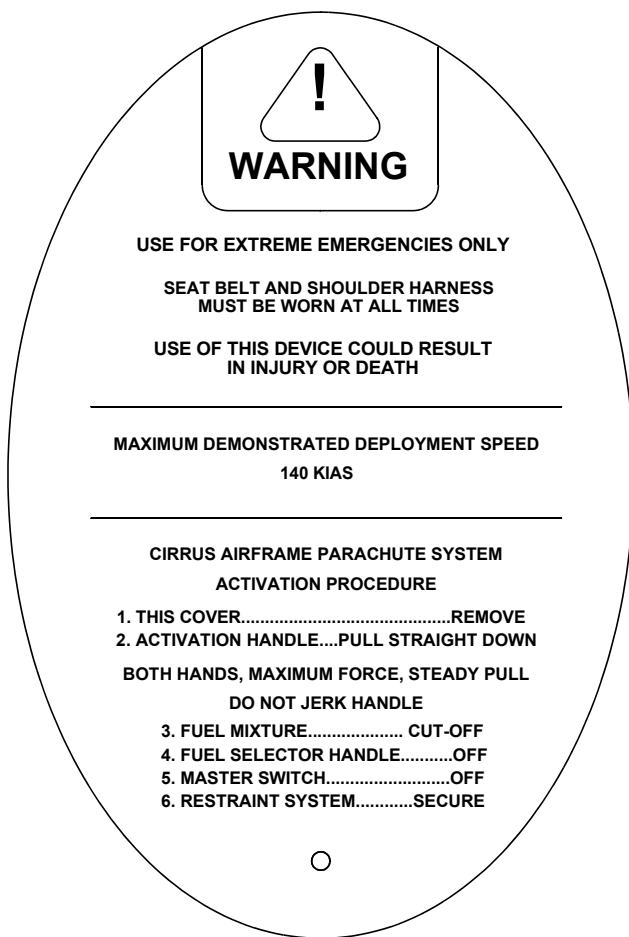
*Baggage Compartment Door, inside:*



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**Figure 2-7**  
**(Sheet 5 of 6)**

*CAPS Deployment Handle Cover, above pilot's right shoulder:*



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**Figure 2-8**  
**(Sheet 6 of 6)**



## Smoke and Fire

### Cabin Fire In Flight

1. Bat-Alt Master Switches..... OFF, AS REQ'D

2. Fire Extinguisher.....ACTIVATE

*If airflow is not sufficient to clear smoke or fumes from cabin:*

3. Cabin Doors..... PARTIALLY OPEN

4. Avionics Power Switch..... OFF

5. All other switches..... OFF

6. Land as soon as possible.

*If setting master switches off eliminated source of fire or fumes and airplane is in night, weather, or IFR conditions:*

7. Airflow Selector..... OFF

8. Bat-Alt Master Switches..... ON

9. Avionics Power Switch..... ON

10. Required Systems ..... ACTIVATE one at a time

11. Temperature Selector ..... COLD

12. Vent Selector ..... FEET/PANEL/DEFROST POSITION

13. Airflow Selector ..... SET AIRFLOW TO MAXIMUM

14. Panel Eyeball Outlets..... OPEN

15. Land as soon as possible.

#### **Amplification**

With Bat-Alt Master Switches OFF, engine will continue to run. However, no electrical power will be available.

If the airplane is in IMC conditions, turn ALT 1, ALT 2, and BAT 1 switches OFF. Power from battery 2 will keep the Primary Flight Display operational for approximately 30 minutes. If airplane is in day VFR conditions and turning off the master switches eliminated the fire situation, leave the master switches OFF. Do not attempt to isolate the source of the fire by checking each individual electrical component.

*(Continued on following page)*

If the cause of the fire is readily apparent and accessible, use the fire extinguisher to extinguish flames and land as soon as possible. Opening the vents or doors may feed the fire, but to avoid incapacitating the crew from smoke inhalation, it may be necessary to rid cabin of smoke or fire extinguishant.

If required to re-activate systems, pause several seconds between activating each system to isolate malfunctioning system. Continue flight to earliest possible landing with malfunctioning system off. Activate only the minimum amount of equipment necessary to complete a safe landing.

### Engine Fire In Flight

1. Mixture ..... CUTOFF
2. Fuel Pump..... OFF
3. Fuel Selector..... OFF
4. Airflow Selector..... OFF
5. Power Lever ..... IDLE
6. Ignition Switch..... OFF
7. Cabin Doors..... PARTIALLY OPEN
8. Land as soon as possible.

#### **Amplification**

If an engine fire occurs during flight, do not attempt to restart the engine.

## Ditching

1. Radio ..... Transmit (121.5 MHz) MAYDAY  
giving location and intentions
2. Transponder..... SQUAWK 7700
3. CAPS ..... ACTIVATE
4. Airplane..... EVACUATE
5. Flotation Devices ..... INFLATE WHEN CLEAR OF AIRPLANE

### ***Amplification***

If available, life preservers should be donned and life raft should be prepared for immediate evacuation upon touchdown.

Consider unlatching a door prior to assuming the emergency landing body position in order to provide a ready escape path.

It may be necessary to allow some cabin flooding to equalize pressure on the doors. If the doors cannot be opened, break out the windows with the egress hammer and crawl through the opening.

## Landing Without Elevator Control

1. Flaps ..... SET 50%
2. Trim..... SET 80 KIAS
3. Power..... AS REQUIRED FOR GLIDE ANGLE

### ***Amplification***

The pitch trim spring cartridge is attached directly to the elevator and provides a backup should you lose the primary elevator control system. Set elevator trim for a 80 KIAS approach to landing. Thereafter, do not change the trim setting until in the landing flare. During the flare, the nose-down moment resulting from a power reduction may cause the airplane to hit on the nosewheel. At touchdown, bring the power lever to idle.

## Engine System Emergencies

### Engine Partial Power Loss

1. Air Conditioner (if installed) ..... OFF
2. Fuel Pump ..... HIGH BOOST/PRIME
3. Fuel Selector ..... SWITCH TANKS
4. Mixture ..... CHECK appropriate for flight conditions
5. Power Lever ..... SWEEP
6. Ignition Switch ..... BOTH, L, then R
7. Land as soon as practical.

#### ***Amplification***

#### **• WARNING •**

If there is a strong smell of fuel in the cockpit, divert to the nearest suitable landing field. Fly a forced landing pattern and shut down the engine fuel supply once a safe landing is assured.

Indications of a partial power loss include fluctuating RPM, reduced or fluctuating manifold pressure, low oil pressure, high oil temperature, and a rough-sounding or rough-running engine. Mild engine roughness in flight may be caused by one or more spark plugs becoming fouled. A sudden engine roughness or misfiring is usually evidence of a magneto malfunction.

If for any reason the aircraft experiences an unexpected loss of normal manifold pressure perform the [\*Unexpected Loss Of Manifold Pressure\*](#) Checklist.

Low oil pressure may be indicative of an imminent engine failure. See [\*Oil Pressure Out of Range\*](#) Checklist in this Section for special procedures with low oil pressure.

A damaged (out-of-balance) propeller may cause extremely rough operation. If an out-of-balance propeller is suspected, immediately shut down engine and perform [\*Forced Landing\*](#) Checklist.

If the power loss is due to a fuel leak in the injector system, fuel sprayed over the engine may be cooled by the slipstream airflow which may prevent a fire at altitude. However, as the Power Lever is reduced during descent and approach to landing the cooling air may not be sufficient to prevent an engine fire.

### ***Amplification***

#### **• WARNING •**

Use caution after shutdown if STARTER circuit breaker required pull (failed relay or solenoid). If breaker is unknowingly or unintentionally reset, starter will instantly engage if Battery 1 power is supplied; creating a hazard for ground personnel.

Starter has been engaged for more than 15 seconds (starter limit is 10 seconds); if not manually engaged, such as during difficult start, this annunciation may indicate a failure of the starter solenoid or a stuck keyswitch.

### **Emergency Ground Egress**

1. Engine.....SHUTDOWN
2. Seat belts.....RELEASE
3. Airplane.....EXIT

### ***Amplification***

#### **• WARNING •**

While exiting the airplane, make sure evacuation path is clear of other aircraft, spinning propellers, and other hazards.

If the engine is left running, set the Parking Brake prior to evacuating the airplane.

If the doors cannot be opened, break out the windows with egress hammer, located in the console between the front seats, and crawl through the opening.

## CAPS Deployment

### • WARNING •

The maximum demonstrated deployment speed is 140 KIAS.

1. Activation Handle Cover ..... REMOVE
2. Activation Handle (Both Hands)..... PULL STRAIGHT DOWN

*After Deployment as time permits:*

3. Mixture ..... CUTOFF
4. Fuel Selector ..... OFF
5. Fuel Pump..... OFF
6. Bat-Alt Master Switches ..... OFF

Turn the Bat-Alt Master Switches off after completing any necessary radio communications.

7. Ignition Switch ..... OFF
8. ELT ..... ON
9. Seat Belts and Harnesses ..... TIGHTEN
10. Loose Items ..... SECURE
11. Assume emergency landing body position.
12. After the airplane comes to a complete stop, evacuate quickly and move upwind.

### ***Amplification***

### • WARNING •

Jerking or rapidly pulling the activation T-handle will greatly increase the pull forces required to activate the rocket. Use a firm and steady pulling motion – a “chin-up” type pull ensures successful activation.

The Cirrus Airframe Parachute System (CAPS) should be activated immediately in the event of a spin. It should also be used in other life-threatening emergencies where CAPS deployment is determined to be safer than continued flight and landing.

Expected impact in a fully stabilized deployment is equivalent to a drop from approximately 13 feet.

• **Caution** •

CAPS deployment will likely result in damage or loss to the airframe.

Several possible scenarios in which the activation of the CAPS would be appropriate are discussed in [Section 10: Safety Information](#) of this Handbook. These include:

- Mid-air collision
- Structural failure
- Loss of control
- Landing on an unprepared surface
- Pilot incapacitation

All pilots should carefully review the information on CAPS activation and deployment in Section 10 before operating the airplane.

CAPS Deployment at High Altitudes

For any indicated airspeed, as altitudes increase the true air speed of the deployment increases. Higher true air speeds increase the parachute inflation loads. Therefore, it is important for the operator takes all reasonable efforts to slow to the minimum possible airspeed prior to deploying the CAPS.

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## Introduction

This section provides procedures for handling abnormal system and/or flight conditions which, if followed, will maintain an acceptable level of airworthiness or reduce operational risk. The guidelines described in this section are to be used when an abnormal condition exists and should be considered and applied as necessary.

### • Caution •

If a Warning annunciation is illuminated in combination with any of the following Abnormal annunciations, the Warning annunciation takes precedence and shall be performed first.

## Abnormal Procedures Guidance

Although this section provides procedures for handling most abnormal system and/or flight conditions that could arise in the aircraft, it is not a substitute for thorough knowledge of the airplane and general aviation techniques. A thorough study of the information in this handbook while on the ground will help you prepare for time-critical situations in the air.

Sound judgment as well as thorough knowledge of the aircraft, its characteristics, and the flight manual procedures are essential in the handling of any abnormal system and/or flight condition. In addition to the outlined items in the Abnormal Procedures, the following steps are considered part of all abnormal situations:

- ***Maintain Aircraft Control***
- ***Analyze the Situation***
- ***Take Appropriate Action***

### Circuit Breakers

Many procedures involve manipulating circuit breakers. The following criteria should be followed during “Circuit Breaker” steps:

- Circuit breakers that are “SET” should be checked for normal condition. If the circuit breaker is not “Set”, it may be reset only once. If the circuit breaker opens again, do not reset.
- Circuit breakers that “PULL” should only be pulled and not reset.
- Circuit breakers that “CYCLE” should be pulled, delayed for several seconds, and reset only once. Allow sufficient cooling time for circuit breakers that are reset through a “CYCLE” procedure.

## Engine System

### Low Idle Oil Pressure

#### ***OIL PRESS Caution***

OIL PRESS

1. If In-Flight..... LAND AS SOON AS PRACTICAL

#### ***Amplification***

Oil pressure between 10 psi and 30 psi at or above 1000 RPM

This message will appear prior to engine start and should clear after engine start.

### Manifold Pressure High

#### ***MAN PRESSURE Caution***

MAN PRESSURE

1. Power Lever..... REDUCE to less than 36.5"
2. Flight..... CONTINUE

*If noticeable surging is present:*

3. Complete **Overboost / Pressure Relief Valve** Emergency Checklist

#### ***Amplification***

Manifold Pressure has exceeded caution limits. High Manifold Pressure may be a result of cold oil and the affect of high associated oil pressure on the wastegate controller. Maintain power at or below 36.5" by power lever management. If High Manifold Pressure persists when oil temperatures are greater than 150°F, MAP controller requires a maintenance adjustment. If engine surges are associated, MAP may be exceeding pressure relief valve (pop-off valve) threshold. Relief valve will protect induction manifolds from excessive pressure, but it may be a sign of a failed closed wastegate; if this is observed or suspected, complete the **Overboost / Pressure Relief Valve** Emergency Checklist.

## Starter Engaged

### **STARTER ENGAGED Caution**

START ENGAGE

#### **On-Ground**

1. Ignition Switch ..... DISENGAGE prior to 10 Seconds
2. Battery Switches ..... Wait 20 seconds before next start attempt  
*If starter does not disengage (relay or solenoid failure):*
3. BAT 1 Switch ..... OFF
4. Engine ..... SHUTDOWN
5. STARTER Circuit breaker ..... PULL

#### **In-Flight**

1. Ignition Switch ..... Ensure not stuck in START
2. STARTER Circuit breaker ..... PULL
3. Flight ..... CONTINUE  
Engine start will not be available at destination.

#### **Amplification**

##### **• WARNING •**

Use caution after shutdown if STARTER circuit breaker required pull (failed relay or solenoid). If breaker is unknowingly or unintentionally reset, starter will instantly engage if Battery 1 power is supplied; creating a hazard for ground personnel.

Starter has been engaged for more than 15 seconds (starter limit is 10 seconds); if not manually engaged, such as during difficult start, this annunciation may indicate a failure of the starter solenoid or a stuck keyswitch.

## Integrated Avionics System

### Avionics Switch Off

#### **AVIONICS OFF Caution**

AVIONICS OFF

1. AVIONICS Switch ..... ON, AS REQUIRED

#### **Amplification**

The AVIONICS master switch is off.

### PFD Cooling Fan Failure

#### **PFD FAN FAIL Advisory**

PFD FAN FAIL

1. AVIONICS FAN 2 Circuit Breaker ..... CYCLE

*If annunciation does not extinguish:*

- a. Hot cabin temperatures ..... LAND AS SOON AS PRACTICAL
- b. Cool cabin temperatures ..... CONTINUE, MONITOR

#### **Amplification**

The cooling fan for the PFD is inoperative.

### MFD Cooling Fan Failure

#### **MFD FAN FAIL Advisory**

MFD FAN FAIL

1. AVIONICS FAN 1 Circuit Breaker ..... CYCLE

*If annunciation does not extinguish:*

- a. High cabin temperatures .... LAND AS SOON AS PRACTICAL
- b. Low cabin temperatures ..... CONTINUE, MONITOR

#### **Amplification**

The cooling fan for the MFD is inoperative.

## Flight Displays Too Dim

1. INSTRUMENT dimmer knob..... OFF (full counter-clockwise)

*If flight displays do not provide sufficient brightness:*

2. Revert to standby instruments.

### **Amplification**

The instrument dimmer knob provides manual dimming control of the display screens, key and text backlighting, flap and Environmental Control System (ECS) status indicators, and standby instruments. Rotation of the dimmer knob fully counterclockwise disables the dimmer, and reverts to daytime lighting for all components. In the event of a dimmer control circuit failure, or to override the manual dimming circuit, pull the CABIN LIGHTS circuit breaker.

In daytime lighting (knob OFF/full counterclockwise, or with CABIN LIGHTS circuit breaker pulled):

- Electro-mechanical standby instruments, all avionics system keypads and the bolster switch panel are unlit
- MFD, PFD, and MD302 Standby Attitude Module (optional) screen illumination is controlled by automatic photocell (providing full brightness in high light conditions, only slightly reduced by darkness)
- ECS and control panels are backlit and their status lights are at maximum intensity

With active dimming (knob moved clockwise), the full bright position (full clockwise) applies maximum illumination to keys and switches, to standby instruments and to status lights, but the PFD, MFD, and MD302 Standby Attitude Module (optional) screen illumination is at a substantially reduced level (levels still appropriate for night flight). Maximum screen illumination (appropriate for daytime use) is with the dimmer OFF/full counterclockwise.

## ***Section 4: Normal Procedures***

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## Introduction

This section provides amplified procedures for normal operation. Normal procedures associated with optional systems can be found in [Section 9: Log of Supplements](#).

## Airspeeds for Normal Operation

Unless otherwise noted, the following speeds are based on a maximum weight of 3600 lb. and may be used for any lesser weight. However, to achieve the performance specified in Section 5 for takeoff and landing distance, the speed appropriate to the particular weight must be used.

### Takeoff Rotation:

- Normal, Flaps 50% ..... 77 KIAS
- Obstacle Clearance, Flaps 50% ..... 85 KIAS

### Enroute Climb, Flaps Up:

- Best Rate of Climb, SL ..... 103 KIAS
- Best Rate of Climb, 10,000 ..... 102 KIAS
- Best Angle of Climb, SL ..... 88 KIAS
- Best Angle of Climb, 10,000 ..... 88 KIAS
- Normal, Full Power, Full Rich Climb ..... 120 KIAS

### Landing Approach:

- Normal Approach, Flaps Up ..... 90-95 KIAS
- Normal Approach, Flaps 50% ..... 85-90 KIAS
- Normal Approach, Flaps 100% ..... 80-85 KIAS
- Short Field, Flaps 100% ( $V_{REF}$ ) ..... 79 KIAS

### Go-Around, Flaps 50%:

- Full Power ..... 80 KIAS

### Maximum Recommended Turbulent Air Penetration:

- 3600 lb ..... 140 KIAS
- 2900 lb ..... 123 KIAS

### Maximum Demonstrated Crosswind Velocity:

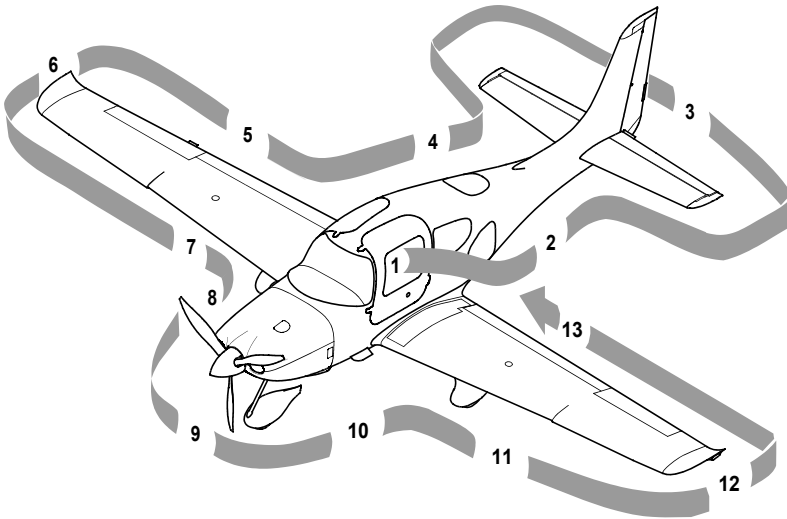
- Takeoff or Landing ..... 21 Knots

# Normal Procedures

## Preflight Inspection

Before carrying out preflight inspections, ensure that all required maintenance has been accomplished. Review your flight plan and compute weight and balance. Throughout the walk-around: check all hinges, hinge pins, and bolts for security; check skin for damage, condition, and evidence of delamination; check all control surfaces for proper movement and excessive free play; check area around liquid reservoirs and lines for evidence of leaking.

In cold weather, remove all frost, ice, or snow from fuselage, wing, stabilizers and control surfaces. Ensure that control surfaces are free of internal ice or debris. Check that wheel fairings are free of snow and ice accumulation. Check that Pitot probe warms within 30 seconds of setting Pitot Heat to ON.



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### 1. Cabin

- a. Required Documents..... On Board
- b. Avionics Power Switch .....OFF

*(Continued on following page)*

- d. Wheel Fairings ..... Security, Accumulation of Debris
- e. Tire ..... Condition, Inflation, and Wear

• **Caution** •

*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21:*  
Clean and inspect temperature indicator installed to piston housing. If indicator center is black, the brake assembly has been overheated. The brake linings must be inspected and O-rings replaced.

- f. Wheel and Brakes ..... Fluid Leaks, Evidence of Overheating, General Condition, and Security.
  - g. Chocks and Tiedown Ropes.....Remove
8. Nose, Right Side
- a. Vortex Generator ..... Condition
  - b. Cowling.....Attachments Secure
  - c. Exhaust Pipe ..... Condition, Security, and Clearance
  - d. Gascolator (underside).....Drain for 3 seconds, Sample
9. Nose gear, Propeller, and Spinner

• **WARNING** •

Keep clear of propeller rotation plane. Do not allow others to approach propeller.

- a. Tow Bar ..... Remove and Stow
  - b. Strut..... Condition
  - c. Wheel Fairing ..... Security, Accumulation of Debris
  - d. Wheel and Tire ..... Condition, Inflation, and Wear
  - e. Propeller ..... Check adequate Ground Clearance
  - f. Spinner ..... Condition, Security, and Oil Leaks
  - g. Air Inlets ..... Unobstructed
  - h. Alternator..... Condition
10. Nose, Left Side
- a. Landing Light..... Condition
  - b. Engine Oil.....Check 6-8 quarts, Leaks, Cap & Door Secure

*(Continued on following page)*

- c. Cowling ..... Attachments Secure
- d. External Power ..... Door Secure
- e. Vortex Generator ..... Condition
- f. Exhaust Pipe(s) ..... Condition, Security, and Clearance

11. Left Main Gear and Forward Wing

- a. Wheel fairings ..... Security, Accumulation of Debris
- b. Tire ..... Condition, Inflation, and Wear

• Caution •

*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21:*  
Clean and inspect temperature indicator installed to piston housing. If indicator center is black, the brake assembly has been overheated. The brake linings must be inspected and O-rings replaced.

- c. Wheel and Brakes ..... Fluid Leaks, Evidence of Overheating, General Condition, and Security.
- d. Chocks and Tiedown Ropes ..... Remove
- e. Fuel Drains (2 underside) ..... Drain and Sample
- f. Fuel Cap ..... Check Quantity and Secure
- g. Leading Edge and Stall Strips ..... Condition

12. Left Wing Tip

- a. Fuel Vent (underside) ..... Unobstructed
- b. Pitot Mast (underside) ..... Cover Removed, Tube Clear
- c. Strobe, Nav Light and Lens ..... Condition and Security
- d. Tip ..... Attachment

13. Left Wing Trailing Edge

- a. Flap And Rub Strips (If installed) ..... Condition and Security
- b. Aileron ..... Freedom of movement
- c. Aileron Gap Seal ..... Security
- d. Hinges, actuation arm, bolts, and cotter pins ..... Secure

indicate faulty grounding of one side of the ignition system or magneto timing set in advance of the specified setting.

## **Maximum Power Fuel Flow**

For maximum power operations (Power Lever full forward - 2500 RPM, 36.0 in.Hg manifold pressure) fuel flow should be in the green arc.

For any power setting greater than 30.5 in.Hg (cruise power) fuel flow is indicated by a dynamically calculated green arc displayed on the fuel gage. Fuel flow should be maintained within this arc by use of the mixture lever.

## Normal Takeoff

1. Brakes ..... RELEASE (Steer with Rudder Only)
2. Power Lever ..... FULL FORWARD
3. Engine Parameters ..... CHECK
4. Elevator Control ..... ROTATE Smoothly at 77-80 KIAS
5. At 90 KIAS, Flaps ..... UP

## Short Field Takeoff

1. Flaps ..... 50%
2. Brakes ..... HOLD
3. Power Lever ..... FULL FORWARD
4. Engine Parameters ..... CHECK
5. Brakes ..... RELEASE (Steer with Rudder Only)
6. Elevator Control ..... ROTATE Smoothly at 77 KIAS
7. Airspeed at Obstacle ..... 85 KIAS

HIGH BOOST /PRIME position and reset the mixture as required to maintain adequate stable fuel flow. Vapor lock is most often indicated by any or a combination of the following:

- Fluctuations in normal fuel flow possibly coupled with abnormal engine operation;
- Rising EGTs and TIT coupled with falling fuel flow
- Rising CHTs (late in the process)

## Cruise Climb: Lean of Peak Technique

1. Power Lever ..... REDUCE to 30.5 in.Hg
2. Mixture ..... LEAN to cyan target or less
3. Minimum Airspeed ..... 120 KIAS
4. Fuel Pump ..... BOOST
5. Oxygen ..... AS REQUIRED
  - a. Oxygen Masks/Cannulas ..... DON
  - b. Oxygen System ..... ON
  - c. Flow Rate ..... ADJUST for planned cruise altitude
  - d. Flowmeters and Quantity ..... MONITOR
6. Cylinder Head Temperatures ..... MONITOR

### ***Amplification***

Cruise climb with the mixture lever set to a lean mixture setting (LOP) is acceptable provided CHTs remain under 420°F. This climb procedure may not be possible in hot weather, but in moderate temperature conditions, LOP cruise climbs provide extended range and better fuel economy. Depending on aircraft weight and OAT, LOP cruise climbs will result in 600 to 700 FPM rates of climb at 130-140 KIAS.

Target fuel flow is calculated to provide the approximate Lean of Peak / "Best Economy" fuel-to-air ratio. Dependant on OAT and airspeed, this setting may not guarantee cylinder head temperatures below 420°F. If any CHT's are greater than 420°F, lean the mixture to maintain cylinders below 420°F. If cylinder head temperatures consistently exceed 420°F, climbs should be made at full rich mixture as described in the Climb Checklist.



## Cruise

1. Oxygen ..... AS REQUIRED
2. Cruise Altitude ..... ESTABLISHED
3. Power Lever ..... REDUCE to 30.5 in.Hg or less
4. Fuel Pump ..... AS REQUIRED

### • Note •

The Fuel Pump must be set to BOOST during maneuvering flight (i.e. flight training maneuvers, chandelles, stalls, etc.).

5. Mixture ..... ADJUST
6. Engine Parameters ..... MONITOR
7. Fuel Flow and Balance ..... MONITOR  
*If any CHT's exceed 420°F:*
8. Mixture ..... LEAN 0.5 GPH and MONITOR

## Amplification

Recommended cruise is at a Lean of Peak / “Best Economy” mixture setting. Cruise leaning, i.e. leaning below full rich fuel flow, is only approved with manifold pressure settings of 30.5 in.Hg or less. Once power is reduced below this level, the green arc expands and a cyan colored Target Fuel Flow que is displayed on the fuel flow gage. With higher manifold pressures, the fuel flow gage provides a narrow green arc which defines full rich fuel flow settings.

Target fuel flow is determined using a calculated engine air flow based on Engine Speed, Manifold Air Temperature and Manifold Air Pressure and indicates a fuel flow that will give the approximate air-to-fuel ratio for best economy operation. Alternatively, the mixture can be set by finding a fuel flow that provides peak TIT and then leaning until TIT is 50°-75°F less than its peak value.

Target Fuel Flow is advisory only. This indicator or the Peak leaning method will provide an initial lean point only. As this setting is dependant on ambient air temperatures, it may not ensure sufficient cylinder cooling. If any CHT's are greater than 420°F, lean the mixture to maintain cylinders below 420°F. As an approximation, a 0.5 GPH reduction in fuel flow will reduce CHT's by 15°F.

Running the engine at mixture levels leaner than the target will improve cooling, but provide lower cruise power because engine power scales in proportion to fuel flow when the engine is running at

lean of peak. Other than lower cruise power, the only undesirable affect of an overly lean-of-peak setting is engine misfire. Cruise mixture should be rich enough to avoid lean misfire, but no richer than target indicator for cruise.

• Note •

The fuel pump is used for vapor suppression during climb. It is also recommended that the fuel pump be left in BOOST after leveling off for 30 minutes following a climb and anytime fuel flow or EGT anomalies occur. Under some previously described extreme environmental conditions, the use of HIGH BOOST/PRIME may be required for vapor suppression during cruise flight. The fuel pump can be returned to the BOOST or OFF position as conditions permit.

## Descent

1. Oxygen ..... AS REQUIRED
2. Altimeter ..... SET
3. Cabin Heat/Defrost ..... AS REQUIRED
4. Landing Light ..... ON
5. Fuel System ..... CHECK
6. Power Lever ..... AS REQUIRED

### *For Rapid Descent:*

- a. Power Lever ..... Smoothly REDUCE MAP 18 to 20 in.Hg
7. Mixture ..... AS REQUIRED

### *For Rapid Descent:*

- a. Mixture ..... Maintain CHTs above 240°F
8. Brake Pressure ..... CHECK

## **Amplification**

Avoid prolonged idle settings. Maintain a CHT of 240°F (116°C) or greater.

## Before Landing

1. Seat Belt and Shoulder Harness ..... SECURE
2. Fuel Pump ..... BOOST
3. Mixture ..... FULL RICH
4. Flaps ..... AS REQUIRED
5. Autopilot ..... AS REQUIRED

## Normal Landing

1. Flaps ..... 100%
  2. Airspeed ..... 80-85 KIAS
  3. Power Lever ..... AS REQUIRED
- After touchdown:*
4. Brakes ..... AS REQUIRED

### ***Amplification***

#### **• Caution •**

Landings should be made with full flaps. Landings with less than full flaps are recommended only if the flaps fail to deploy or to extend the aircraft's glide distance due to engine malfunction. Landings with flaps at 50% or 0%; power should be used to achieve a normal glide path and low descent rate. Flare should be minimized.

Normal landings are made with full flaps with power on or off. Surface winds and air turbulence are usually the primary factors in determining the most comfortable approach speeds.

Actual touchdown should be made with power off and on the main wheels first to reduce the landing speed and subsequent need for braking. Gently lower the nose wheel to the runway after airplane speed has diminished. This is especially important for rough or soft field landings.

### ***Crosswind Landings***

Normal crosswind landings are made with full flaps. Avoid prolonged slips. After touchdown, hold a straight course with rudder and brakes as required. The maximum allowable crosswind velocity is dependent upon pilot capability as well as aircraft limitations. Operation in direct crosswinds of 21 knots has been demonstrated.

## Short Field Landing

1. Flaps ..... 100%
2. Airspeed ..... 79 KIAS
3. Power Lever ..... AS REQUIRED  
*After clear of obstacles:*
4. Power Lever ..... REDUCE TO IDLE  
*After touchdown:*
5. Brakes ..... MAXIMUM

### **Amplification**

For a short field landing in smooth air conditions, make an approach at 79 KIAS with full flaps using enough power to control the glide path (slightly higher approach speeds should be used under turbulent air conditions).

After all approach obstacles are cleared, progressively reduce power to reach idle just before touchdown and maintain the approach speed by lowering the nose of the airplane.

Touchdown should be made power-off and on the main wheels first. Immediately after touchdown, lower the nose wheel and apply braking as required. For maximum brake effectiveness, retract the flaps, hold the control yoke full back, and apply maximum brake pressure without skidding.

## Balked Landing/Go-Around

1. Autopilot ..... DISENGAGE
2. Power Lever ..... FULL FORWARD
3. Flaps ..... 50%
4. Airspeed ..... 80-85 KIAS  
*After clear of obstacles:*
5. Flaps ..... UP

### Amplification

In a balked landing (go around) climb, disengage autopilot, apply full power, then reduce the flap setting to 50%. If obstacles must be cleared during the go around, climb at 80-85 KIAS with 50% flaps. After clearing any obstacles, retract the flaps and accelerate to the normal flaps up climb speed.

## After Landing

1. Power Lever ..... 1000 RPM
2. Fuel Pump ..... OFF or BOOST
3. Mixture ..... LEAN to obtain maximum idle RPM
4. Flaps ..... UP
5. Transponder ..... STBY
6. Lights ..... AS REQUIRED
7. Pitot Heat ..... OFF

• Note •

As the airplane slows the rudder becomes less effective and taxiing is accomplished using differential braking.

## Shutdown

1. Fuel Pump (if used) ..... OFF
2. Throttle..... IDLE
3. Ignition Switch..... CYCLE
4. Mixture .....CUTOFF
5. All Switches ..... OFF
6. Magnetos ..... OFF
7. ELT..... TRANSMIT LIGHT OUT
8. Chocks, Tie-downs, Pitot Covers..... AS REQUIRED

### ***Amplification***

#### **• Caution •**

Note that the engine hesitates as the switch cycles through the “OFF” position. If the engine does not hesitate, one or both magnetos are not grounded. Prominently mark the propeller as being “Hot,” and contact maintenance personnel immediately.

After a hard landing, the ELT may activate. If this is suspected, press the RESET button.

## Stalls

Aircraft stall characteristics are conventional. Power-off stalls may be accompanied by a slight nose bobbing if full aft stick is held. Power-on stalls are marked by a high sink rate at full aft stick. Power-off stall speeds at maximum weight for both forward and aft CG positions are presented in Section 5 - *Stall Speeds*.

When practicing stalls at altitude, as the airspeed is slowly reduced, you will notice a slight airframe buffet, hear the stall speed warning horn sound between 5 and 10 knots before the stall, and see the Crew Alerting System display a STALL Warning annunciation. Normally, the stall is marked by a gentle nose drop and the wings can easily be held level or in the bank with coordinated use of the ailerons and rudder. Upon stall warning in flight, recovery is accomplished by immediately by reducing back pressure to maintain safe airspeed, adding power if necessary and rolling wings level with coordinated use of the controls.

### • WARNING •

Extreme care must be taken to avoid uncoordinated, accelerated or abused control inputs when close to the stall, especially when close to the ground.



## Environmental Considerations

### Cold Weather Operation

#### • Caution •

An engine that has been superficially warmed, may start and appear to run satisfactorily, but can be damaged from lack of lubrication due to the congealed oil blocking proper oil flow through the engine. The amount of damage will vary and may not become evident for many hours. However, the engine may be severely damaged and may fail shortly following application of high power. Proper procedures require thorough application of preheat to all parts of the engine. Hot air must be applied directly to the oil sump and external oil lines as well as the cylinders, air intake and oil cooler. Because excessively hot air can damage non-metallic components such as composite parts, seals, hoses, and drives belts, do not attempt to hasten the preheat process.

#### ***Starting***

If the engine has been cold soaked, it is recommended that the propeller be pulled through by hand several times to break loose or limber the oil. This procedure will reduce power draw on the battery if a battery start is made.

When the engine has been exposed to temperatures at or below 20°F (-7°C) for a period of two hours or more, the use of an external pre-heater and external power is recommended. Failure to properly preheat a cold-soaked engine may result in oil congealing within the engine, oil hoses, and oil cooler with subsequent loss of oil flow, possible internal damage to the engine, and subsequent engine failure.

If the engine does not start during the first few attempts, or if engine firing diminishes in strength, the spark plugs have probably frosted over. Preheat must be used before another start is attempted.

#### • Note •

When the oil temperature has reached 100°F (38°C) and oil pressure does not exceed 70 psi at 2500 RPM, the engine has been warmed sufficiently to accept full rated power.

1. Ignition Switch ..... OFF

• **WARNING** •

Use caution when pulling the propeller through by hand. Make sure ignition switch is OFF, keys are out of ignition, and then act as if the engine will start.

2. Propeller ..... Hand TURN several rotations  
3. External Power (If applicable) ..... CONNECT  
4. Brakes ..... HOLD  
5. Bat Master Switches ..... ON (check voltage)  
6. Mixture ..... FULL RICH  
7. Power lever ..... FULL FORWARD  
8. Fuel Pump ..... HIGH BOOST/PRIME, then BOOST

• **Note** •

In temperatures down to 20°F, hold Fuel Pump switch to HIGH BOOST/PRIME for 15 seconds prior to starting.

9. Propeller Area ..... CLEAR  
10. Power Lever ..... OPEN ¼ INCH  
11. Ignition Switch ..... START (Release after engine starts)

• **Caution** •

Limit cranking to intervals of 10 seconds with a 20 second cooling period between cranks.

12. Power Lever ..... RETARD (to maintain 1000 RPM)  
13. Oil Pressure ..... CHECK  
14. Alt Master Switches ..... ON  
15. Avionics Power Switch ..... ON  
16. Engine Parameters ..... MONITOR  
17. External Power (If applicable) ..... DISCONNECT  
18. Amp Meter/Indication ..... CHECK  
19. Strobe Lights ..... ON

## Hot Weather Operation

Avoid prolonged engine operation on the ground. Fuel BOOST must be ON for engine start and takeoff, and should be ON during climb for vapor suppression which could occur under hot ambient conditions or after extended idle.

### ***Ground Operation of Air Conditioning (If Installed)***

• Note •

To facilitate faster cabin cooling, prior to engine start leave the cabin doors open for a short time to allow hot air to escape cabin.

1. Control Panel ..... SELECT Desired Mode and Temperature
2. Voltage ..... MONITOR

• Note •

Decrease electrical load if battery discharge is noted.

3. Annunciator Lights ..... CHECK
  - a. Verify ALT 1 caution light out and positive amps indication.
4. Engine Parameters ..... CHECK

## Extended Ground Operation

For airplanes that experience prolonged engine operation on the ground, the following procedure is recommended to reduce potential for spark plug lead fouling and lead build-up on engine valve guides.

1. Set throttle to 1200 RPM.
2. Lean the mixture for maximum RPM.
3. Reduce throttle to RPM for continued ground operations (800 - 1000 RPM is recommended).

• WARNING •

Before takeoff, the mixture lever must be returned to the full forward/rich position.

• Note •

If further ground operations will be required after the BEFORE TAKEOFF checklist is completed, lean the mixture again (as described above) until ready for the TAKEOFF checklist.

## Noise Characteristics/Abatement

The certificated noise levels for the aircraft established in accordance with FAR 36 Appendix G are:

Configuration	Actual	Maximum Allowable
Hartzell 3-blade Propeller PHC-J3Y1F-1N/N7605(B)	81.5 dB(A)	88.0 dB(A)
Hartzell 3-blade Propeller PHC-J3Y1F-1N/N7605C(B)	81.5 dB(A)	88.0 dB(A)

No determination has been made by the Federal Aviation Administration that the noise levels of this airplane are or should be acceptable or unacceptable for operation at, into, or out of, any airport. The above noise levels were established at 3600 pounds takeoff weight and 2500 RPM.

Recently, increased emphasis on improving environmental quality requires all pilots to minimize the effect of airplane noise on the public. The following suggested procedures minimize environmental noise when operating the aircraft.

• Note •

Do not follow these noise abatement procedures where they conflict with Air Traffic Control clearances or instructions, weather considerations, or wherever they would reduce safety.

1. When operating VFR over noise-sensitive areas, such as outdoor events, parks, and recreational areas, fly not less than 2000 feet above the surface even though flight at a lower level may be allowed.
2. For departure from or approach to an airport, avoid prolonged flight at low altitude near noise-sensitive areas.

## Fuel Conservation

Minimum fuel use at cruise will be achieved using Lean-of-Peak Cruise Climb.

## ***Section 5: Performance Data***

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## Introduction

Performance data in this section are presented for operational planning so that you will know what performance to expect from the airplane under various ambient and field conditions. Performance data are presented for takeoff, climb, and cruise (including range & endurance).

*Aircraft with optional Air Conditioning System:* Brake Horsepower is reduced by approximately 6 BHP.

## Associated Conditions Affecting Performance

Computed performance data in this section are based upon data derived from actual flight testing with the airplane and engine in good condition and using average piloting techniques. Unless specifically noted in the “Conditions” notes presented with each table, ambient conditions are for a standard day (refer to Section 1, [Meteorological Terminology](#)). Flap position as well as power setting technique is similarly noted with each table.

The charts in this section provide data over temperature ranges as specified on the chart. If ambient temperature is below the chart value, use the lowest temperature shown to compute performance. This will result in more conservative performance calculations. If ambient temperature is above the chart value, use caution as performance degrades rapidly at higher temperatures.

## Demonstrated Operating Temperature

Satisfactory engine cooling has been demonstrated for this airplane with an outside air temperature 23°C above standard. The value given is not considered an operating limitation. Reference should be made to Section 2, [Powerplant Limitations](#) for operating limitations.

## Airspeed Calibration: Normal Static Source

**Conditions:**

- Power for level flight or maximum continuous, whichever is less.

• Note •

Indicated airspeed values assume zero instrument error.

KIAS	KCAS		
	Flaps 0%	Flaps 50%	Flaps 100%
60	57	50	56
70	68	66	69
80	79	80	80
90	89	92	91
100	100	102	102
110	111	113	113
120	121	121	
130	132	133	
140	142	144	
150	152	154	
160	163		
170	173		
180	183		
190	193		
200	203		
210	213		



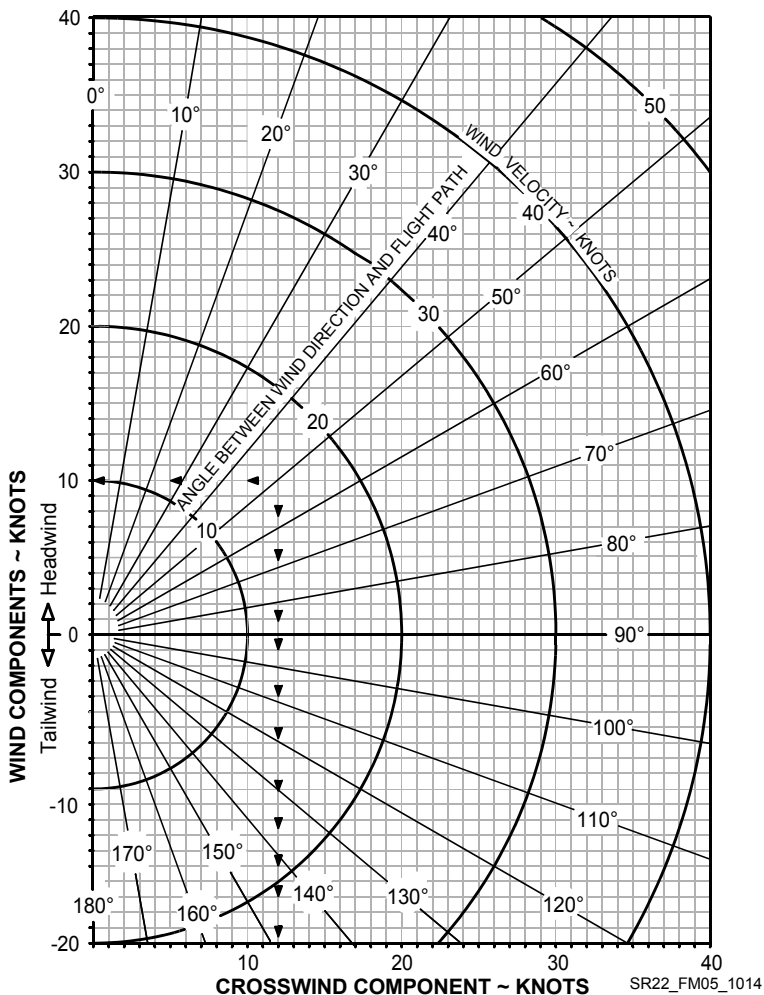
## Wind Components

### Example:

- Runway Heading ..... 10°
- Wind Direction ..... 60°
- Wind Velocity ..... 15 Knots

• Note •

The max demonstrated crosswind is 21 knots. Value not considered limiting.



## Takeoff Distance

### Conditions:

- Winds.....Zero
  - Runway.....Dry, Level, Paved
  - Flaps.....50%
  - Air Conditioner.....OFF
  - Power:
  - Throttle .....Full Open
  - Mixture.....Set per Placard
- Set prior to brake release for short field takeoff.

The following factors are to be applied to the computed takeoff distance for the noted condition:

Headwind - Subtract 10% from computed distance for each 12 knots headwind.

Tailwind - Add 10% for each 2 knots tailwind up to 10 knots.

Dry Grass Runway - Increase distances by 15% of the ground roll distance.

Sloped Runways - Increase distances by 22% of the ground roll value at Sea Level, 30% of the ground roll value at 5000 ft, 43% of the ground roll value at 10000 ft for each 1% of upslope; decrease distances by 7% of the ground roll value at Sea Level, 10% of the ground roll value at 5000 ft, and 14% of the ground roll value at 10000 ft for each 1% of downslope.

### • Caution •

The above corrections for runway slope are required to be included herein under FAR 23. They should be used with caution since published runway slope data is usually the net slope from one end of the runway to the other. Many runways will have portions of their length at greater or lesser slopes than the published slope, lengthening (or shortening) takeoff ground run values estimated from the published slope as described above.

If brakes are not held while applying power, distances apply from the point where full throttle and mixture setting is complete.

For operation in outside air temperatures colder than the Takeoff Distance table provides, use the coldest data shown.

For operation in outside air temperatures warmer than the Takeoff Distance table provides, use caution.

Aircraft with optional Air Conditioning System: Add 100 feet to ground roll distance and 150 feet to distance over 50' obstacle if Air Conditioner is ON during takeoff.

## Takeoff Climb Gradient

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Full Rich
- Flaps ..... 50%
- Airspeed ..... Best Rate of Climb

### • Note •

Climb Gradient values shown are the gain in altitude for the horizontal distance traversed expressed as Feet per Nautical Mile.

For operation in air colder than this table provides, use the coldest (left-most) data shown.

For operation in air warmer than this table provides, use caution.

Weight  LB	Press Altitude  FT	Climb Speed  KIAS	CLIMB GRADIENT ~ Feet per Nautical Mile					
			Temperature ~°C					ISA
			-20	0	20	40	50	
3600	SL	91	1020	879	752	634	579	782
	2000	91	958	823	701	589	537	755
	4000	91	898	770	654	547	496	728
	6000	91	841	719	608	506	458	702
	8000	91	787	671	565	468	422	676
	10000	91	735	625	524	431	387	651
2900	SL	94	1303	1148	1002	864	797	1038
	2000	94	1251	1097	952	815	750	1016
	4000	93	1196	1043	900	765	701	991
	6000	93	1137	986	845	713	650	964
	8000	92	1077	928	790	660	599	935
	10000	92	1015	869	733	607	546	904

## Takeoff Rate of Climb

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Full Rich
- Flaps ..... 50%
- Airspeed ..... Best Rate of Climb

### • Note •

Rate-of-Climb values shown are change in altitude in ft per unit time expressed in Feet per Minute

For operation in air colder than this table provides, use the coldest (left-most) data shown.

For operation in air warmer than this table provides, use caution.

Aircraft with optional Air Conditioning System: Maximum rate of climb performance is reduced by approximately 50 feet per minute if system is ON.

For maximum climb performance the air-conditioner should be off.

Weight  LB	Press Altitude  FT	Climb Speed  KIAS	RATE OF CLIMB ~ Feet per Minute					
			Temperature ~°C					ISA
			-20	0	20	40	50	
3600	SL	91	1462	1314	1166	1019	946	1203
	2000	91	1425	1277	1130	983	910	1196
	4000	91	1388	1240	1093	947	874	1189
	6000	91	1352	1204	1057	910	837	1182
	8000	91	1315	1167	1020	874	801	1175
	10000	91	1278	1131	984	838	765	1168
2900	SL	94	1880	1730	1570	1404	1318	1611
	2000	94	1867	1709	1542	1370	1282	1618
	4000	93	1847	1681	1508	1329	1238	1621
	6000	93	1819	1646	1466	1282	1189	1619
	8000	92	1784	1604	1418	1228	1132	1613
	10000	92	1742	1555	1364	1169	1070	1602

## Enroute Climb Gradient

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Maintain Fuel Flow in GREEN ARC
- Flaps ..... 0% (UP)
- Airspeed ..... 120 KIAS

### • Note •

Climb Gradient values shown are the gain in altitude for the horizontal distance traversed expressed as Feet per Nautical Mile.

For operation in air colder than this table provides, use the coldest (left-most) data shown.

For operation in air warmer than this table provides, use caution.

Weight  LB	Press Altitude  FT	Climb Speed  KIAS	CLIMB GRADIENT - Feet per Nautical Mile						
			Temperature ~°C						
			-40	-20	0	20	40	50	ISA
3600	S.L.	120	931	798	679	571	473	427	597
	2000	120	866	740	627	524	430	386	569
	4000	120	804	685	577	480	390	349	542
	6000	120	746	632	530	438	353	313	516
	8000	120	690	583	486	398	317	279	490
	10000	120	638	536	444	360	284	248	466
	12000	120	588	491	404	325	252	218	442
	14000	120	541	449	367	292	222	190	419
	16000	120	497	410	332	260	195	164	397
	18000	120	455	373	299	231	169	139	376
	20000	120	415	337	267	203	144	117	356
	22000	120	390	245	119	9			306
	24000	120	280	147	32				230
	25000	120	229	102					194
2900	S.L.	120	1173	998	856	736	629	579	765
	2000	120	1083	932	806	695	594	546	744
	4000	120	1012	878	763	657	559	511	725
	6000	120	953	831	722	619	521	472	706
	8000	120	903	787	680	578	478	428	685
	10000	120	856	743	636	531	428	377	662
	12000	120	808	695	585	478	371	318	634
	14000	120	757	642	528	416	305	250	601
	16000	120	701	581	463	346	230	173	562
	18000	120	637	512	388	266	146	86	517
	20000	120	564	433	303	176	51		465
	22000	120	482	344	209	77			405
	24000	120	389	245	105				337
	25000	120	339	192	49				300

## Enroute Rate of Climb

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Maintain Fuel Flow in GREEN ARC
- Flaps ..... 0% (UP)
- Airspeed ..... 120 KIAS

### • Note •

Rate-of-Climb values shown are change in altitude in ft per unit time expressed in Feet per Minute.

For operation in air colder than this table provides, use the coldest (left-most) data shown.

For operation in air warmer than this table provides, use caution.

Aircraft with optional Air Conditioning System: Maximum rate of climb performance is reduced by approximately 50 feet per minute if system is ON.

For maximum climb performance the air-conditioner should be off.

Weight LB	Press Altitude FT	Climb Speed KIAS	RATE OF CLIMB ~ Feet per Minute						
			Temperature ~°C						
			-40	-20	0	20	40	50	ISA
3600	S.L.	120	1635	1465	1298	1133	970	890	1174
	2000	120	1580	1410	1243	1079	917	836	1153
	4000	120	1524	1355	1189	1025	863	783	1131
	6000	120	1469	1301	1135	971	810	730	1110
	8000	120	1414	1246	1081	918	757	677	1089
	10000	120	1359	1191	1027	864	703	624	1067
	12000	120	1304	1137	973	811	650	571	1046
	14000	120	1249	1083	919	757	597	518	1025
	16000	120	1194	1028	865	704	544	465	1003
	18000	120	1140	974	811	650	491	412	982
	20000	120	1085	920	758	597	439	360	961
	22000	120	1064	698	353	27			855
	24000	120	799	438	98				666
	25000	120	668	309					571
2900	S.L.	120	2045	1822	1630	1456	1289	1206	1498
	2000	120	1964	1768	1594	1427	1262	1179	1502
	4000	120	1908	1731	1566	1401	1233	1146	1508
	6000	120	1869	1704	1540	1370	1193	1100	1515
	8000	120	1841	1677	1508	1329	1137	1037	1517
	10000	120	1815	1646	1466	1271	1060	949	1512
	12000	120	1784	1603	1405	1189	956	833	1496
	14000	120	1742	1541	1320	1079	819	682	1466
	16000	120	1680	1454	1205	934	643	491	1418
	18000	120	1593	1336	1054	749	424	255	1348
	20000	120	1473	1180	860	517	156		1253
	22000	120	1314	979	618	235			1130
	24000	120	1109	729	323				975
	25000	120	988	583	154				884

## Cruise Performance

### Conditions:

- Weight ..... 3400 LB
- Winds ..... Zero

### • Note •

Subtract 10 KTAS if nose wheel pant and fairing removed. Lower KTAS by 10% if nose and main wheel pants and fairings are removed.

Aircraft with optional Air Conditioning System: Cruise performance is reduced by 2 knots. For maximum performance, turn air conditioner off.

*Aircraft with optional Enhanced Vision System:* Cruise performance is reduced by up to 1 knot.

The Values shown in gray in the table below may not be achievable for engine temperature management reasons. This data is provided for purposes of interpolating between points.

CRUISE PERFORMANCE			ISA -30°C		ISA		ISA +30°C	
Altitude (ft MSL)	Power (% of 315)	FF (GPH)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)
2000	85%	18.3	164	9.0	170	9.3	176	9.6
	75%	16.4	157	9.6	162	9.9	167	10.2
	65%	14.6	148	10.2	154	10.5	158	10.8
	55%	12.7	138	10.9	143	11.2	147	11.5
4000	85%	18.3	168	9.1	174	9.5	179	9.8
	75%	16.4	160	9.7	165	10.1	170	10.4
	65%	14.6	151	10.3	156	10.7	161	11.0
	55%	12.7	140	11.0	145	11.4	149	11.8
6000	85%	18.3	171	9.3	177	9.7	183	10.0
	75%	16.4	163	9.9	168	10.2	174	10.6
	65%	14.6	153	10.5	159	10.9	163	11.2
	55%	12.7	143	11.2	147	11.6	152	11.9
8000	85%	18.3	174	9.5	180	9.8	186	10.2
	75%	16.4	166	10.1	171	10.4	177	10.8
	65%	14.6	156	10.7	161	11.1	166	11.4
	55%	12.7	145	11.4	150	11.8	154	12.1
10000	85%	18.3	177	9.7	184	10.0	190	10.4
	75%	16.4	169	10.3	175	10.6	180	11.0
	65%	14.6	159	10.9	164	11.3	169	11.6
	55%	12.7	148	11.6	152	12.0	157	12.3

CRUISE PERFORMANCE			ISA -30°C		ISA		ISA +30°C	
Altitude (ft MSL)	Power (% of 315)	FF (GPH)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)
12000	85%	18.3	181	9.9	187	10.2	193	10.6
	75%	16.4	172	10.4	178	10.8	183	11.2
	65%	14.6	162	11.1	167	11.5	172	11.8
	55%	12.7	150	11.8	155	12.2	159	12.5
14000	85%	18.3	184	10.0	191	10.4	197	10.8
	75%	16.4	175	10.6	181	11.0	187	11.4
	65%	14.6	165	11.3	170	11.7	175	12.0
	55%	12.7	153	12.0	157	12.4	162	12.7
16000	85%	18.3	187	10.2	194	10.6	201	11.0
	75%	16.4	178	10.8	185	11.2	191	11.6
	65%	14.6	167	11.5	173	11.9	179	12.2
	55%	12.7	155	12.2	160	12.6	164	12.9
18000	85%	18.3	191	10.4	198	10.8	205	11.0
	75%	16.4	181	11.0	188	11.4	194	11.8
	65%	14.6	171	11.7	176	12.1	182	12.5
	55%	12.7	158	12.4	162	12.8	167	13.1
20000	85%	18.3	195	10.6	202	11.0	209	11.4
	80%	17.4	190	10.9	197	11.3	204	11.7
	75%	16.4	185	11.2	192	11.7	198	12.0
	65%	14.6	174	11.9	180	12.3	185	12.7
	55%	12.7	160	12.6	165	13.0	169	13.3
22000	85%	18.3	199	10.8	206	11.3	213	11.6
	80%	17.4	194	11.1	201	11.6	208	12.0
	75%	16.4	188	11.5	195	11.9	202	12.3
	65%	14.6	177	12.1	183	12.5	188	12.9
	55%	12.7	163	12.8	168	13.2	172	13.5
24000	85%	18.3	202	11.1	210	11.5	218	11.9
	80%	17.4	197	11.4	205	11.8	212	12.2
	75%	16.4	192	11.7	199	12.1	206	12.5
	65%	14.6	180	12.3	186	12.8	191	13.1
	55%	12.7	165	13.0	170	13.4	174	13.7



CRUISE PERFORMANCE			ISA -30°C		ISA		ISA +30°C	
Altitude (ft MSL)	Power (% of 315)	FF (GPH)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)	TAS (KTAS)	Econ (nm/gal)
25000	85%	18.3	204	11.2	213	11.6	220	12.0
	80%	17.4	199	11.5	207	11.9	214	12.3
	75%	16.4	194	11.8	201	12.2	208	12.6
	65%	14.6	181	12.4	188	12.9	193	13.2
	55%	12.7	166	13.1	171	13.5	176	13.8

## Range / Endurance: Full Power Climb

### Conditions:

- Mixture ..... Best Economy - Target Fuel Flow or less
- Weight ..... 3600 LB for Climb, Avg 3400 LB for Cruise
- Winds ..... Zero
- Total Fuel ..... 92 Gallons

### • Note •

Fuel Remaining for Cruise is equal to 92.0 gallons usable, less 1.5 gallons (pre-takeoff fuel consumed), 11 gallons (45 minute IFR reserve at 65% power), and listed volume for fuel consumed in Full Power Climb.

Range is decreased by 5% if nose wheel pant and fairings removed.

Range is decreased by 15% of nose wheel and main wheel pants and fairings removed.

For aircraft with optional Air Conditioning System: range is decreased by 1% if system in operation.

Aircraft with optional Enhanced Vision System: range is decreased by ½%.

Range / Endurance: 85% Power Cruise - Full Power Climb							
Press Alt  FT	Climb Fuel  Gal	Fuel Remaining For Cruise Gal	Airspee d  KTAS	Fuel Flow  GPH	Endurance  Hours	Range  NM	Specific Range  Nm/Gal
2000	1.0	78.5	170	18.3	4.3	734	9.3
4000	2.1	77.5	174	18.3	4.2	742	9.5
6000	3.2	76.4	177	18.3	4.2	749	9.7
8000	4.3	75.3	180	18.3	4.1	756	9.8
10000	5.4	74.2	184	18.3	4.0	763	10.0
12000	6.5	73.0	187	18.3	4.0	770	10.2
14000	7.7	71.9	191	18.3	3.9	777	10.4
16000	8.9	70.7	194	18.3	3.9	784	10.6
18000	10.1	69.4	198	18.3	3.8	791	10.8
20000	11.3	68.2	202	18.3	3.7	798	11.0
22000	12.6	66.9	206	18.3	3.7	805	11.3
24000	14.2	65.4	210	18.3	3.6	810	11.5
25000	15.1	64.4	213	18.3	3.5	812	11.6

Range / Endurance: 75% Power Cruise - Full Power Climb							
Press Alt  FT	Climb Fuel  Gal	Fuel Remaining For Cruise Gal	Airspee d  KTAS	Fuel Flow  GPH	Endurance  Hours	Range  NM	Specific Range  Nm/Gal
2000	1.0	78.5	162	16.4	4.8	779	9.9
4000	2.1	77.5	165	16.4	4.7	786	10.1
6000	3.2	76.4	168	16.4	4.6	793	10.2
8000	4.3	75.3	171	16.4	4.6	800	10.4
10000	5.4	74.2	175	16.4	4.5	807	10.6
12000	6.5	73.0	178	16.4	4.4	814	10.8
14000	7.7	71.9	181	16.4	4.4	821	11.0
16000	8.9	70.7	185	16.4	4.3	827	11.2
18000	10.1	69.4	188	16.4	4.2	834	11.4
20000	11.3	68.2	192	16.4	4.2	841	11.7
22000	12.6	66.9	195	16.4	4.1	847	11.9
24000	14.2	65.4	199	16.4	4.0	852	12.1
25000	15.1	64.4	201	16.4	3.9	853	12.2

Range / Endurance: 65% Power Cruise - Full Power Climb							
Press Alt  FT	Climb Fuel  Gal	Fuel Remaining For Cruise Gal	Airspee d  KTAS	Fuel Flow  GPH	Endurance  Hours	Range  NM	Specific Range  Nm/Gal
2000	1.0	78.5	153	14.6	5.4	828	10.5
4000	2.1	77.5	156	14.6	5.3	835	10.7
6000	3.2	76.4	159	14.6	5.2	842	10.9
8000	4.3	75.3	161	14.6	5.2	848	11.1
10000	5.4	74.2	164	14.6	5.1	855	11.3
12000	6.5	73.0	167	14.6	5.0	861	11.5
14000	7.7	71.9	170	14.6	4.9	868	11.7
16000	8.9	70.7	173	14.6	4.8	874	11.9
18000	10.1	69.4	176	14.6	4.8	879	12.1
20000	11.3	68.2	180	14.6	4.7	885	12.3
22000	12.6	66.9	183	14.6	4.6	890	12.5

Range / Endurance: 65% Power Cruise - Full Power Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise	Airspee d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
24000	14.2	65.4	186	14.6	4.5	893	12.8
25000	15.1	64.4	188	14.6	4.4	894	12.9

Range / Endurance: 55% Power Cruise - Full Power Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise	Airspee d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
2000	1.0	78.5	143	12.7	6.2	884	11.2
4000	2.1	77.5	145	12.7	6.1	890	11.4
6000	3.2	76.4	147	12.7	6.0	896	11.6
8000	4.3	75.3	150	12.7	5.9	902	11.8
10000	5.4	74.2	152	12.7	5.8	908	12.0
12000	6.5	73.0	155	12.7	5.7	913	12.2
14000	7.7	71.9	157	12.7	5.6	918	12.4
16000	8.9	70.7	160	12.7	5.6	922	12.6
18000	10.1	69.4	162	12.7	5.5	926	12.8
20000	11.3	68.2	165	12.7	5.4	930	13.0
22000	12.6	66.9	168	12.7	5.3	933	13.2
24000	14.2	65.4	170	12.7	5.1	934	13.4
25000	15.1	64.4	171	12.7	5.1	932	13.5

## Range / Endurance: Cruise Climb

### Conditions:

- Mixture ..... Best Economy - Target Fuel Flow or less
- Weight ..... 3600 LB for Climb, Avg 3400 LB for Cruise
- Winds ..... Zero
- Total Fuel.....92 Gallons

### • Note •

Fuel Remaining for Cruise in this table is based on AFM Cruise Climb: Lean of Peak Technique; if Full Power Climb: Rich of Peak Technique is performed, use Range/Endurance: Full Power Climb tables.

Fuel Remaining for Cruise is equal to 92.0 gallons usable, less 1.5 gallons (pre-takeoff fuel consumed), 11 gallons (45 minute IFR reserve at 65% power), and listed volume for fuel consumed in Full Power Climb.

Range is decreased by 5% if nose wheel pant and fairings removed.

Range is decreased by 15% of nose wheel and main wheel pants and fairings removed.

For aircraft with optional air conditioning System: range is decreased by 1% if system in operation.

Aircraft with optional Enhanced Vision System: range is decreased by ½%.

Range / Endurance: 85% Power Cruise - Cruise Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise	Airspe d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
2000	0.7	78.9	170	18.3	4.3	739	9.3
4000	1.3	78.2	174	18.3	4.3	751	9.5
6000	2.0	77.5	177	18.3	4.2	763	9.7
8000	2.7	76.8	180	18.3	4.2	775	9.8
10000	3.5	76.1	184	18.3	4.2	788	10.0
12000	4.2	75.3	187	18.3	4.1	801	10.2
14000	5.0	74.6	191	18.3	4.1	813	10.4
16000	5.8	73.8	194	18.3	4.0	827	10.6
18000	6.6	73.0	198	18.3	4.0	840	10.8
20000	7.4	72.2	202	18.3	3.9	854	11.0
22000	8.1	71.4	206	18.3	3.9	869	11.3
24000	9.1	70.5	210	18.3	3.8	883	11.5

Range / Endurance: 85% Power Cruise - Cruise Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise Gal	Airspee d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
25000	9.6	69.9	213	18.3	3.8	890	11.6

Range / Endurance: 75% Power Cruise - Cruise Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise Gal	Airspee d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
2000	0.7	78.9	162	16.4	4.8	784	9.9
4000	1.3	78.2	165	16.4	4.8	796	10.1
6000	2.0	77.5	168	16.4	4.7	808	10.2
8000	2.7	76.8	171	16.4	4.7	820	10.4
10000	3.5	76.1	175	16.4	4.6	833	10.6
12000	4.2	75.3	178	16.4	4.6	845	10.8
14000	5.0	74.6	181	16.4	4.5	859	11.0
16000	5.8	73.8	185	16.4	4.5	872	11.2
18000	6.6	73.0	188	16.4	4.4	885	11.4
20000	7.4	72.2	192	16.4	4.4	899	11.7
22000	8.1	71.4	195	16.4	4.3	913	11.9
24000	9.1	70.5	199	16.4	4.3	927	12.1
25000	9.6	69.9	201	16.4	4.3	934	12.2

Range / Endurance: 65% Power Cruise - Cruise Climb							
Press Alt	Climb Fuel	Fuel Remaining For Cruise Gal	Airspee d	Fuel Flow	Endurance	Range	Specific Range
FT	Gal	Gal	KTAS	GPH	Hours	NM	Nm/Gal
2000	0.7	78.9	153	14.6	5.4	833	10.5
4000	1.3	78.2	156	14.6	5.4	845	10.7
6000	2.0	77.5	159	14.6	5.3	857	10.9
8000	2.7	76.8	161	14.6	5.3	869	11.1

Range / Endurance: 65% Power Cruise - Cruise Climb							
Press Alt  FT	Climb Fuel  Gal	Fuel Remaining For Cruise Gal	Airspeed  KTAS	Fuel Flow  GPH	Endurance  Hours	Range  NM	Specific Range  Nm/Gal
10000	3.5	76.1	164	14.6	5.2	882	11.3
12000	4.2	75.3	167	14.6	5.2	894	11.5
14000	5.0	74.6	170	14.6	5.1	907	11.7
16000	5.8	73.8	173	14.6	5.1	920	11.9
18000	6.6	73.0	176	14.6	5.0	933	12.1
20000	7.4	72.2	180	14.6	4.9	946	12.3
22000	8.1	71.4	183	14.6	4.9	959	12.5
24000	9.1	70.5	186	14.6	4.8	972	12.8
25000	9.6	69.9	188	14.6	4.8	978	12.9

Range / Endurance: 55% Power Cruise - Cruise Climb							
Press Alt  FT	Climb Fuel  Gal	Fuel Remaining For Cruise Gal	Airspeed  KTAS	Fuel Flow  GPH	Endurance  Hours	Range  NM	Specific Range  Nm/Gal
2000	0.7	78.9	143	12.7	6.2	889	11.2
4000	1.3	78.2	145	12.7	6.1	900	11.4
6000	2.0	77.5	147	12.7	6.1	912	11.6
8000	2.7	76.8	150	12.7	6.0	924	11.8
10000	3.5	76.1	152	12.7	6.0	936	12.0
12000	4.2	75.3	155	12.7	5.9	948	12.2
14000	5.0	74.6	157	12.7	5.9	959	12.4
16000	5.8	73.8	160	12.7	5.8	971	12.6
18000	6.6	73.0	162	12.7	5.7	982	12.8
20000	7.4	72.2	165	12.7	5.7	994	13.0
22000	8.1	71.4	168	12.7	5.6	1005	13.2
24000	9.1	70.5	170	12.7	5.5	1016	13.4
25000	9.6	69.9	171	12.7	5.5	1021	13.5

## Balked Landing Climb Gradient

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Full Rich
- Flaps ..... 100% (DN)
- Climb Airspeed .....  $V_{REF}$

### • Note •

Climb Gradient values shown are the gain in altitude for the horizontal distance traversed expressed as Feet per Nautical Mile.

For operation in air colder than this table provides, use the coldest (left-most) data shown.

For operation in air warmer than this table provides, use caution.

Weight  LB	Press Alt  FT	Climb Speed ( $V_{REF}$ ) KIAS	CLIMB GRADIENT ~ Feet/Nautical Mile					
			Temperature ~°C					
			-20	0	20	40	50	ISA
3600	SL	79	1111	921	751	596	524	792
	2000	79	1015	835	674	528	459	744
	4000	79	924	754	602	463	398	699
	6000	79	838	678	534	402	341	655
	8000	79	758	606	470	346	287	613
	10000	79	682	539	410	293	237	573
2900	SL	79	1519	1274	1057	861	771	1109
	2000	79	1394	1164	959	775	689	1049
	4000	79	1277	1061	868	694	612	991
	6000	79	1168	965	783	618	541	936
	8000	79	1066	874	703	547	474	883
	10000	79	970	790	628	481	412	832



## Balked Landing Rate of Climb

### Conditions:

- Power ..... Full Throttle
- Mixture ..... Full Rich
- Flaps ..... 100% (DN)
- Climb Airspeed .....  $V_{REF}$

### • Note •

Rate-of-Climb values shown are change in altitude in ft per unit time expressed in Feet per Minute

For operation in air colder than this table provides, use the coldest (life-times) data shown.

For operation in air warmer than this table provides, use caution.

Weight  LB	Press Alt  FT	Climb Speed ( $V_{REF}$ ) KIAS	RATE OF CLIMB - Feet per Minute					
			Temperature ~°C					
			-20	0	20	40	50	ISA
3600	SL	79	1344	1163	986	811	725	1030
	2000	79	1276	1096	919	745	659	998
	4000	79	1208	1028	852	679	594	966
	6000	79	1140	961	786	613	528	934
	8000	79	1072	894	719	548	463	903
	10000	79	1005	827	653	482	397	871
2900	SL	79	1812	1592	1377	1166	1062	1431
	2000	79	1732	1514	1300	1090	986	1396
	4000	79	1653	1436	1224	1014	911	1361
	6000	79	1575	1359	1147	939	836	1327
	8000	79	1497	1282	1072	864	762	1293
	10000	79	1420	1206	996	790	688	1259

# Landing Distance

**Conditions:**

- Winds.....Zero
- Runway..... Dry, Level, Paved
- Flaps..... 100%, 50% or 0%
- Power ..... 3° Power Approach  
to 50 FT obstacle, then reduce power passing the estimated 50 foot point and  
smoothly continue power reduction to reach idle just prior to touchdown.

• Note •

The following factors are to be applied to the computed landing distance for the noted condition:

Headwind - Subtract 10% from table distances for each 13 knots headwind.

Tailwind - Add 10% to table distances for each 2 knots tailwind up to 10 knots.

Grass Runway, Dry - Add 20% to ground roll distance.

Grass Runway, Wet - Add 60% to ground roll distance.

Sloped Runway - Increase table distances by 27% of the ground roll distance for each 1% of downslope. Decrease table distances by 9% of the ground roll distance for each 1% of upslope.

• Note •

The above corrections for runway slope are required to be included herein. These corrections should be used with caution since published runway slope data is usually the net slope from one end of the runway to the other. Many runways will have portions of their length at greater or lesser slopes than the published slope, lengthening (or shortening) landing ground roll estimated from the table.

For operation in outside air temperatures colder than this table provides, use coldest data shown.

For operation in outside air temperatures warmer than this table provides, use caution.

## Landing Distance - Flaps 100%

<b>WEIGHT:</b> 3600 LB <b>Speed over 50 Ft Obstacle:</b> 79 KIAS <b>Flaps:</b> 100% <b>Power:</b> Idle <b>Runway:</b> Dry, Paved, Level				<b>Headwind:</b> Subtract 10% for each 13 knots headwind. <b>Tailwind:</b> Add 10% for each 2 knots tailwind up to 10 knots. <b>Runway Slope:</b> Reference Notes <b>Dry Grass:</b> Add 20% to Ground Roll <b>Wet Grass:</b> Add 60% to Ground Roll				
PRESS ALT FT	DISTANCE  FT	TEMPERATURE ~°C						ISA
		0	10	20	30	40	50	
<b>SL</b>	<b>Grnd Roll</b>	1117	1158	1198	1239	1280	1321	1178
	<b>Total</b>	2447	2505	2565	2625	2685	2747	2535
<b>1000</b>	<b>Grnd Roll</b>	1158	1200	1243	1285	1327	1370	1213
	<b>Total</b>	2506	2567	2630	2693	2757	2821	2585
<b>2000</b>	<b>Grnd Roll</b>	1201	1245	1289	1333	1377	1421	1250
	<b>Total</b>	2568	2633	2699	2765	2832	2900	2636
<b>3000</b>	<b>Grnd Roll</b>	1246	1292	1337	1383	1428	1474	1287
	<b>Total</b>	2635	2702	2771	2841	2911	2983	2691
<b>4000</b>	<b>Grnd Roll</b>	1293	1340	1388	1435	1482	1530	1326
	<b>Total</b>	2705	2776	2848	2922	2996	3070	2748
<b>5000</b>	<b>Grnd Roll</b>	1342	1391	1440	1489	1539	1588	1367
	<b>Total</b>	2779	2854	2930	3007	3085	3163	2808
<b>6000</b>	<b>Grnd Roll</b>	1393	1444	1495	1546	1598	1649	1409
	<b>Total</b>	2857	2936	3016	3097	3179	3261	2871
<b>7000</b>	<b>Grnd Roll</b>	1447	1500	1553	1606	1659	1712	1453
	<b>Total</b>	2941	3024	3108	3193	3279	3365	2937
<b>8000</b>	<b>Grnd Roll</b>	1503	1558	1613	1668	1724	1779	1499
	<b>Total</b>	3029	3116	3205	3294	3384	3475	3006
<b>9000</b>	<b>Grnd Roll</b>	1562	1619	1677	1734	1791	1848	1546
	<b>Total</b>	3122	3214	3307	3401	3496	3592	3079
<b>10000</b>	<b>Grnd Roll</b>	1624	1683	1743	1802	1862	1921	1595
	<b>Total</b>	3221	3318	3416	3515	3614	3715	3155

# Landing Distance - Flaps 50%

<b>WEIGHT:</b> 3600 LB <b>Speed over 50 Ft Obstacle:</b> 87 KIAS <b>Flaps:</b> 50% <b>Power:</b> Idle <b>Runway:</b> Dry, Paved, Level		<b>Headwind:</b> Subtract 10% for each 13 knots headwind. <b>Tailwind:</b> Add 10% for each 2 knots tailwind up to 10 knots. <b>Runway Slope:</b> Reference Notes <b>Dry Grass:</b> Add 20% to Ground Roll <b>Wet Grass:</b> Add 60% to Ground Roll						
PRESS ALT FT	DISTANCE  FT	TEMPERATURE ~°C						ISA
		0	10	20	30	40	50	
SL	Grnd Roll	1166	1209	1251	1294	1337	1379	1230
	Total	2681	2745	2810	2875	2942	3010	2777
1000	Grnd Roll	1209	1253	1298	1342	1386	1430	1267
	Total	2745	2813	2881	2950	3020	3091	2833
2000	Grnd Roll	1254	1300	1346	1392	1438	1484	1305
	Total	2814	2885	2957	3029	3103	3178	2892
3000	Grnd Roll	1301	1349	1396	1444	1491	1539	1344
	Total	2886	2961	3037	3113	3191	3269	2954
4000	Grnd Roll	1350	1399	1449	1498	1548	1597	1385
	Total	2963	3042	3121	3202	3283	3366	3019
5000	Grnd Roll	1401	1453	1504	1555	1607	1658	1427
	Total	3045	3127	3211	3296	3382	3468	3087
6000	Grnd Roll	1455	1508	1561	1615	1668	1721	1472
	Total	3131	3218	3306	3395	3485	3576	3158
7000	Grnd Roll	1511	1566	1622	1677	1732	1788	1517
	Total	3223	3314	3407	3501	3595	3691	3233
8000	Grnd Roll	1570	1627	1685	1742	1800	1857	1565
	Total	3320	3416	3514	3612	3712	3812	3312
9000	Grnd Roll	1631	1691	1751	1810	1870	1930	1614
	Total	3423	3524	3627	3731	3835	3941	3395
10000	Grnd Roll	1695	1758	1820	1882	1944	2006	1666
	Total	3532	3639	3747	3856	3966	4077	3481

## Landing Distance - Flaps 0%

<b>WEIGHT:</b> 3600 LB <b>Speed over 50 Ft Obstacle:</b> 94 KIAS <b>Flaps:</b> 0% <b>Power:</b> Idle <b>Runway:</b> Dry, Paved, Level				<b>Headwind:</b> Subtract 10% for each 13 knots headwind. <b>Tailwind:</b> Add 10% for each 2 knots tailwind up to 10 knots. <b>Runway Slope:</b> Reference Notes <b>Dry Grass:</b> Add 20% to Ground Roll <b>Wet Grass:</b> Add 60% to Ground Roll				
PRESS ALT FT	DISTANCE  FT	TEMPERATURE ~°C						ISA
		0	10	20	30	40	50	
SL	Grnd Roll	1365	1415	1465	1515	1565	1615	1440
	Total	3165	3241	3319	3398	3478	3558	3280
1000	Grnd Roll	1415	1467	1519	1571	1623	1675	1483
	Total	3242	3323	3404	3487	3571	3656	3347
2000	Grnd Roll	1468	1522	1576	1629	1683	1737	1527
	Total	3324	3409	3495	3582	3670	3759	3418
3000	Grnd Roll	1523	1579	1635	1690	1746	1802	1574
	Total	3411	3500	3590	3682	3775	3868	3491
4000	Grnd Roll	1581	1638	1696	1754	1812	1870	1621
	Total	3503	3597	3692	3788	3885	3984	3569
5000	Grnd Roll	1641	1701	1761	1821	1881	1941	1671
	Total	3600	3699	3799	3900	4003	4106	3650
6000	Grnd Roll	1703	1766	1828	1890	1953	2015	1723
	Total	3703	3807	3913	4019	4127	4236	3736
7000	Grnd Roll	1769	1834	1899	1963	2028	2093	1776
	Total	3813	3922	4033	4145	4258	4373	3825
8000	Grnd Roll	1838	1905	1972	2040	2107	2174	1832
	Total	3929	4044	4161	4279	4398	4518	3919
9000	Grnd Roll	1910	1980	2049	2119	2189	2259	1890
	Total	4052	4173	4296	4420	4545	4671	4018
10000	Grnd Roll	1985	2058	2130	2203	2276	2348	1950
	Total	4183	4310	4439	4569	4701	4833	4122

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## ***Section 6: Weight and Balance Data***

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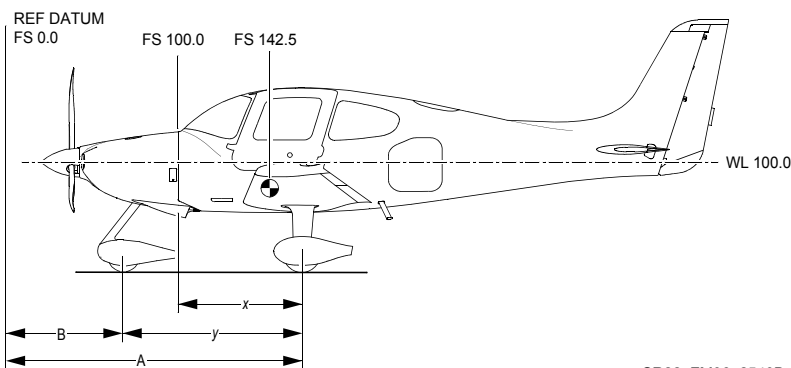


## Introduction

This section describes the procedure for calculating the weight and moment for various operations. A comprehensive list of all equipment available for this airplane is included at the back of this section.

It should be noted that specific information regarding the weight, arm, moment, and installed equipment for this airplane as delivered from the factory can be found at the back of this section.

It is the responsibility of the pilot to ensure that the airplane is loaded properly. All changes to the basic empty weight and center of gravity are the responsibility of the operator.



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Basic empty weight, moment, and center of gravity are provided in inches aft of datum, where 0 inches datum is 100.0 inches forward of the cabin firewall. CG can also be expressed in terms of its location as a percentage of the airplane Mean Aerodynamic Cord (MAC) using the following formula:

$$CG\% \text{ MAC} = 100 \times (CG \text{ Inches} - LEMAC) / MAC$$

Where:

$$LEMAC = 133.1$$

$$MAC = 47.7$$

• Note •

Leveling and Weighing procedures are not described in this section. Refer to Airplane Maintenance Manual (AMM), Chapter 8, Leveling and Weighing.

## Loading Instructions

It is the responsibility of the pilot to ensure that the airplane is properly loaded and operated within the prescribed weight and center of gravity limits. The following information enables the pilot to calculate the total weight and moment for the loading. The calculated moment is then compared to the Moment Limits chart or table (*Figure 6-3*) for a determination of proper loading.

Airplane loading determinations are calculated using the Weight & Balance Loading Form (*Figure 6-1*), the Loading Data chart and table (*Figure 6-2*), and the Moment Limits chart and table (*Figure 6-3*).

1. **Basic Empty Weight** – Enter the current Basic Empty Weight and Moment from the Weight & Balance Record (*Figure 6-4*).
2. **Front Seat Occupants** – Enter the total weight and moment/1000 for the front seat occupants from the Loading Data (*Figure 6-2*).
3. **Rear Seat Occupants** – Enter the total weight and moment/1000 for the rear seat occupants from the Loading Data (*Figure 6-2*).
4. **Baggage** – Enter weight and moment for the baggage from the Loading Data (*Figure 6-2*).
  - If desired, subtotal the weights and moment/1000 from steps 1 through 4. This is the *Zero Fuel Condition*. It includes all useful load items excluding fuel.
5. **Fuel Loading** – Enter the weight and moment of usable fuel loaded on the airplane from the Loading Data (*Figure 6-2*).
  - Subtotal the weight and moment/1000. This is the *Ramp Condition* or the weight and moment of the aircraft before taxi.
6. **Fuel for start, taxi, and run-up** – This value is pre-entered on the form. Normally, fuel used for start, taxi, and run-up is approximately 9 pounds at an average moment/1000 of 1.394.
7. **Takeoff Condition** – Subtract the weight and moment/1000 for step 8 (start, taxi, and run-up) from the Ramp Condition values (step 7) to determine the Takeoff Condition weight and moment/1000.
  - The total weight at takeoff must not exceed the maximum weight limit of 3600 pounds. The total moment/1000 must not be above the maximum or below the minimum moment/1000 for the Takeoff Condition Weight as determined from the Moment Limits chart or table (*Figure 6-3*).

## Weight and Balance Loading Form

• Note •

The Takeoff Condition Weight must not exceed 3600 lb.

The Takeoff Condition Moment must be within the Minimum Moment to Maximum Moment range at the Takeoff Condition Weight. (Refer to *Moment Limits*).

Serial Num: \_\_\_\_\_ Date: \_\_\_\_\_

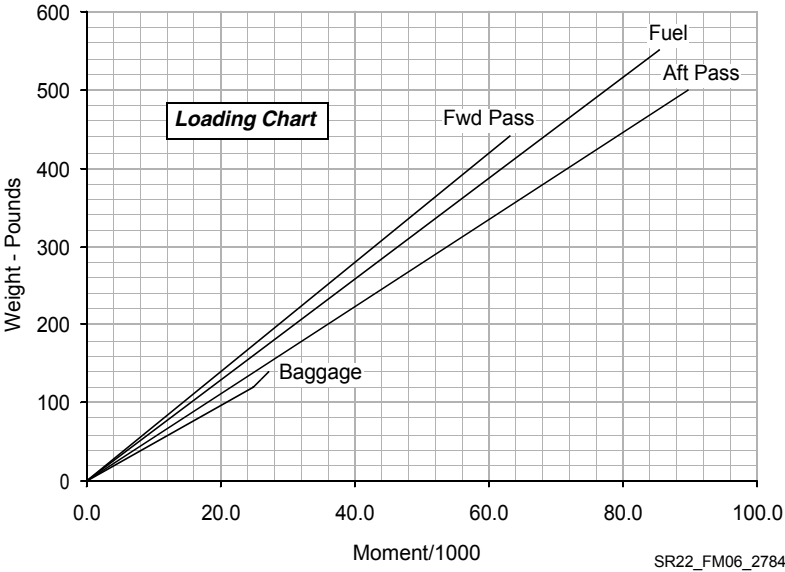
Reg. Num: \_\_\_\_\_ Initials: \_\_\_\_\_

Item	Description	Weight LB	Moment/ 1000
1.	Basic Empty Weight <i>Includes unusable fuel &amp; full oil</i>		
2.	Front Seat Occupants <i>Pilot &amp; Passenger (total)</i>		
3.	Rear Seat Occupants		
4.	Baggage Area <i>130 lb maximum</i>		
5.	Zero Fuel Condition Weight <i>Sub total item 1 thru 4 3400 lb maximum</i>		
6.	Fuel Loading <i>92 Gallon @ 6.0 lb/gal. Maximum</i>		
7.	Ramp Condition Weight <i>Sub total item 5 and 6</i>		
8.	Fuel for start, taxi, and run-up (negative number) <i>Normally 9 lb at average moment of 1394 (1.4).</i>		
9.	Takeoff Condition Weight <i>Subtract item 8 from item 7</i>		

Figure 6-1

Loading Data

Use the following chart or table to determine the moment/1000 for fuel and payload items to complete the Loading Form.



Weight LB	Fwd Pass FS 143.5	Aft Pass FS 180.0	Baggage FS 208.0	Fuel FS 154.9	Weight LB	Fwd Pass FS 143.5	Aft Pass FS 180.0	Fuel FS 154.9
20	2.9	3.6	4.2	3.1	300	43.1	54.0	46.5
40	5.7	7.2	8.3	6.2	320	45.9	57.6	49.6
60	8.6	10.8	12.5	9.3	340	48.8	61.2	52.7
80	11.5	14.4	16.6	12.4	360	51.7	64.8	55.8
100	14.4	18.0	20.8	15.5	380	54.5	68.4	58.9
120	17.2	21.6	25.0	18.6	400	57.4	72.0	62.0
140	20.1	25.2	27.04*	21.7	420	60.3	75.6	65.1
160	23.0	28.8		24.8	440	63.1	79.2	68.2
180	25.8	32.4		27.9	460		82.8	71.3
200	28.7	36.0		31.0	480		86.4	74.4
220	31.6	39.6		34.1	500		90.0	77.5
240	34.4	43.2		37.2	520			80.5
260	37.3	46.8		40.3	552**			85.5
280	40.2	50.4		43.4				

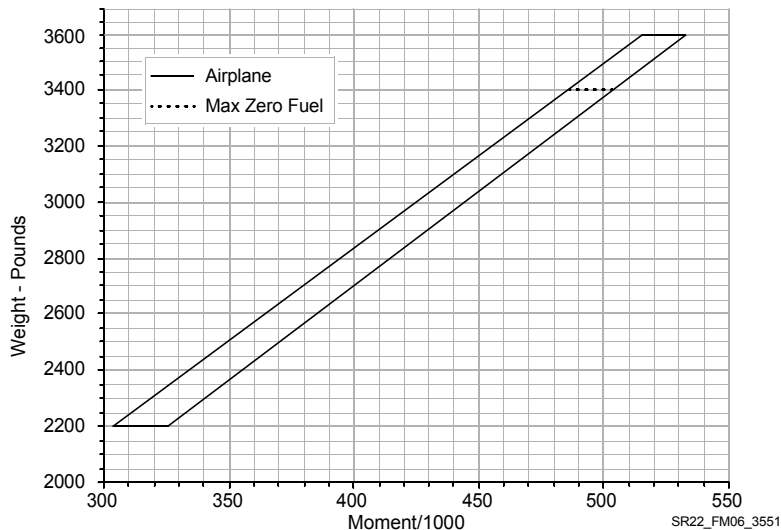
\*130 lb Maximum

\*\*92 U.S. Gallons Usable

Figure 6-2

Moment Limits

Use the following chart or table to determine if the weight and moment from the completed Weight and Balance Loading Form (Figure 6-1) are within limits.



Weight			Moment/1000		
LB	Minimum	Maximum	LB	Minimum	Maximum
2200	304	326	2950	414	437
2250	311	333	3000	422	444
2300	318	341	3050	430	452
2350	325	348	3100	438	459
2400	332	356	3150	445	467
2450	340	363	3200	453	474
2500	347	370	3250	461	481
2550	354	378	3300	469	489
2600	361	385	3350	477	496
2650	368	393	*3400	484	504
2700	375	400	3450	494	511
2750	383	407	3500	501	519
2800	391	415	3550	508	526
2850	399	422	3600	515	533
2900	407	430			

\*NOTE: Maximum zero fuel weight.

Figure 6-3



## Equipment List

This list will be determined after the final equipment has been installed in the aircraft.

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## ***Section 7: Systems Description***

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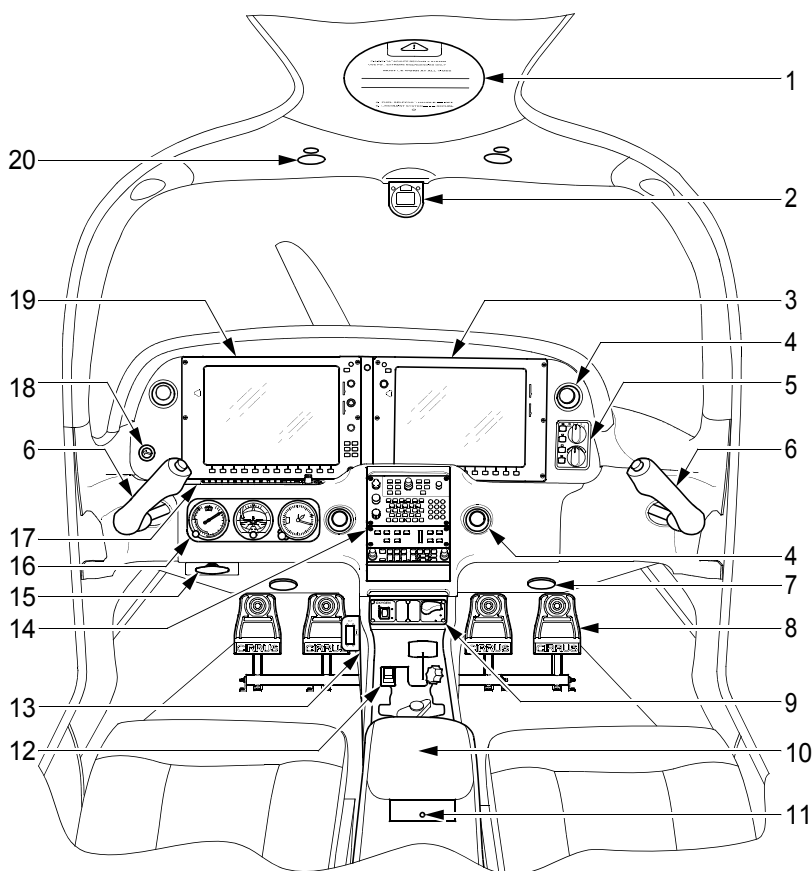
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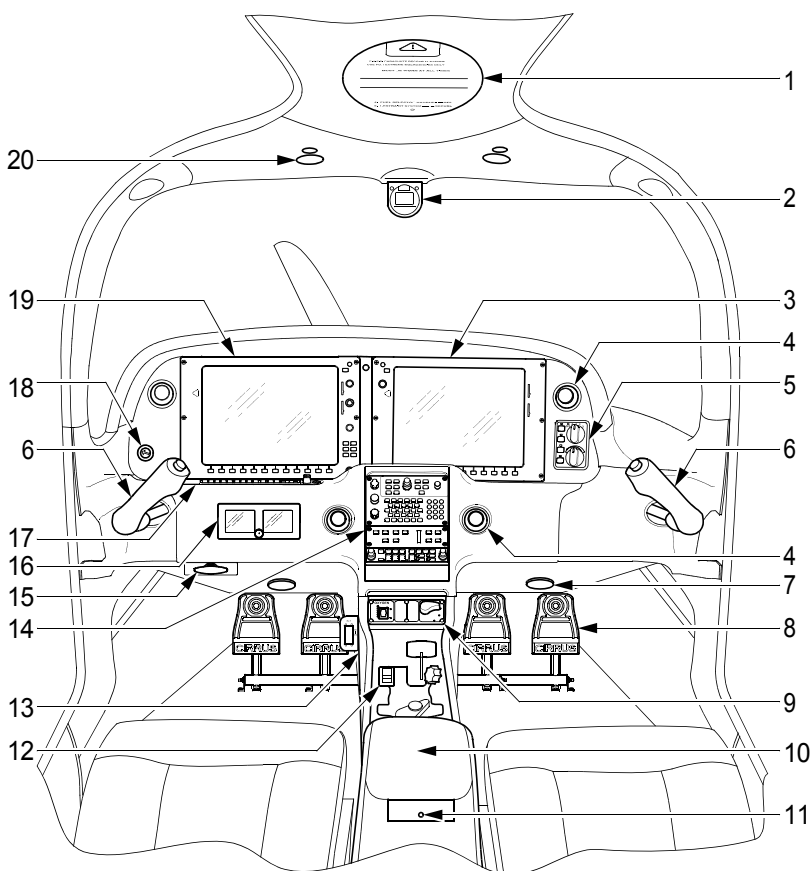


### Legend

- |  |                                   |
|--|-----------------------------------|
| 1. Cirrus Airframe Parachute System (CAPS) Activation T-Handle Cover | 12. Engine & Fuel System Controls |
| 2. Magnetic Compass  | 13. Left Side Console             |
| 3. Multifunction Display   | · Circuit Breaker Panel           |
| 4. Fresh Air "Eyeball" Outlet  | · ELT Remote Switch               |
| 5. Temperature/Ventilation Controls                                  | · Alternate Static Source         |
| 6. Control Yoke  | 14. Avionics Panel                |
| 7. Conditioned Air Outlet  | 15. Parking Brake                 |
| 8. Rudder Pedals   | 16. Flight Instrument Panel       |
| 9. Flap Control & Position Indicators                                | 17. Bolster Switch Panel          |
| 10. Armrest  | 18. Start/Ignition Key Switch     |
| 11. Passenger Audio Jack(s)  | 19. Primary Flight Display        |
|  | 20. Overhead Light & Switch       |

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**Figure 7-4**  
**Instrument Panel and Console - *Serials w/o MD302* (1 of 2)**



**Legend**

- |  |   |
|--|---|
| 1. Cirrus Airframe Parachute System (CAPS) Activation T-Handle Cover | 13. Left Side Console   |
| 2. Magnetic Compass  | <ul style="list-style-type: none"> <li>· Circuit Breaker Panel</li> <li>· Alternate Engine Air</li> <li>· ELT Remote Switch</li> <li>· Alternate Static Source</li> </ul> |
| 3. Multifunction Display   | 14. Avionics Panel  |
| 4. Fresh Air "Eyeball" Outlet  | 15. Parking Brake   |
| 5. Temperature/Ventilation Controls                                  | 16. Flight Instrument Panel   |
| 6. Control Yoke  | 17. Bolster Switch Panel  |
| 7. Conditioned Air Outlet  | 18. Start/Ignition Key Switch   |
| 8. Rudder Pedals   | 19. Primary Flight Display  |
| 9. Flap Control & Position Indicators                                | 20. Overhead Light & Switch   |
| 10. Armrest  |   |
| 11. Passenger Audio Jack(s)  |   |
| 12. Engine & Fuel System Controls                                    |   |

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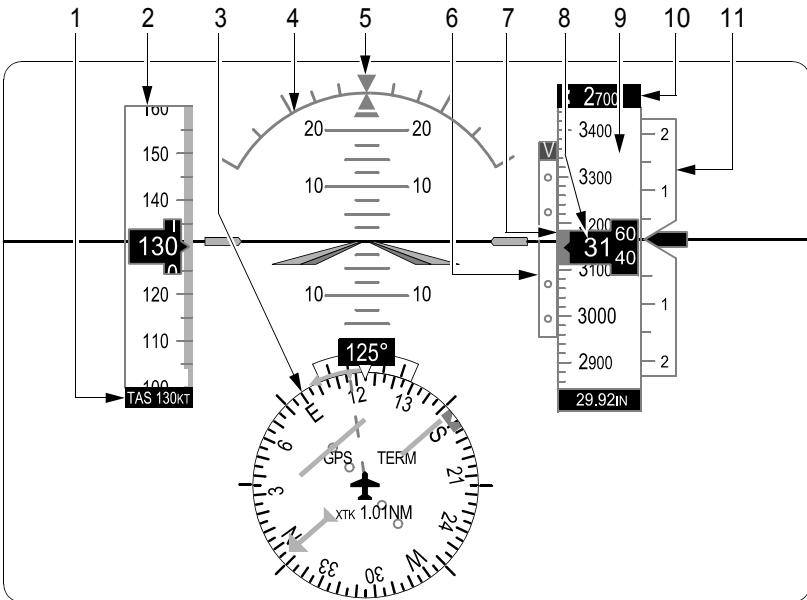
**Figure 7-4**

**Instrument Panel and Console - Serials 0954, 0963 & subs w/ MD302 (2 of 2)**

## Flight Instruments

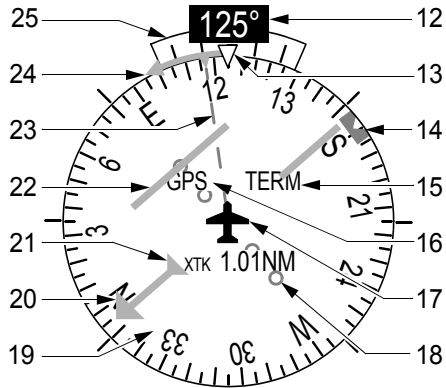
Flight instruments and annunciations are displayed on the Primary Flight Display (PFD) located directly in front of the pilot. The PFD presents the primary flight instruments arranged in the conventional basic “T” configuration. Standby instruments for airspeed, attitude, and altitude are mounted on the LH bolster panel and are powered independently of the PFD.

Knobs, knob sets, and membrane-type push button switches are located along the inboard edge of the PFD and MFD and provide control for communication (COM), navigation (NAV), heading (HDG), barometric pressure set (BARO), and various Flight Management functions. For electrical requirements and additional information on PFD and MFD integration, refer to the [Perspective Integrated Avionics System](#) description in this section.



**LEGEND**

1. True Airspeed
2. Airspeed Indicator
3. Horizontal Situation Indicator (HSI)
4. Attitude Indicator
5. Slip/Skid Indicator
6. Vertical Deviation Indicator (VDI)
7. Selected Altitude Bug
8. Current Altitude
9. Altimeter
10. Selected Altitude
11. Vertical Speed Indicator (VSI)
12. Current Heading
13. Lubber Line
14. Selected Heading Bug
15. Flight Phase
16. Navigation Source
17. Aircraft Symbol
18. Course Deviation Indicator
19. Rotating Compass Rose
20. Course Pointer



**HSI DETAIL**

21. To/From Indicator
22. Course Deviation Indicator
23. Current Track Indicator
24. Turn Rate/Heading Trend Vector
25. Turn Rate Indicator

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**Figure 7-5**  
**Flight Instruments**



## Attitude Indicator

The primary attitude indicator is shown on the upper center of the PFD and displays pitch, roll, and slip/skid information provided by the Attitude and Heading Reference System (AHRS).

Above and below the horizon line, major pitch marks and labels are shown for every 10°, up to 80°. Between 25° below and 45° above the horizon line, the pitch index scale is graduated in 5° increments with every 10° of pitch labeled. Between 20° below and 20° above the horizon line, minor pitch marks occur every 2.5°. If pitch limits are exceeded in either the nose-up or nose-down attitude, red warning chevrons will appear and point the way back to level flight. The roll index scale is graduated with major tick marks at 30° and 60° and minor tick marks at 10°, 20°, and 45°. The roll pointer is slaved to the airplane symbol. The slip-skid indicator is the bar beneath the roll pointer. The indicator moves with the roll pointer and moves laterally away from the pointer to indicate lateral acceleration. Slip/skid is indicated by the location of the bar relative to the pointer. One bar displacement is equal to one ball displacement on a traditional slip/skid indicator.

### ***Standby Attitude Indicator***

*Serials w/o MD302 Standby Attitude Module:*

The standby attitude indicator is mounted on the LH bolster panel and gives backup indication of flight attitude. Bank attitude is indicated by a pointer at the top of the indicator relative to the bank scale with index marks at 10°, 20°, 30°, 60°, and 90° either side of the center mark. A fixed miniature airplane superimposed over a movable mask containing a white symbolic horizon bar, which divides the mask into two sections, indicates pitch and roll attitudes. The upper “blue sky” section and the lower “earth” sections have pitch reference lines useful for pitch attitude control. A knob at the bottom of the instrument allows adjustment of the miniature airplane to the horizon bar for a more accurate flight attitude indication. A PULL TO CAGE knob on the indicator is used for quick erection of the gyro. When the caging knob is pulled, the pitch and roll indications will align to within 2° of their respective fixed references. The standby attitude indicator is electrically driven. A red GYRO flag indicates loss of electrical power. Redundant circuits paralleled through diodes at the indicator supply DC electrical power for gyro operation.

*Serials 0954, 0963 & subs w/ MD302 Standby Attitude Module:*

The MD302 Standby Attitude Module is mounted on the LH bolster panel and gives backup indication of flight attitude. Bank attitude is indicated by a pointer at the top of the indicator relative to the bank scale with index marks at 0° (triangle), 10°, 20°, 30°, 45° (small triangle), and 60° either side of the center mark. A fixed, user-configurable airplane symbol is superimposed over a movable background containing a white horizon bar that divides the attitude display into two sections: upper “blue sky” and lower “earth”. The pitch scale on the attitude display is graduated in 5° increments. Chevrons appear on the pitch scale at extreme pitch attitudes. The MD302 Standby Attitude Module is electrically driven. A red X indicates the attitude display is absent due to exceedance of internal rate sensors, loss of airspeed, or other reasons. Redundant circuits paralleled through diodes supply DC electrical power to the unit.

*All Serials:*

28 VDC for the standby attitude indicator is supplied through the 5-amp STDBY ATTD 1 circuit breaker on the ESS BUS 1 and the 5-amp STDBY ATTD 2 circuit breaker on the MAIN BUS 1.

## **Airspeed Indicator**

Primary airspeed data is provided by the Air Data Computer and is shown as a vertical tape along the upper left side of the PFD. The airspeed scale is graduated with major tick marks at intervals of 10 knots and minor tick marks at intervals of 5 knots. Speed indication starts at 20 knots, with 56 knots of airspeed viewable at any time. The actual airspeed is displayed inside the black pointer. The pointer remains black until reaching the never-exceed speed ( $V_{NE}$ ), at which point it turns red. Color coded bars are provided to indicate flap operating range, normal operating range, caution range, and never-exceed speed. Speeds above the never-exceed speed, appear in the high speed warning range, represented on the airspeed tape by red/white “barber pole” coloration. Calculated true airspeed is displayed in window at the bottom edge of the airspeed tape. Airspeed trend is also displayed as a bar along side of the airspeed tape.

### ***Standby Airspeed Indicator***

*Serials w/o MD302 Standby Attitude Module:*

The standby airspeed indicator is mounted on the LH bolster panel and displays indicated and true airspeeds on a dual-scale, internally lit

precision airspeed indicator installed in the pilot's instrument panel. The instrument senses difference in static and Pitot pressures and displays the result in knots on an airspeed scale. A single pointer sweeps an indicated airspeed scale calibrated from 40 to 220 knots. The 'zero' index is at the 12 o'clock position. A sub-scale aligns true airspeed with the corresponding indicated airspeed when the altitude/temperature correction is set in the correction window. A knob in the lower left corner of the instrument is used to rotate the pressure altitude scale in the correction window to align the current pressure altitude with the outside air temperature.

*Serials 0954, 0963 & subs w/ MD302 Standby Attitude Module:*

The MD302 Standby Attitude Module is mounted on the LH bolster panel and displays the current Indicated Airspeed (IAS). The instrument senses difference in static and pitot pressures and displays the result in knots in the Airspeed Window. The Airspeed Window/Pointer sweeps the indicated airspeed tape and denotes the current airspeed. Color coded bars are provided to indicate flap operating range, normal operating range, caution range, and never-exceed speed. The airspeed indication markings on the MD302 do not automatically compensate for changes in  $V_{NE}$  or  $V_{NO}$  at altitudes above 17,500 feet.

28 VDC for the MD302 Standby Attitude Module is supplied through the 5-amp STDBY ATTD 1 circuit breaker on the ESS BUS 1 and the 5-amp STDBY ATTD 2 circuit breaker on the MAIN BUS 1.

## Altimeter

Primary altitude data is provided by the Air Data Computer and is shown as a vertical tape along the upper right side of the PFD. The altimeter scale is graduated with major tick marks at intervals of 100 feet and minor tick marks at intervals of 20 feet. Six hundred (600) feet of barometric altitude is viewable at any time.

The local barometric pressure is set using the barometric adjustment knob on the PFD. The selectable altitude reference bug is displayed on the altimeter tape and is set using the altitude selection knob on the Flight Management System Keyboard. Barometric minimum descent altitude (MDA, or Decision Height, DH), can be preset. Altimeter trend is also displayed as a bar along side of the altimeter tape.

The PFD Altitude is corrected for static source position error (normal static source / 0% flaps), the altitude calibration errors for the PFD are zero with flaps up and normal source (typical cruise flight). Calibration

corrections are only necessary when flaps are extended or the alternate static source is selected.

### ***Standby Altimeter***

#### ***Serials w/o MD302 Standby Attitude Module:***

Airplane altitude is depicted on a conventional, three-pointer, internally lit barometric altimeter installed on the LH bolster panel. The instrument senses the local barometric pressure adjusted for altimeter setting and displays the result on the instrument in feet. The altimeter is calibrated for operation between -1000 and 20,000 feet altitude. The scale is marked from 0 to 10 in increments of 2. The long pointer indicates hundreds of feet and sweeps the scale every 1000 feet (each increment equals 20 feet). The short, wide pointer indicates thousands of feet and sweeps the scale every 10,000 feet (each increment equals 200 feet). The short narrow pointer indicates tens of thousands feet and sweeps from 0 (zero) to 2 (20,000 feet with each increment equal to 2000 feet). Barometric windows on the instrument's face allow barometric calibrations in either inches of mercury (in.Hg) or millibars (mb). The barometric altimeter settings are input through the barometric adjustment knob at the lower left of the instrument.

#### ***Serials 0954, 0963 & subs w/ MD302 Standby Attitude Module:***

The MD302 Standby Attitude Module is mounted on the LH bolster panel and displays the current barometric corrected altitude. The instrument senses the local barometric pressure adjusted for altimeter setting and displays the result in the Altitude Window. The altitude units are user-configurable in feet or meters. The Altitude Window/Pointer sweeps the altitude tape and denotes the current BARO-corrected altitude. The BARO Window shows the currently selected barometric altitude. The BARO units are user-configurable in IN HG or MBAR. The barometric setting on the MD302 will automatically synchronize to the setting on the Garmin avionics, and can be manually adjusted by turning the Control Knob while in Flight Mode.

28 VDC for the MD302 Standby Attitude Module is supplied through the 5-amp STDBY ATTD 1 circuit breaker on the ESS BUS 1 and the 5-amp STDBY ATTD 2 circuit breaker on the MAIN BUS 1.

#### ***All Serials:***

The standby altimeter does not have automatic position error corrections, calibration corrections are necessary. Because the PFD has automatic corrections and the standby does not, differences between the two indications are typical (difference is the greatest at

high altitudes and high airspeeds, where the position error corrections are the highest).

## **Horizontal Situation Indicator**

The horizontal situation indicator is displayed along the lower center of the PFD. Heading data is provided by the Attitude and Heading Reference System (AHRS) and the onboard magnetometers. The HSI displays a rotating compass card in a heading-up orientation. Letters indicate the cardinal points and numeric labels occur every 30°. Major tick marks are at 10° intervals and minor tick marks at 5° intervals. Reference index marks are provided at 45° intervals around the compass card. A circular segment scale directly above the rotating compass card shows half and standard rates of turn based on the length of the turn rate trend vector.

The HSI presents heading, turn rate, course deviation, bearing, and navigation source information in a 360° compass-rose format. The HSI contains a Course Deviation Indicator (CDI) with a course pointer arrow, a To/From arrow, a sliding deviation bar, and scale. The course pointer is a single line arrow (GPS, VOR1, and LOC1) or a double line arrow (VOR2 and LOC2) which points in the direction of the set course. The To/From arrow rotates with the course pointer and is displayed when the active NAVAID is received.

The HSI heading reference bug is set using the heading selection knob on the Flight Management System Keyboard. The selected heading is displayed in a window above the upper LH 45° index mark and will disappear approximately 3 seconds after the heading selection knob stops turning.

The Course Deviation Indicator (CDI) navigation source shown on the HSI is set using the CDI softkey to select GPS, NAV1, or NAV2 inputs. The course pointer is set using the course selection knob on the Flight Management System Keyboard. The selected course is displayed in a window above the upper RH 45° index mark and will disappear approximately 3 seconds after the heading selection knob stops turning.

## **Vertical Speed Indicator**

Vertical Speed data is provided by the Air Data Computer and is shown as a vertical tape along the right side of the altimeter on the PFD. The VSI scale is graduated with major tick marks at 1000 and 2000 fpm in each direction and minor tick marks at intervals of 500 feet

The vertical speed pointer moves up and down the fixed VSI scale and shows the rate of climb or descent in digits inside the pointer. A reference notch at the RH edge of the scale indicates 0 feet/min. Vertical speed must exceed 100 feet/min before digits will appear in the VSI pointer. If the rate of ascent/descent exceeds 2000 fpm, the pointer appears at the corresponding edge of the tape and the rate appears inside the pointer.

*Serials 0954, 0963 & subs w/ MD302 Standby Attitude Module:*

The Altitude Trend Bar is located along the right margin of the Altitude Display. This feature is optional and can be turned on or off by the user.

## Magnetic Compass

A conventional, internally lighted, liquid filled, magnetic compass is installed on the cabin headliner immediately above the windshield. A compass correction card is installed with the compass.

### • Note •

Refer to FAA Advisory Circular (AC) 43.13-1B for a list of occasions requiring a compass swing. If a compass swing is required, perform Operational Test - Magnetic Compass Calibration (refer to AMM 34-20, Attitude and Direction).

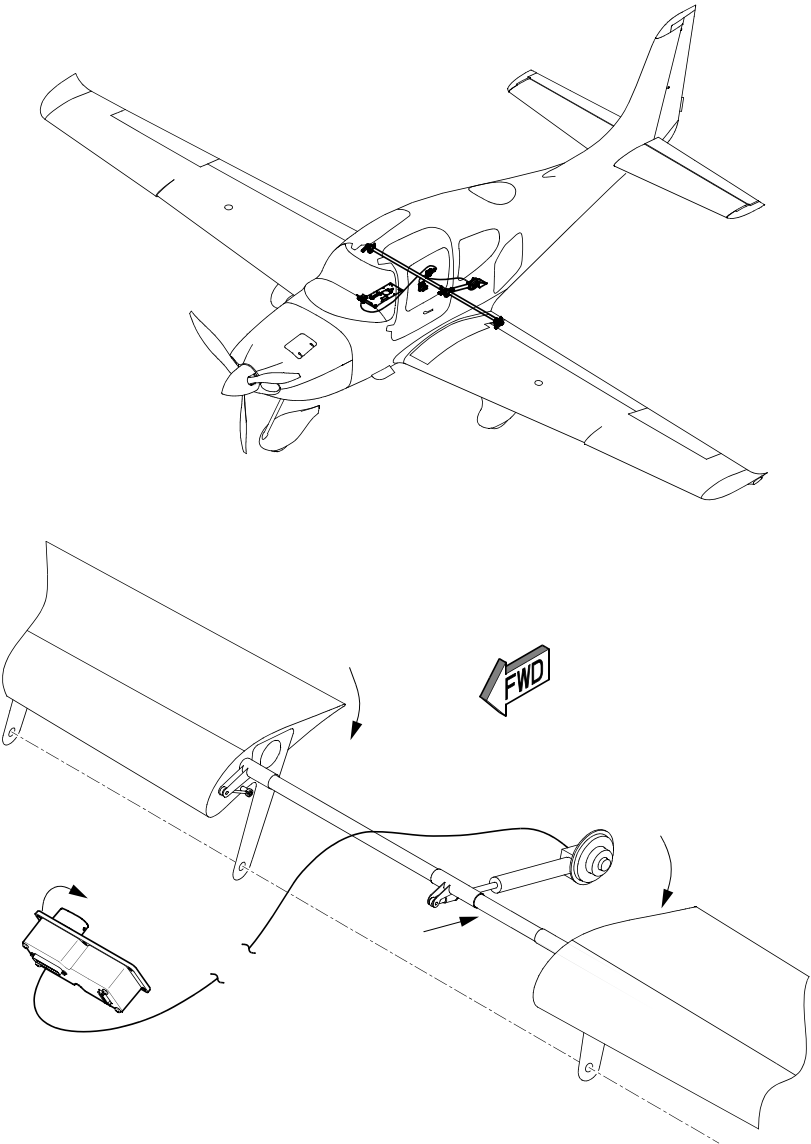
## Wing Flaps

The electrically controlled, single-slotted flaps provide low-speed lift enhancement. Each flap is manufactured of aluminium and connected to the wing structure at three hinge points. Rub strips are installed on the top leading edge of each flap to prevent contact between the flap and wing flap cove. The flaps are selectively set to three positions: 0%, 50% (16°) and 100% (35.5°) by operating the FLAP control switch. The FLAP control switch positions the flaps through a motorized linear actuator mechanically connected to both flaps by a torque tube. Proximity switches in the actuator limit flap travel to the selected position and provide position indication.

The wing flaps actuator, flap control switch and indicator lights are powered by 28 VDC through the 10-amp FLAPS circuit breaker on the NON ESS BUS.

### Flap Control Switch

An airfoil-shaped FLAPS control switch is located at the bottom of the vertical section of the center console. The control switch is marked and has detents at three positions: UP (0%), 50% and 100%. The appropriate  $V_{FE}$  speed is marked at the Flap 50% and 100% switch positions. Setting the switch to the desired position will cause the flaps to extend or retract to the appropriate setting. An indicator light at each control switch position illuminates when the flaps reach the selected position. The UP (0%) light is green and the 50% and 100% lights are yellow.



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**Figure 7-6**  
**Wing Flaps**



## Landing Gear

### Main Gear

The main landing gear are bolted to composite wing structure between the wing spar and shear web. The landing gear struts are constructed of composite material for fatigue resistance. The composite construction is both rugged and maintenance free. The main wheels and wheel pants are bolted to the struts.

Each main gear wheel has a 15 x 6.00 x 6 tire with an inner-tube installed (*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21*), or with a tubeless tire installed (*0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs*).

Standard wheel pants are easily removable to provide access to tires and brakes. Access plugs in the wheel pants can be removed to allow tire inflation and pressure checking. Each main gear wheel is equipped with an independent, hydraulically operated single cylinder, dual piston, disc brake.

### Nose Gear

The nose gear strut is of tubular steel construction and is attached to the steel engine mount structure. Shock absorption is accomplished by an oleo shock absorber. The nosewheel is free casting and can turn through an arc of approximately 170 degrees (85 degrees either side of center). Steering is accomplished by differential application of individual main gear brakes.

Each nosewheel has a 5.00 x 5 tire with an inner-tube installed (*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21*), or with a tubeless tire installed (*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs*).

### Brake System

The main wheels have hydraulically operated, single-disc type brakes, individually activated by floor mounted toe pedals at both pilot stations. A parking brake mechanism holds induced hydraulic pressure on the disc brakes for parking. The brake system consists of a master cylinder for each rudder pedal, a hydraulic fluid reservoir, a parking brake valve, a single disc brake assembly on each main landing gear wheel, temperature sensors, and associated hydraulic plumbing and wiring.

Braking pressure is initiated by depressing the top half of a rudder pedal (toe brake). The brakes are plumbed so that depressing either the pilot's or copilot's left or right toe brake will apply the respective (left or right) main wheel brake.

The reservoir is serviced with MIL-H-5606 hydraulic fluid (*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21*), or MIL-PRF-87257 hydraulic fluid (*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs*).

Brake system malfunction or impending brake failure may be indicated by a gradual decrease in braking action after brake application, noisy or dragging brakes, soft or spongy pedals, excessive travel, and/or weak braking action. A temperature sensor is mounted to each brake assembly and provides measured brake temperatures to the avionics system for caution and warning annunciation.

Should any of these symptoms occur, immediate maintenance is required. If, during taxi or landing roll, braking action decreases, let up on the toe brakes and then reapply the brakes with heavy pressure. If the brakes are spongy or pedal travel increases, pumping the pedals may build braking pressure.

Refer to Section 10, *Taxiing, Steering, and Braking Practices* for Brake System operational considerations.

### **Parking Brake**

#### **• Caution •**

Do not set the PARK BRAKE in flight. If a landing is made with the parking brake valve set, the brakes will maintain any pressure applied after touchdown.

The main wheel brakes are set for parking by using the PARK BRAKE handle on the right side kick plate near the pilot's right knee. Brake lines from the toe brakes to the main wheel brake calipers are plumbed through a parking brake valve. For normal operation, the handle is pushed in. With the handle pushed in, poppets in the valve are mechanically held open allowing normal brake operation. When the handle is pulled out, the parking brake valve holds applied brake pressure, locking the brakes. To apply the parking brake, set the brakes with the rudder-pedal toe brakes, and then pull the PARK BRAKE handle aft.

## Baggage Compartment

The baggage compartment, located on the left side of the fuselage aft of the wing, allows entry to the baggage compartment. The baggage door is hinged on the forward edge and latched on the rear edge. The door is locked from the outside with a key lock. The baggage compartment key will also open the cabin doors.

The baggage compartment extends from behind the rear passenger seat to the aft cabin bulkhead. The rear seats can be folded forward to provide additional baggage area for long or bulky items.

### Baggage Tie-Downs/Cargo Net

#### • Caution •

If not adequately restrained, baggage compartment items may pose a projectile hazard to cabin occupants in the event of rapid deceleration. Secure all baggage items with tie-down straps or cargo net.

Four baggage tie-down straps are provided to secure items in the baggage compartment. Each strap assembly has a hook at each end and a cam-lock buckle in the middle. The hook ends clip over loop fittings installed in the baggage floor and in the rear bulkhead. The tie-down straps should be stowed attached and tightened to the fittings.

The aircraft is equipped with a retractable cargo net to secure items in the baggage compartment. Integral inertia reels attached to the rear bulkhead allow the cargo net to be extended forward, placed over baggage, and secured to the seat back via four latch assemblies. The cargo net should be stowed attached to the seat back fittings.

The cargo net is not functional when rear seats are folded forward. Use conventional tie-down straps in this configuration.

For baggage area and door dimensions see Section 1, [Airplane Interior Dimensions](#).

## Seats

The seating arrangement consists of two individually adjustable seats for the pilot and front seat passenger and a “2+1” configuration with a one-piece bench seat and fold-down seat backs for the rear seat passengers.

### • Caution •

Do not kneel or stand on the seats. The seat bottoms have an integral aluminum honeycomb core designed to crush under impact to absorb downward loads.

## Front Seats

The front seats are adjustable fore and aft and the seat backs can be reclined for passenger comfort or folded forward for rear seat access. Integral headrests are provided. The fore and aft travel path is adjusted through the seat position control located below the forward edge of the seat cushion. The seat track is angled upward for forward travel so that shorter people will be positioned slightly higher as they adjust the seat forward. Recline position is controlled through levers located on each side of the seat backs. Depressing the recline release control while there is no pressure on the seat back will return the seat back to the full up position.

To position front seat fore and aft:

1. Lift the position control handle.
2. Slide the seat into position.
3. Release the handle and check that the seat is locked in place.

To adjust recline position:

1. Actuate and hold the seat back control lever.
2. Position the seat back to the desired angle.
3. Release the control lever.

## Rear Seats

The rear seats employ a one-piece bench seat and two seat backs configured in 60/40 split. This “2+1” seating configuration provides for a center seat/restraint area for a third passenger on the wider left hand seat.

Each seat back reclines independently of each other and can be folded forward to provide a semi-flat surface for cargo extending

forward from the baggage compartment. Recline position is controlled through a lever located on either side of the seat.

To fold seat back forward:

1. With no pressure on the seat back, rotate the lever to the recline position and fold the seat back forward.

## **Seat Belt and Shoulder Harness**

Integrated seat belt and shoulder harness assemblies with inertia reels are provided for the pilot and each passenger.

The front seats use a 4-point inflatable restraint system. Forward seat belts are attached to the seat frame. The shoulder harnesses are attached to inertia reels mounted in the seat back.

The rear seats use a 3-point safety harness consisting of one shoulder harness and a lap belt. The rear seat belts are attached to fittings on the cabin floor. The shoulder harnesses are attached to inertia reels mounted to the baggage compartment rear bulkhead.

Each front and rear seat shoulder harness is attached to the seat belt. The inertia reels allow complete freedom of movement of the occupant's upper torso. In the event of a sudden deceleration, the reels lock automatically to protect the occupants. It is recommended that the seat belts be stowed in the latched position when not in use.

### ***Front Seat Inflatable Restraints***

An inflatable shoulder harness is integral to each front seat harness. The electronic module assembly, mounted below the cabin floor, contains a crash sensor, battery, and related circuitry to monitor the deceleration rate of the airplane. In the event of a crash, the sensor evaluates the crash pulse and sends a signal to an inflator assembly mounted to the aft seat frame. This signal releases the gas in the inflator and rapidly inflates the airbag within the shoulder harness cover. After airbag deployment, the airbag deflates to enable the pilot/co-pilot to egress the airplane without obstruction.

The crash sensor's predetermined deployment threshold does not allow inadvertent deployment during normal operations, such as hard landings, strikes on the seat, or random vibration.

#### **• Caution •**

No slack may exist between the occupant's shoulder and restraint harness shoulder strap.

Stow the seat belts in the latched position when not in use.

To use the restraints:

1. Slip arms behind the harness so that the harness extends over shoulders.
2. Hold the buckle and firmly insert the link.
3. Grasp the seat belt tabs outboard of the link and buckle and pull to tighten. Buckle should be centered over hips for maximum comfort and safety.
4. Restraint harnesses should fit snug against the shoulder with the lap buckle centered and tightened around the hips.

To release the restraints:

1. Grasp the top of the buckle opposite the link and pull outward. The link will slip free of buckle.
2. Slip arms from behind the harness.

### ***Child Restraint System***

The aircraft is equipped with provisions for installing two LATCH compliant child seats in the outboard rear seat positions, OR one non-LATCH compliant seat in the center rear seat position.

Lower anchors for the LATCH compliant seats are located in the outboard seat positions. The non-LATCH compliant seat must be

installed using the center seat belt. Three top tether anchors for the child seats are located on the rear bulkhead.

To install a child seat:

1. Fasten lower seat attachments to bench seat:
  - a. *LATCH Compliant Outboard Seat*: Fasten lower seat attachment to the outboard anchors in the bench seat.
  - b. *Non-LATCH Complaint Center Seat*: Using the center seat belt, fasten lower seat attachments to the bench seat as described by the manufacturer's instructions
2. Locate top tether pass-through - a narrow slit in the seat back upholstery - near the top, outboard section of the seat back.

**• Caution •**

Do not route child seat top tether over or around seat back. The top tether must be routed through the seat back pass-through for the child seat to function properly.

3. Route child seat's top tether through the seat back pass-through.
4. Fasten top tether to rear bulkhead anchor.

Firmly tension the child seat straps according to the manufacturer's instructions.

## Cabin Doors

Two large forward hinged doors allow crew and passengers to enter and exit the cabin. The door handles engage striker pins in the door frame receptacles at the upper aft and lower aft door perimeter. Gas charged struts provide assistance in opening the doors and hold the doors open against gusts. Front seat armrests are integrated with the doors. A key lock in each door provides security. The cabin door keys also fit the baggage compartment door lock. Separate keys are provided for the fuel caps.

### Key Fob

*Serials 1233 & subs w/ Convenience Lighting:*

Remote operation of the door locks is provided by a battery-powered key fob.

This device complies with part 15 of the FCC Rules. Operation is subject to the following two conditions:

1. This device may not cause harmful interference.
2. This device must accept any interference received, including interference that may cause undesired operation.

• Note •

Key fob will not actuate door locks when BAT 1 switch is ON.

### Windshield and Windows

The windshield and side windows are manufactured of acrylic. Use only clean soft cloths and mild detergent to clean acrylic surfaces. Refer to Section 8, [Windshield and Windows](#) for detailed cleaning instructions.



## Engine

The airplane is powered by a Teledyne Continental TSIO-550-K, twin turbocharged, direct drive, fuel injected, air cooled, horizontally opposed 6 cylinder engine that uses a high pressure, wet sump style oil system for lubrication. The engine has a 550 cubic inch displacement and is rated to 315 bhp at 2500 RPM with a 2000-hour Time Between Overhaul (TBO) schedule. The engine utilizes a top air induction system, engine mounted throttle body, bottom exhaust system, and a full flow spin-on disposable oil filter. Engine front accessories include a hydraulically operated propeller governor, a gear-driven primary alternator and a belt-driven secondary alternator. Rear engine accessories include a starter, gear driven oil pump, gear driven fuel pump, and dual gear driven magnetos. The engine is attached to the firewall by a six-point steel engine mount. The firewall attach points are structurally reinforced with gusset-type attachments that transfer thrust and bending loads into the fuselage shell.

## Engine Controls

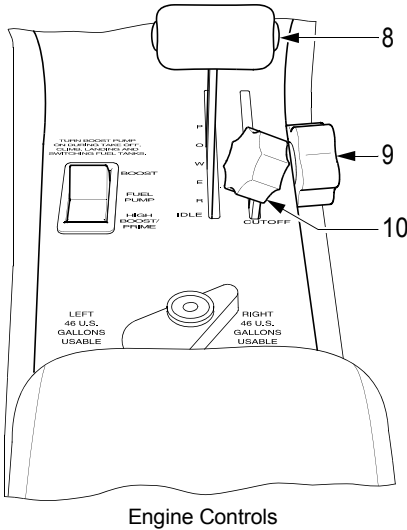
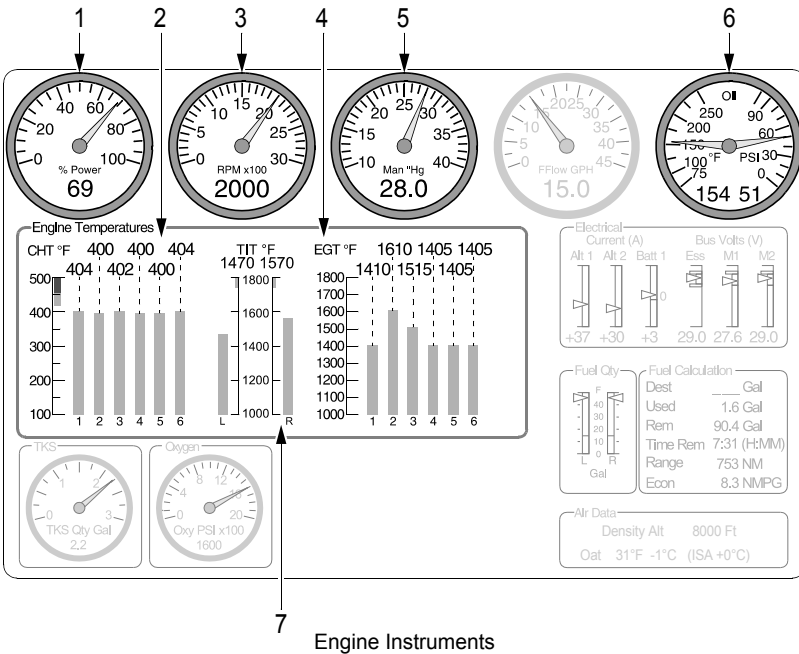
Engine controls are easily accessible to the pilot on a center console. They consist of a single-lever power (throttle) control and a mixture control lever. A friction control wheel, labeled FRICTION, on the right side of the console is used to adjust control lever resistance to rotation for feel and control setting stability.

### ***Power (Throttle) Lever***

The single-lever throttle control, labeled MAX-POWER-IDLE, on the console adjusts the engine throttle setting. The lever is mechanically linked by cable to the air throttle body/fuel-metering valve. Moving the lever towards MAX opens the air throttle butterfly and meters more fuel to the fuel manifold. No propeller control is required. The governor is set to 2500 maximum RPM in climb and cruise.

### ***Mixture Control***

The mixture control lever, labeled RICH-MIXTURE-CUTOFF, on the console adjusts the proportion of fuel to air for combustion. The Mixture Control Lever is mechanically linked to the mixture control valve in the engine-driven fuel pump. Moving the lever forward (towards RICH) repositions the valve allowing greater proportions of fuel and moving the lever aft (towards CUTOFF) reduces (leans) the proportion of fuel. Full aft position (CUTOFF) closes the control valve.



- LEGEND**
1. Percent Power
  2. CHT
  3. Tachometer
  4. EGT
  5. Manifold Pressure
  6. Oil Temperature and Pressure
  7. Turbine Inlet Temperature
  8. Power Lever
  9. Friction Control
  10. Mixture Control

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**Figure 7-7**  
**Engine Controls and Indicating**

## Engine Indicating

Engine information is displayed as analog-style gages, bar graphs, and text on the MFD's ENGINE page. When the ENGINE page is not active or in the case of an electronic display failure (backup mode), all essential engine information is displayed along the LH edge of the display. Engine data is acquired by the Engine Airframe Unit which transmits the data to the Engine Indicating System for display as described in the following pages.

- Note •

A "Red X" through any electronic display field indicates that the display field is not receiving valid data and should be considered inoperative.

### ***Engine System Annunciations***

Engine system health, caution, and warning messages are displayed in color-coded text in the Crew Alerting System (CAS) window located to the right of the Altimeter and Vertical Speed Indicator. In combination with a CAS alert, the affected engine parameter displayed on the ENGINE page changes to the corresponding color of CAS alert and the annunciation system issues an audio alert.

For specific pilot actions in response to Engine System Annunciations, refer to Section 3 - Emergency Procedures, [Engine System Emergencies](#), and Section 3A - Abnormal Procedures, [Engine System](#).

For additional information on Engine Instrument Markings and Annunciations, refer to [Section 2: Limitations](#).

For additional information on the System Annunciations And Alerts, refer to the [Perspective Integrated Avionics System](#) description in this section.

### ***Tachometer***

Engine speed (RPM) is shown in the upper mid-left corner of the ENGINE page as both a simulated tachometer and as a digital value. The tachometer pointer sweeps a scale range from 0 to 3000 RPM in 100 RPM increments. The digital RPM value is displayed in increments of 10 RPM in white numerals below the gage.

The tachometer receives a speed signal from a magnetic pickup sensor on the right hand magneto from the Engine Indicating System via the Engine Airframe Unit.

### ***Exhaust Gas Temperature (EGT)***

Exhaust gas temperatures for all six cylinders are displayed in the Engine Temperature block of the ENGINE page as vertical bars. The EGT graph is marked from 1000°F to 1800°F in 100°F increments. The digital EGT value of the cylinder is displayed above the bar in white numerals. A sensor in the exhaust pipe of each cylinder measures exhaust gas temperature and provides a voltage signal to the Engine Airframe Unit which processes and transmits the data to the Engine Indicating System.

### ***Cylinder Head Temperature (CHT)***

Cylinder head temperatures for all six cylinders are displayed in the Engine Temperature block of the ENGINE page as vertical bars. The CHT graph is marked from 100°F to 500°F in 100°F increments. The digital CHT value of the cylinder is displayed above the bar in white numerals.

A sensor in each cylinder head measures cylinder head temperature and provides a voltage signal to the Engine Airframe Unit which processes and transmits the data to the Engine Indicating System.

### ***Turbine Inlet Temperature (TIT)***

Turbine inlet temperature for the LH and RH turbochargers is displayed in the Engine Temperature block of the ENGINE page as vertical bars. The TIT graph is marked from 1000°F to 1800°F in 100°F increments. The digital TIT value of the turbine inlet is displayed above the bar in white numerals.

A sensor in each turbocharger measures turbine inlet temperature and provides a voltage signal to the Engine Airframe Unit which processes and transmits the data to the Engine Indicating System. The TIT gage is used as the primary source to lean fuel mixture.

### ***Oil Temperature***

Oil temperature is shown in the upper right corner of the ENGINE page, opposite the oil pressure scale, as both a simulated temperature gage and as a digital value. The gage pointer sweeps a scale range from 75°F to 250°F in 50°F increments. The digital temperature value is displayed in white numerals below the gage.

The oil temperature sensor is mounted below the oil cooler and provides a signal to the Engine Airframe Unit that is processed and transmitted to the Engine Indicating System for display.

### ***Oil Pressure***

Oil Pressure is shown in the upper right corner of the ENGINE page, opposite the oil temperature scale, as both a simulated pressure gage and as a digital value. The gage pointer sweeps a scale range from 0 to 90 PSI in 10 PSI increments. The digital pressure value is displayed in white numerals below the gage.

The oil pressure sensor is mounted below the oil cooler and provides a signal to the Engine Airframe Unit that is processed and transmitted to the Engine Indicating System for display.

### ***Manifold Pressure Gage***

Manifold pressure is shown in the upper center portion of the ENGINE page as both a simulated pressure gage and as a digital value. The gage pointer sweeps a scale range from 10 to 40 inches of mercury in 1 in.Hg increments. The digital MAP value is displayed in white numerals below the gage. The manifold pressure sensor is mounted in the induction air manifold near the throttle body and provides a signal to the Engine Airframe Unit that is processed and transmitted to the Engine Indicating System for display.

### ***Percent Power Gage***

Percent power is shown in the upper left corner of the ENGINE page as both a simulated gage and as a digital value. The percent power gage sweeps a scale marked from 0 to 100 percent in 5 percent increments. The digital percent power value is displayed in white numerals below the gage. The display units calculate the percentage of maximum engine power produced by the engine based on an algorithm employing manifold pressure, indicated air speed, outside air temperature, pressure altitude, engine speed, and fuel flow.

## Engine Lubrication System

The engine is provided with a wet-sump, high-pressure oil system for engine lubrication and cooling. Oil for engine lubrication is drawn from an eight-quart capacity sump through an oil suction strainer screen and directed through the oil filter to the engine-mounted oil cooler by a positive displacement oil pump. The oil pump is equipped with a pressure relief valve at the pump output end to bypass oil back to the pump inlet should the pump exceed limits. The oil cooler is equipped with a temperature control valve set to bypass oil if the temperature is below approximately 180°F (82°C). Bypass or cooled oil is then directed through oil galleries to the engine rotating parts and piston inner domes. Oil is also directed to the propeller governor to regulate propeller pitch. The complete oil system is contained in the engine. An oil filler cap and dipstick are located at the left rear of the engine. The filler cap and dipstick are accessed through a door on the top left side of the engine cowling.

## Ignition and Starter System

Two engine-driven magnetos and two spark plugs in each cylinder provide engine fuel ignition. The right magneto fires the lower right and upper left spark plugs, and the left magneto fires the lower left and upper right spark plugs. Normal operation is conducted with both magnetos, as more complete burning of the fuel-air mixture occurs with dual ignition. A rotary-type key switch, located on the instrument panel, controls ignition and starter operation. The switch is labeled OFF-R-L- BOTH-START. In the OFF position, the starter is electrically isolated, the magnetos are grounded and will not operate. Normally, the engine is operated on both magnetos (switch in BOTH position) except for magneto checks and emergency operations. The R and L positions are used for individual magneto checks and for single magneto operation when required. When the battery master switch is ON, rotating the switch to the spring loaded START position energizes the starter and activates both magnetos. The switch automatically returns to the BOTH position when released.

28 VDC for Starter operation is supplied through the 2-amp STARTER circuit breaker on NON-ESSENTIAL BUS.

## Air Induction System

Induction air enters the engine compartment through two NACA ducts located in the lower engine cowl. The air continues to the air boxes where it is filtered and ducted into the compressor housing where the compressor wheel, spinning at a high RPM, increases the air pressure which provides a greater mass of air available to the engine for the combustion cycle. From the compressor housing, the air is ducted through the intercoolers where the air temperature is lowered which further increases the density of the induction air. Downstream of the intercoolers the airflow joins at the “Y” junction of intake tubes, then passes through the throttle body and into the intake manifold where it is divided by the intake pipes flowing to each cylinder.

In the case of filter blockage or induction ice, alternate air can enter the engine via the alternate air assembly located in the lower front of the engine and connected to the LH and RH air boxes. Under normal conditions, the alternate air assembly door is held closed by magnetic force. If the air induction system should become blocked, suction created by the engine will open the door and draw unfiltered air from inside the cowl. When the door opens, a switch opens which causes an annunciation on the Primary Flight Display to alert the pilot.

Use of alternate air will result in a reduction of engine power due to lower inlet air pressure and higher air temperature. Additionally loss of manifold pressure will occur when operating at high altitude and low RPM where the turbocharger wastegate is closed.

## Engine Exhaust System

After leaving the cylinders, exhaust gases flow through the exhaust collector to the turbocharger turbine housing inlet. The exhaust gas flow provides turbine wheel rotation then exits through the turbine housing discharge port and overboard through tailpipes exiting through the lower cowl.

## Engine Fuel Injection

The multi-nozzle, continuous-flow fuel injection system supplies fuel for engine operation. An engine driven fuel pump draws fuel from the selected wing tank and passes it to the mixture control valve integral to the pump. The mixture control valve proportions fuel in response to the pilot operated mixture control lever position. From the mixture control, fuel is routed to the fuel-metering valve on the air-induction system throttle body. The fuel-metering valve adjusts fuel flow in

response to the pilot controlled Power Lever position. From the metering valve, fuel is directed to the fuel manifold valve (spider) and then to the individual injector nozzles. The system meters fuel flow in proportion to engine RPM, mixture setting, and throttle angle. Manual mixture control and idle cut-off are provided. An electric fuel pump provides fuel boost for vapor suppression and for priming.

## **Engine Cooling**

Engine cooling is accomplished by discharging heat to the oil and then to the air passing through the oil cooler, and by discharging heat directly to the air flowing past the engine. Cooling air enters the engine compartment through the two inlets in the cowling. Aluminum baffles direct the incoming air to the engine and over the engine cylinder cooling fins where the heat transfer takes place. The heated air exits the engine compartment through louvered vents in the bottom of the cowlings. No movable cowl flaps are used.

## **Turbochargers**

The TSIO-550-K has twin turbochargers which use exhaust gas flow to boost induction air pressure for increased power. There is one turbocharger on each side of the engine. The turbochargers compress and raise the temperature of the incoming air before going to the intercoolers. The dual turbochargers are lubricated from external oil supply lines from a source at the bottom of the oil cooler. There is one oil pressure actuated wastegate on the left side of the engine controlling the amount of exhaust gas used by the turbochargers. Control is accomplished by a diaphragm actuated valve sensing differential pressure across the throttle plate and controlling the oil return flow rate from the wastegate. An overboost valve in the induction system provides protection from too much pressure by actuating at overly high manifold pressures.



## **Propeller**

The airplane is equipped with a composite, three-blade, constant speed, governor-regulated propeller.

The propeller governor automatically adjusts propeller pitch to regulate propeller and engine RPM by controlling the flow of engine oil - boosted to high pressure by the governing pump - to or from a piston in the propeller hub. Oil pressure acting on the piston twists the blades toward high pitch (low RPM). When oil pressure to the piston in the propeller hub is relieved, centrifugal force, assisted by an internal spring, twists the blades toward low pitch (high RPM). Any change in airspeed or load on the propeller results in a change in propeller pitch.

During climb and cruise, the governor automatically adjusts propeller pitch in order to maintain a 2500 RPM setting.

## Fuel System

An 92-gallon usable wet-wing fuel storage system provides fuel for engine operation. The system consists of a 47.25-gallon capacity (46-gallon usable) vented integral fuel tank and a fuel collector/sump in each wing, a three position selector valve, an electric fuel pump, and an engine-driven fuel pump. Fuel is gravity fed from each tank to the associated collector sumps where the engine-driven fuel pump draws fuel through a filter and selector valve to pressure feed the engine fuel injection system. The electric fuel pump is provided for engine priming and vapor suppression.

Each integral wing fuel tank has a filler cap in the upper surface of each wing for fuel servicing. Access panels in the lower surface of each wing allow access to the associated wet compartment (tank) for inspection and maintenance. Float-type fuel quantity sensors in each wing tank supply fuel level information to the fuel quantity gages. Positive pressure in the tank is maintained through a vent line from each wing tank. Fuel, from each wing tank, gravity feeds through strainers and a flapper valve to the associated collector tank in each wing. Each collector tank/sump incorporates a flush mounted fuel drain and a vent to the associated fuel tank.

The engine-driven fuel pump pulls filtered fuel from the two collector tanks through a three-position (LEFT-RIGHT-OFF) selector valve. The selector valve allows tank selection. From the fuel pump, the fuel is metered to a flow divider, and delivered to the individual cylinders. Excess fuel is returned to the selected tank.

A simulated fuel quantity gage is located on the Engine Strip along the left edge of the MFD and in the Fuel Qty block on the MFD's Engine page.

Fuel shutoff and tank selection is positioned nearby for easy access. Fuel system venting is essential to system operation. Blockage of the system will result in decreasing fuel flow and eventual engine fuel starvation and stoppage. Venting is accomplished independently from each tank by a vent line leading to a NACA-type vent mounted in an access panel underneath the wing near each wing tip.

The airplane may be serviced to a reduced capacity to permit heavier cabin loadings. This is accomplished by filling each tank to a tab visible below the fuel filler, giving a reduced fuel load of 30.0 gallons usable in each tank (60 gallons total usable in all flight conditions).

Drain valves at the system low points allow draining the system for maintenance and for examination of fuel in the system for contamination and grade. The fuel must be sampled prior to each flight. A sampler cup is provided to drain a small amount of fuel from the wing tank drains, the collector tank drains, and the gascolator drain. If takeoff weight limitations for the next flight permit, the fuel tanks should be filled after each flight to prevent condensation.

## **Fuel Selector Valve**

A fuel selector valve, located at the rear of the center console, provides the following functions:

- LEFT Allows fuel to flow from the left tank
- RIGHT Allows fuel to flow from the right tank
- OFF Cuts off fuel flow from both tanks

The valve is arranged so that to feed off a particular tank the valve should be pointed to the fuel indicator for that tank. To select RIGHT or LEFT, rotate the selector to the desired position. To select Off, first raise the fuel selector knob release and then rotate the knob to OFF.

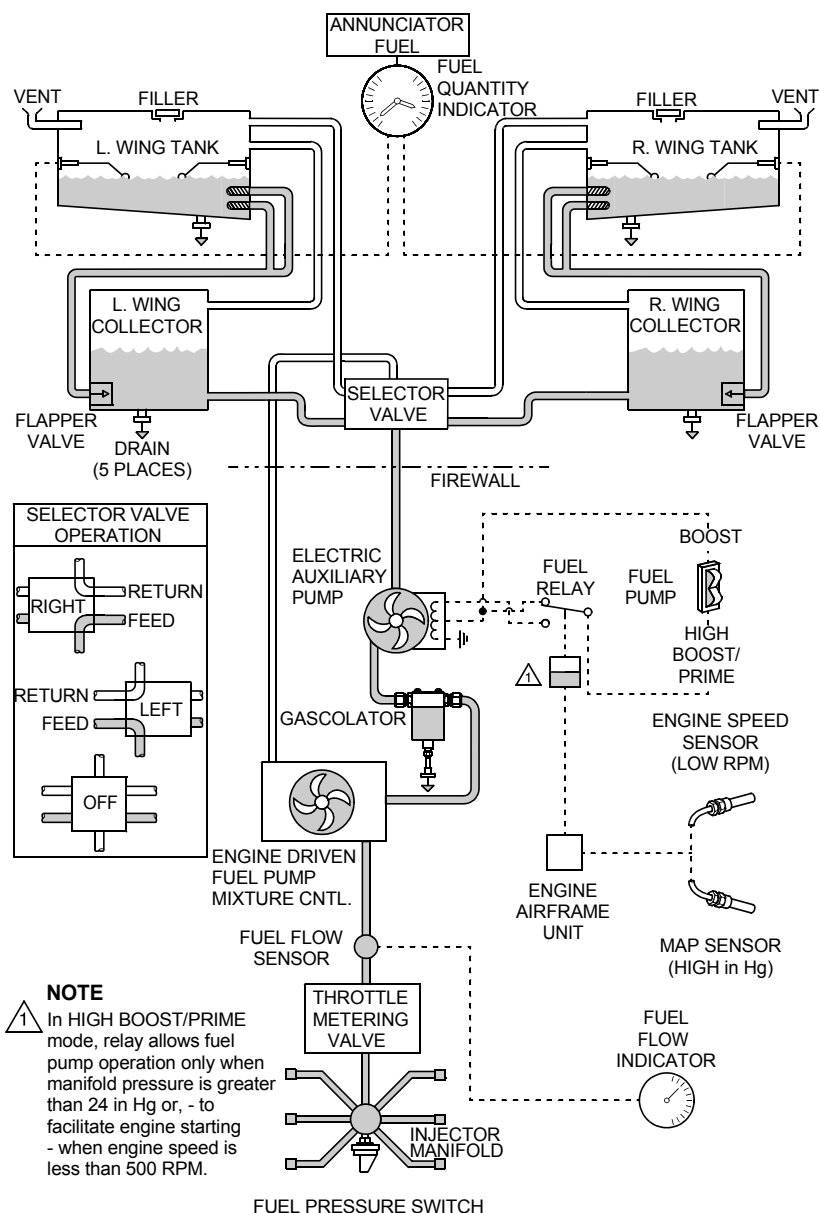
## **Fuel Pump Operation**

Fuel pump operation and engine prime is controlled through the Fuel Pump rocker switch located adjacent to the fuel selector valve.

To prevent over-priming, the system uses a lockout relay that only allows HIGH BOOST/PRIME for engine start and operations at high power settings. When the manifold pressure is less than 24 in-Hg and engine RPM is greater than 500 RPM, pressing the HIGH BOOST/PRIME limits the fuel pump to low-speed mode BOOST.

Selecting BOOST energizes the fuel pump in low-speed mode regardless of engine speed or manifold pressure to deliver a continuous 4-6 psi boost to the fuel flow for vapor suppression in a hot fuel condition.

The fuel pump operates on 28 VDC supplied through the 5-amp FUEL PUMP circuit breaker on MAIN BUS 2.



FUEL PRESSURE SWITCH

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**Figure 7-8**  
**Fuel System Schematic**

## Fuel Indicating

Fuel quantity is measured by float-type quantity sensors installed in each fuel tank and displayed on the Fuel Quantity Gage.

### • Caution •

When the fuel tanks are 1/4 full or less, prolonged uncoordinated flight such as slips or skids can uncover the fuel tank outlets. Therefore, if operating with one fuel tank dry or if operating on LEFT or RIGHT tank when 1/4 full or less, do not allow the airplane to remain in uncoordinated flight for periods in excess of 30 seconds.

### • Note •

A “Red X” through any electronic display field indicates that the display field is not receiving valid data and should be considered inoperative.

### ***Fuel Quantity Gage***

A dual reading fuel quantity gage is displayed on the Engine Strip along the left edge of the MFD and in the Fuel Qty block of the ENGINE page. In the case of an electronic display failure (backup mode), all essential fuel information is displayed on the Engine Strip along the left edge of the PFD. The LEFT pointer indicates left tank fuel quantity and sweeps a vertical bar scale marked from 0 to 46 U.S. gallons in 5-gallon increments. The RIGHT pointer sweeps an identical scale for the right tank. Each scale is marked with a yellow band from 0 to 14 U.S. gallons and a green band from 14 to 46 U.S. gallons. The indicators are calibrated to read 0 gallons when no usable fuel remains.

Fuel quantity is measured by a float type quantity sensors installed in the fuel tanks. Fluid quantity information is sent to the Engine Airframe Unit, processed, and transmitted to the simulated Fuel Quantity Gage and CAS window for display.

### ***Fuel Flow***

Fuel Flow is shown in the upper mid right corner of the ENGINE page as both a simulated gage and as a digital value. The gage pointer sweeps a scale range from 0 to 45 Gallons Per Hour (GPH). The fuel flow value is displayed in white numerals below the gage. Fuel flow is measured by a transducer on the right side of the engine in the fuel line between the engine driven fuel pump and distribution block. The

fuel flow signal is sent to the Engine Airframe Unit, processed, and transmitted to the Engine Indicating System for display.

The gage displays a green normal arc, which is dynamically updated to display the range of normal values appropriate to engine power settings.

- When manifold pressures is above 30.6 in.Hg, the green arc covers a narrow range depicting the full rich fuel flow appropriate for that power setting.
- For manifold pressures of 30.5 in.Hg and below, the normal arc extends from full rich limit to 10 gallons per hour (indicating that cruise leaning is permitted).

Target Fuel Flow is a cyan pointer placed on the fuel gage to indicate best economy target fuel flow. This pointer is displayed when cruise leaning is allowed (manifold pressure of 30.5 in.Hg or below), it will be presented after the green arc expands when power is reduced. This indicator provides guidance to aide in cruise leaning, it is calculated to provide a fuel flow closely corresponding with the best economy fuel to air ratio.

• Note •

Target Fuel Flow is removed from gage when resulting engine power would be less than 55% (intended for cruise, not descent leaning guidance).

### ***Fuel Totalizer and Calculated Information***

Fuel totalizer calculations are located in the lower right section of the ENGINE page and are separate and independent of the fuel quantity gage and float sensor system. The fuel totalizer monitors fuel flow and calculates fuel-to-destination, fuel used, fuel remaining, time remaining, fuel range, and nautical miles per gallon. Upon system startup, the fuel totalizer initial fuel screen appears and prompts the user to enter the total fuel on board at start. The option to enter the number of gallons added since last fuel fill and the ability to set fuel to "Full" or to "Tabs" buttons is also available.

### ***Fuel System Annunciations***

Fuel system health, caution, and warning messages are displayed in color-coded text in the Crew Alerting System (CAS) window located to the right of the Altimeter and Vertical Speed Indicator. In combination with a CAS alert, the affected fuel parameter displayed on the

ENGINE page changes to the corresponding color of CAS alert and the annunciation system issues an audio alert.

FUEL QTY: Fuel quantity.

- A white Advisory message is generated when either fuel tank goes below 14 gallons.
- An amber Caution message is generated when both fuel tanks go below 14 gallons.
- A red Warning message is generated when the *fuel totalizer* amount goes below 9 gallons, or if the sensed fuel quantity (left plus right) is below 9 gallons.

FUEL IMBALANCE: Imbalanced between Left and Right Tanks.

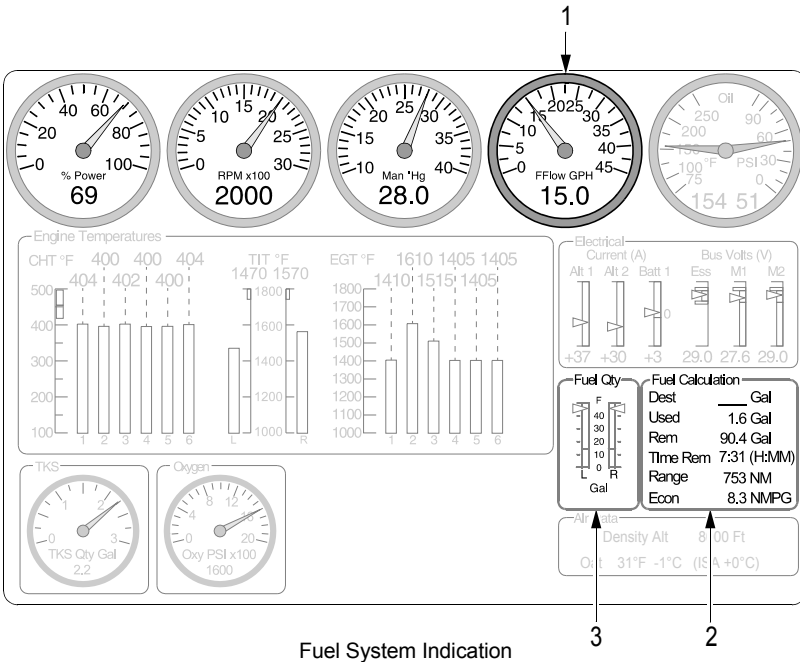
- A white Advisory message is generated when the imbalance between tanks is greater than 8 gallons.
- An amber Caution message is generated when the imbalance between tanks is greater than 10 gallons.
- A red Warning message is generated when the imbalance between tanks is greater than 12 gallons.

• Note •

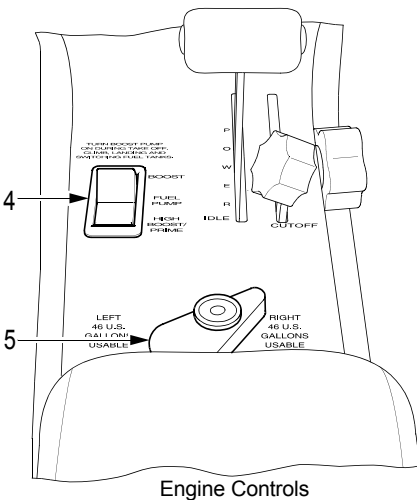
For specific pilot actions in response to Fuel System Annunciations, refer to Section 3 - Emergency Procedures, [Fuel System Emergencies](#), and Section 3A - Abnormal Procedures, [Fuel System](#).

For additional information on Engine Instrument Markings and Annunciations, refer to [Section 2: Limitations](#).

For additional information on the System Annunciations And Alerts, refer to the [Perspective Integrated Avionics System](#) description in this section.



Fuel System Indication



Engine Controls

### LEGEND

- Fuel Flow Gage
- Fuel Calculations:
  - Fuel At Destination (Totalizer)
  - Fuel Used (Totalizer)
  - Fuel Remaining (Totalizer)
  - Time Remaining (Totalizer)
  - Fuel Range (Totalizer)
  - Nautical Miles Per Gallon (Totalizer)
- Fuel Quantity Gage (Float Sensor)
- Fuel Pump Switch
- Fuel Selector Valve

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**Figure 7-9**  
**Fuel System Controls and Indicating**



## Electrical System

The airplane is equipped with a two-alternator, two-battery, 28-volt direct current (VDC) electrical system designed to reduce the risk of electrical system faults. The system provides uninterrupted power for avionics, flight instrumentation, lighting, and other electrically operated and controlled systems during normal operation.

### Power Generation

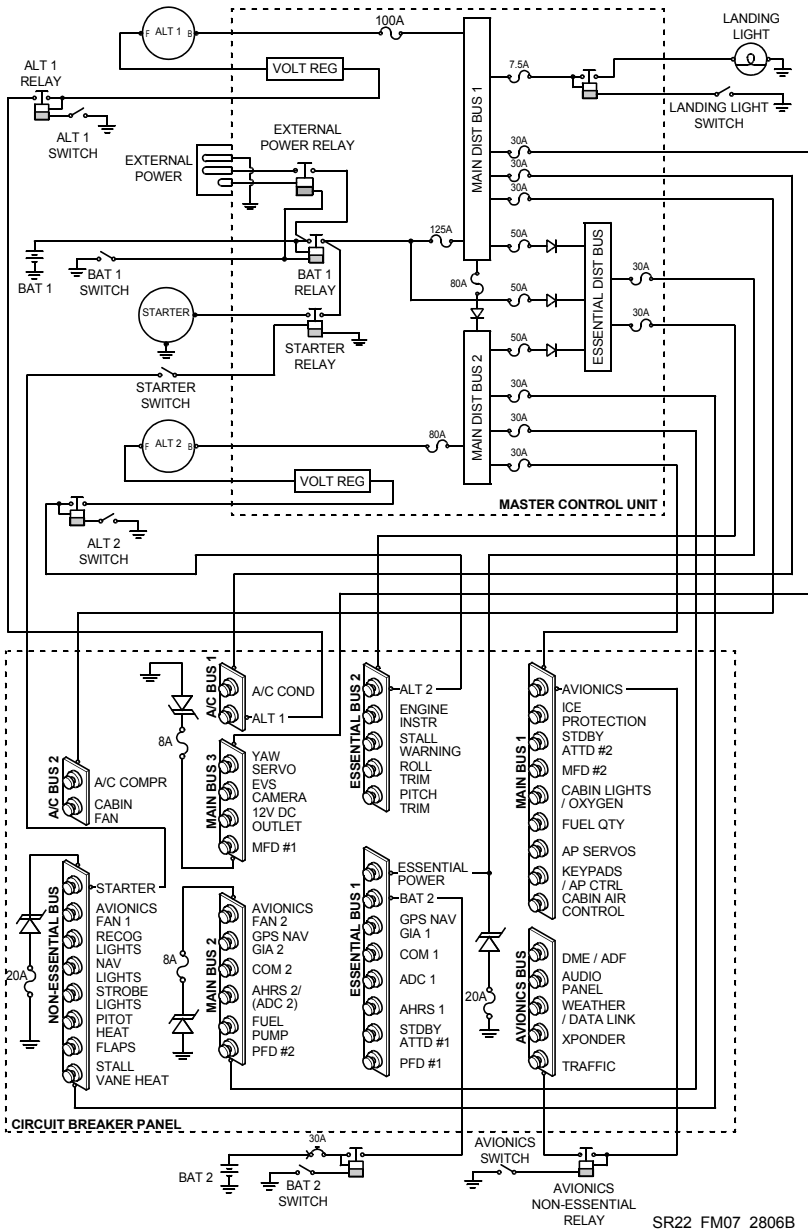
Primary power for the airplane is supplied by a 28-VDC, negative-ground electrical system. The electrical power generation system consists of two alternators controlled by a Master Control Unit (MCU) mounted on the left side of the firewall and two batteries for starting and electrical power storage.

Alternator 1 (ALT 1) is a gear-driven, internally rectified, 100-amp alternator mounted on the right front of the engine. Alternator 2 (ALT 2) is a belt-driven, internally rectified, 70-amp alternator mounted on the front left of the engine. ALT 1 is regulated to 28 volts and ALT 2 is regulated to 28.75 volts. Both alternators are self-exciting and require battery voltage for field excitation in order to start up - for this reason, the batteries should not be turned off in flight.

### Storage

Battery 1 (BAT 1) is an aviation grade 12-cell, lead-acid, 24-volt, 10-amp-hour battery mounted on the right firewall. BAT 1 is charged from the Main Distribution Bus 1 in the MCU.

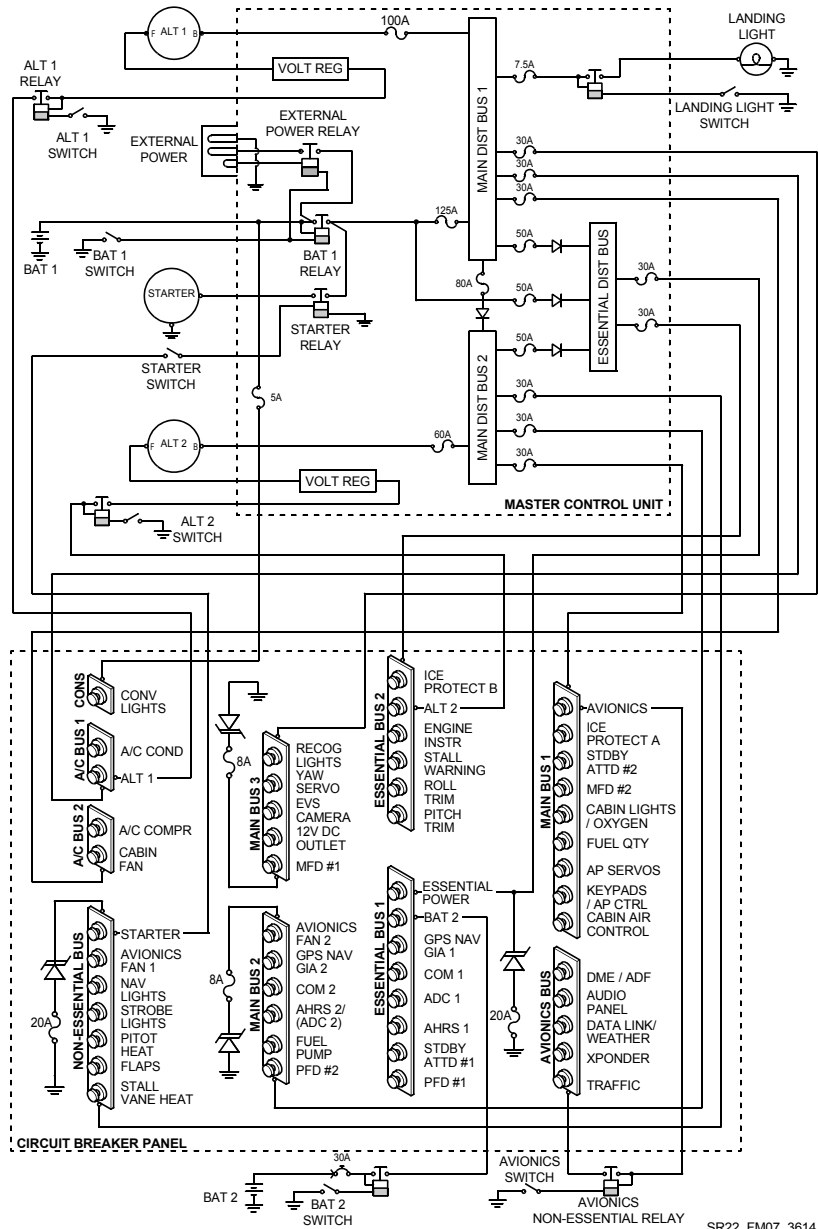
Battery 2 (BAT 2) is composed of two 12-volt, 7-amp-hour, sealed, lead-acid batteries connected in series to provide 24 volts. Both BAT 2 units are located in a vented, acid-resistant container mounted behind the aft cabin bulkhead (FS 222) below the parachute canister. BAT 2 is charged from the circuit breaker panel ESS BUS 1.



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Figure 7-10

Electrical System Schematic - Serials 0442 thru 0656, 0658 thru 0689 (1 of 2)



SR22\_FM07\_3614

**Figure 7-10**  
**Electrical System Schematic - Serials 0657, 0690 & subs (2 of 2)**

## Power Distribution

Power is supplied to the airplane circuits through three distribution buses contained in the MCU: Main Distribution Bus 1, Main Distribution Bus 2, and the Essential Distribution Bus. The three distribution buses power the associated buses on the circuit breaker panel.

### ***Master Control Unit***

The Master Control Unit (MCU) is located on the left firewall. The MCU controls ALT 1, ALT 2, starter, landing light, external power, and power generation functions. In addition to ALT 1 and ALT 2 voltage regulation, the MCU also provides external power reverse polarity protection, alternator overvoltage protection, as well as electrical system health annunciations to the Integrated Avionics System. Power is distributed to the airplane circuit panel buses through Main and Essential buses in the MCU. The Main distribution buses are interconnected by an 80-amp fuse and a diode. The diode prevents ALT 2 from feeding the Main Distribution Bus 1. Additionally, since ALT 2 Bus voltage is slightly higher than ALT 1 voltage, bus separation is further assured.

### ***Essential Distribution Bus***

The Essential Distribution Bus is fed by both Main Distribution Bus 1 and Main Distribution Bus 2 in the MCU through two 50-amp fuses. The Essential Bus powers two circuit breaker buses through 30-amp fuses located in the MCU:

- ESS BUS 1,
- ESS BUS 2.

### ***Main Distribution Bus 1***

The output from ALT 1 is connected to the Main Distribution Bus 1 in the MCU through a 100-amp fuse. Main Distribution Bus 1 directly powers the Landing Light through a 7.5-amp fuse and three circuit breaker buses through 30-amp fuses located in the MCU:

- A/C BUS 1,
- A/C BUS 2,
- MAIN BUS 3.

### **Main Distribution Bus 2**

The output from ALT 2 is connected to the Main Distribution Bus 2 in the MCU through an 80-amp fuse. Main Distribution Bus 2 powers three circuit breaker buses through 30-amp fuses located in the MCU:

- NON ESS BUS,
- MAIN BUS 1,
- MAIN BUS 2.

### **Constant Power Bus - serials 0657, 0690 & subs**

The Constant Power Bus is fed by BAT 1 in the MCU through one 5-amp fuse located on top of the MCU.

## **Electrical System Protection**

### **Circuit Breakers, Fuses and Voltage Suppressors**

Individual electrical circuits connected to the Main, Essential, and Non-Essential Buses in the airplane are protected by re-settable circuit breakers mounted in the circuit breaker panel on the left side of the center console. Loads on circuit breaker panel buses are shed by pulling the individual circuit breakers.

#### Transient Voltage Suppressors

Transient Voltage Suppressors (TVS) are installed in key areas of the electrical system to protect the system from lightning strikes. During lightning strikes, enormous energy spikes can be induced within the airplane electrical system. In the absence of any transient protection, this unwanted energy would typically be dissipated in the form of high-voltage discharge across the avionics and electrical systems of the airplane. By adding a high power TVS at key power entry points on the electrical busses, unwanted energy from electrical transients is allowed to dissipate through a semi-conducting pathway to ground.

#### **• Caution •**

If smoke and/or fumes are detected in the cabin and it is suspected that this event was caused by a TVS failure, the operator should confirm that there is no fire and perform the [Smoke and Fume Elimination](#) Checklist.

### **Essential Buses**

The circuit breaker panel ESS BUS 1 and ESS BUS 2 are powered directly by ALT 1 and ALT 2 from the MCU Essential Distribution Bus

through 30-amp fuses inside the MCU and also by BAT 2 through the 20-amp BAT 2 circuit breaker.

In the event of ALT 1 or ALT 2 failure, the Essential Buses in the circuit breaker panel will be powered by the remaining alternator through the Main Distribution Bus 1 or Main Distribution Bus 2 in the MCU. In the case of both alternators failing, BAT 1 is connected directly to the Essential Distribution Bus in the MCU and will power ESS BUS 1 and ESS BUS 2. In the event of both alternators and BAT 1 failing, BAT 2 is connected directly to ESS BUS 1.

### ***Main Buses***

The circuit breaker panel MAIN BUS 1 and MAIN BUS 2 are powered by ALT 2 from the MCU Main Distribution Bus 2 and - in the event of ALT 2 failure - by ALT 1 and BAT 1 from the Main Distribution Bus 2 via the diode interconnecting the MCU distribution buses through 30-amp fuses inside the MCU.

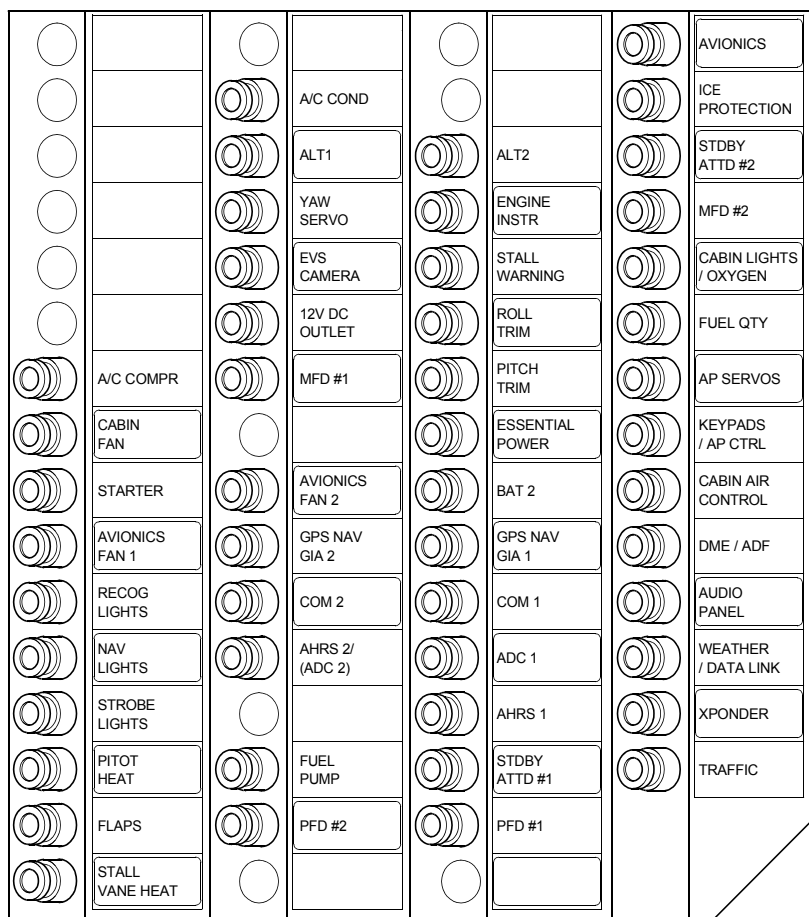
The 10-amp AVIONICS circuit breaker on MAIN BUS 1, controlled through the AVIONICS master switch on the bolster switch panel, powers all loads on the AVIONICS BUS.

The circuit breaker panel MAIN BUS 3 is powered by ALT 1 and BAT 1 from the MCU Main Distribution Bus 1 through a 30-amp fuse inside the MCU. In the event of ALT 1 failure, BAT 1 will power MAIN BUS 3. ALT 2 is prevented from powering MAIN BUS 3 by the isolation diode interconnecting the MCU distribution buses 1 and 2.

### ***Non-Essential Buses***

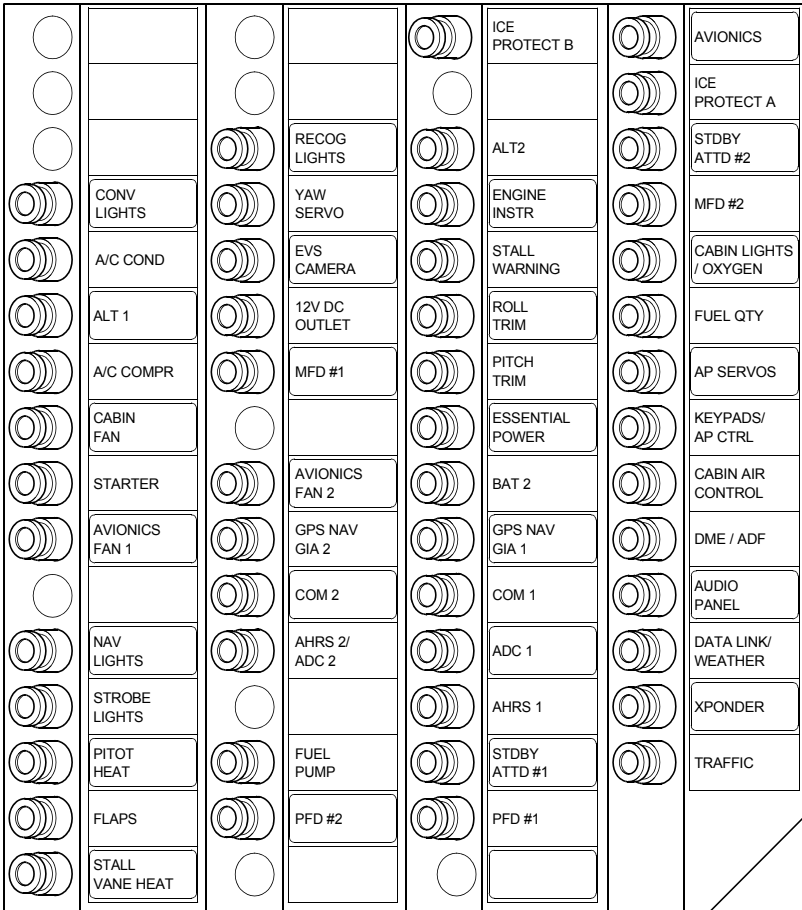
The circuit breaker panel NON ESS BUS is powered by ALT 2 from the MCU Main Distribution Bus 2 and - in the event of ALT 2 failure - by ALT 1 and BAT 1 from the Main Distribution Bus 2 via the diode interconnecting the MCU distribution buses through 30-amp fuses inside the MCU. The Avionics Non-Essential Bus is powered through the 10-amp AVIONICS circuit breaker on MAIN BUS 1 and is discussed above.

The circuit breaker panel A/C BUS 1 and A/C BUS 2, is powered by ALT 1 and BAT 1 from the MCU Main Distribution Bus 1 through a 30-amp fuse inside the MCU. In the event of ALT 1 failure, BAT 1 will power A/C BUS 1 and A/C BUS 2. ALT 2 is prevented from powering A/C BUS 1 and A/C BUS 2 by the isolation diode interconnecting the MCU distribution buses 1 and 2.



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**Figure 7-11**  
**Circuit Breaker Panel - Serials 0442 thru 0656, 0658 thru 0689 (1 of 2)**  
**P/N 13772-005**  
**Revision 1**



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**Figure 7-11**  
**Circuit Breaker Panel - Serials 0657, 0690 & subs (2 of 2)**



## Electrical System Control

The rocker type electrical system MASTER switches are 'on' in the up position and 'off' in the down position. The switches, labeled BAT 2, BAT 1, ALT 1, ALT 2 are located in the bolster switch panel immediately below the instrument panel. These switches, along with the AVIONICS power switch, control all electrical power to the airplane.

### ***Battery Switches***

The BAT 1 and BAT 2 switches control the respective battery. Setting the BAT 1 switch 'on' energizes a relay connecting BAT 1 to the MCU Distribution Buses (also energizing the circuit breaker panel buses) and the open contacts of the starter relay. Setting the BAT 2 switch 'on' energizes a relay connecting BAT 2 to the circuit breaker panel ESS BUS 1. Normally, for flight operations, all master switches will be 'on'. However, the BAT 1 and BAT 2 switches can be turned 'on' separately to check equipment while on the ground. Setting only the BAT 2 switch 'on' will energize those systems connected to the circuit breaker panel's ESS BUS 1 and ESS BUS 2. If any system on the other buses is energized, a failure of the Distribution Bus interconnect isolation diode is indicated. When the BAT 1 switch is set to 'on,' the remaining systems will be energized. To check or use non-essential avionics equipment or radios while on the ground, the AVIONICS power switch must also be turned on.

### ***Alternator Switches***

The ALT 1 and ALT 2 switches control field power to the respective alternator. For ALT 1 to start, the BAT 1 switch must be 'on'. Setting the ALT 1 switch 'on' energizes a relay allowing 28 VDC from the 5 amp ALT 1 circuit breaker on A/C BUS 1 to be applied to a voltage regulator for ALT 1. For ALT 2 to start, either the BAT 1 switch or the BAT 2 switch must be 'on.' Setting the ALT 2 switch 'on' energizes a relay allowing 28 VDC from the 5 amp ALT 2 circuit breaker on ESS BUS 2 to be applied to voltage regulator for ALT 2. Positioning either ALT switch to the OFF position removes the affected alternator from the electrical system.

#### **• Caution •**

Continued operation with the alternators switched off will reduce battery power enough to open the battery relay, remove power from the alternator field, and prevent alternator restart.

### ***Avionics Power Switch***

A rocker switch, labeled AVIONICS, controls electrical power from the circuit breaker panel (MAIN BUS 1) to the Avionics Bus. The switch is located next to the ALT and BAT Master switches. Typically, the switch is used to energize or de-energize all non-essential avionics on the AVIONICS bus simultaneously. With the switch in the OFF position, no electrical power will be applied to the non-essential avionics equipment, regardless of the position of the MASTER switch or the individual equipment switches. For normal operations, the AVIONICS switch should be placed in the OFF position prior to activating the MASTER switches, starting the engine, or applying an external power source.

### **Ground Service Receptacle**

A ground service receptacle is located just aft of the cowl on the left side of the airplane. This receptacle is installed to permit the use of an external power source for cold weather starting and maintenance procedures requiring reliable power for an extended period. The external power source must be regulated to 28 VDC. The external power control contactor is wired through the BAT 1 MASTER switch so that the BAT 1 switch must be 'on' to apply external power.

Refer to Section 8, [Ground Handling](#) for use of external power and special precautions to be followed.

## Electrical Indicating

Electrical system information is displayed as bar graphs and text on the MFD's ENGINE page. When the ENGINE page is not active or in the case of an electronic display failure (backup mode), Battery 1 ampere output and Essential Bus voltage output are displayed along the LH edge of the display. Electrical data is acquired by the Engine Airframe Unit which transmits the data to the Engine Indicating System for display as described in the following pages.

• Note •

A "Red X" through any electronic display field indicates that the display field is not receiving valid data and should be considered inoperative.

### ***Electrical System Annunciations***

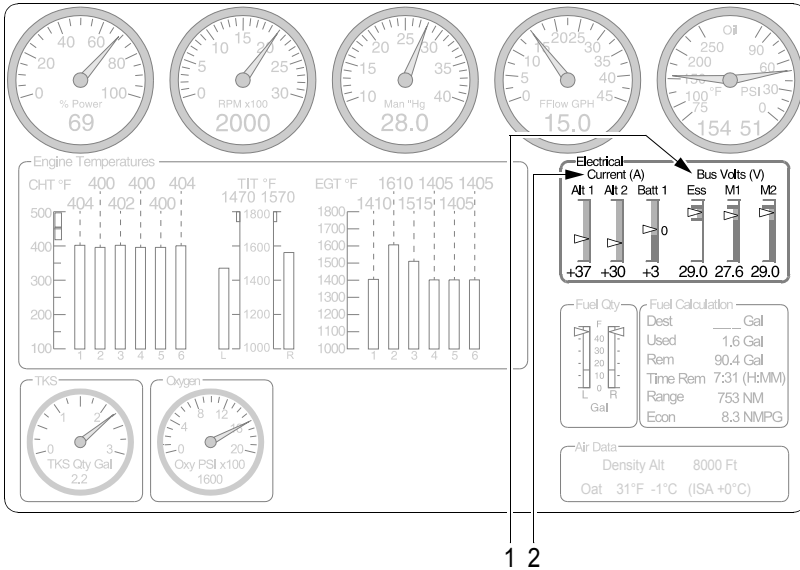
Electrical system health, caution, and warning messages are displayed in color-coded text in the Crew Alerting System (CAS) window located to the right of the Altimeter and Vertical Speed Indicator. In combination with a CAS alert, the affected electrical parameter displayed on the ENGINE page changes to the corresponding color of CAS alert and the annunciation system issues an audio alert.

• Note •

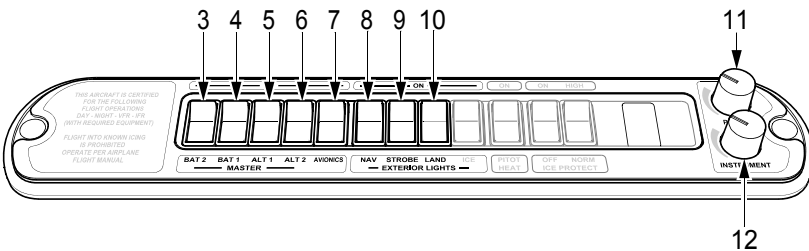
For specific pilot actions in response to Electrical System Annunciations, refer to Section 3 - Emergency Procedures, [Electrical System Emergencies](#), and Section 3A - Abnormal Procedures, [Electrical System](#).

For additional information on Engine Instrument Markings and Annunciations, refer to [Section 2: Limitations](#).

For additional information on the System Annunciations And Alerts, refer to the [Perspective Integrated Avionics System](#) description in this section.



**Electrical System Indication**



**Electrical and Lighting Controls**

**LEGEND**

- |                                 |                       |
|---------------------------------|-----------------------|
| 1. Essential & Main Bus Voltage | 7. Avionics           |
| 2. Alternator & Battery Current | 8. Navigation         |
| 3. Battery 2                    | 9. Strobe             |
| 4. Battery 1                    | 10. Landing Light     |
| 5. Alternator 1                 | 11. Panel Dimmer      |
| 6. Alternator 2                 | 12. Instrument Dimmer |

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**Figure 7-12**  
**Electrical / Lighting Controls and Indicating**

# Lighting Systems

## Exterior Lighting

The airplane is equipped with wing tip navigation lights with integral anti-collision strobe lights and recognition Lights. The landing light is located in the lower cowl.

### ***Navigation Lights***

The airplane is equipped with standard wing tip navigation lights. The lights are controlled through the NAV light switch on the instrument panel bolster.

28 VDC for navigation light operation is supplied through the 5-amp NAV LIGHTS circuit breaker on the NON ESS BUS.

### ***Strobe Light***

Anti-collision strobe lights are installed integral with the standard navigation lights. Each strobe is flashed by a separate power supply. The strobe power supplies are controlled through the STROBE light switch on the instrument panel bolster.

28 VDC for strobe light and control circuits is supplied through the 5-amp STROBE LIGHTS circuit breaker on the NON ESS BUS.

### ***Landing Light***

A High Intensity Discharge (HID) landing light is mounted in the lower engine cowl. The landing light is controlled through the LAND light switch on the instrument panel bolster.

Setting the LAND light switch 'on' energizes the landing light control relay in the Master Control Unit (MCU) completing a 28 VDC circuit from the airplane Main Distribution Bus 1 to the light's ballast located on the firewall. The ballast provides boosted voltage to illuminate the HID lamp.

A 7.5-amp fuse on the Main Distribution Bus 1 in the MCU protects the circuit.

### ***Recognition Lights***

The airplane is equipped with recognition lights on the leading edge of the wing tips. The lights are controlled through the landing light switch on the instrument panel bolster.

28 VDC for recognition light operation is supplied through the 5-amp RECOG LIGHTS circuit breaker on the NON ESS BUS.

## Interior Lighting

Interior lighting for the airplane consists of overhead lights for general cabin lighting, individual lights for the pilots and passengers, and dimmable panel floodlights. The flight instrumentation and avionics equipment lights are dimmable.

### ***Instrument Lights***

Instrument lighting for the airplane includes: Primary Flight and Multifunction Display backlighting and bezel, bolster switch panel, audio panel keys, FMS keyboard, and optionally installed GMC 705 AFCS Control Unit, incandescent lights in the standby instrument bezels, key backlighting and status lighting for the flap and Environmental Control System (ECS) control panels. Associated lighting is adjustable through the INSTRUMENT dimmer control on the instrument panel bolster. The dimmer is OFF when rotated fully counterclockwise, all systems revert to daytime lighting in this position (not full DIM).

In daytime lighting (knob OFF/full counterclockwise):

- Standby instruments, all Avionics system keypads and the bolster switch panel are unlit
- MFD and PFD screen illumination is controlled by automatic photocell (providing full brightness in high light conditions, only slightly reduced by darkness)
- ECS and control panels are backlight and their status lights at maximum intensity

With active dimming (knob moved clockwise), the full bright position (full clockwise) applies maximum illumination to keys and switches, to standby instruments and to status lights, but the PFD/MFD screen illumination is at a substantially reduced level (levels still appropriate for night flight). Maximum screen illumination (appropriate for daytime use) is with the dimmer OFF/full counterclockwise.

The instrument light circuits operate on 28 VDC supplied through the 5-amp CABIN LIGHTS circuit breaker on MAIN BUS.

### ***Panel Flood Lights***

A string of red LEDs mounted under the instrument panel glareshield provide flood lighting for the instrument panel. The lights are controlled through the PANEL dimmer control on the instrument panel bolster.

The panel lights operate on 28 VDC supplied through the 5-amp CABIN LIGHTS circuit breaker on MAIN BUS 1.

### ***Reading Lights***

Individual eyeball-type reading lights are installed in the headliner above each passenger position. Each light is aimed by positioning the lens in the socket and is controlled by a push-button switch located next to the light. The pilot and copilot reading lights are also dimmable through the PANEL lights control on the instrument panel bolster. The reading lights are powered by 28 VDC supplied through the 5-amp CABIN LIGHTS circuit breaker on MAIN BUS 1.

### ***Overhead Dome Light***

General cabin lighting is provided by a dome light located in the headliner at the approximate center of the cabin.

#### ***Serials 0442 thru 1232:***

The dome light is controlled through the OVERHEAD light control on the instrument panel bolster or by the switch next to the light assembly on the ceiling of the airplane. On airplane with OVERHEAD light control on the instrument panel bolster, rotating the knob clockwise from the off position will illuminate the light and control its intensity.

The dome light is powered by 28 VDC supplied through the 5-amp CABIN LIGHTS circuit breaker on MAIN BUS 1.

#### ***Serials 1233 & subs w/o Convenience Lighting:***

The dome light is controlled through the cabin light switch located next to the light assembly on the ceiling of the airplane.

The dome light is powered by 28 VDC supplied through the 5-amp CABIN LIGHTS circuit breaker on MAIN BUS 1.

## Convenience Lighting

### *Serials 1233 & subs w/ Convenience Lighting:*

The convenience lighting option consists of the overhead dome light, overhead baggage compartment lights, interior footwell lights, exterior entry step lights, and a key fob.

#### ***Overhead Dome Light***

General cabin lighting is provided by a dome light located in the headliner at the approximate center of the cabin.

#### ***Overhead Baggage Compartment Lights***

General baggage compartment lighting is provided by lights located in the headliner.

#### ***Footwell Lights***

General floor lighting is provided by footwell lights located throughout the cabin.

#### ***Entry Step Lights***

Illumination of the entry steps is provided by lights located above each step.

Convenience lighting is controlled by the cabin light switch located on the ceiling of the airplane. 28 VDC for convenience lighting is supplied through the 3-amp CONV LIGHTS circuit breaker on the CONS BUS.

#### ***Key Fob***

Remote operation of the door locks is provided by a battery-powered key fob. Refer to [Cabin Doors](#) description in this section.



### ***Convenience Lighting Operation***

When the cabin light switch is in the ON position:

- Dome light and footwell lights will turn on.
- Entry step lights will turn on when either cabin door is opened or the doors are unlocked via the key fob and will turn off when both cabin doors are closed or the doors are locked via the key fob.
- Baggage compartment lights will turn on when baggage door is opened and will turn off when baggage door is closed.

When the cabin light switch is in the OFF position:

- Dome light, baggage compartment lights, footwell lights, and entry step lights will turn off.

When the cabin light switch is in the AUTO position:

- Dome light, footwell lights, and entry step lights will turn on when either cabin door is opened or the doors are unlocked via the key fob and will turn off when both cabin doors are closed or the doors are locked via the key fob.
- Baggage compartment lights will turn on when baggage door is opened and will turn off when baggage door is closed.

When aircraft power is turned off all convenience lighting will turn off after several minutes of illumination.

## Environmental System

• Note •

To facilitate faster cabin cooling, prior to engine start leave the cabin doors open for a short time to allow hot air to escape.

Standard cabin heating and ventilation is accomplished by supplying conditioned air from the heat exchanger for heating and windshield defrost and fresh outside air for ventilation. The environmental system consists of a fresh air inlet in the RH cowl, a heat exchanger around the exhaust system crossover tube, an air mixing chamber, air ducting for distribution, a distribution manifold, a windshield diffuser, crew and passenger air vents, and associated plumbing, controls, actuators, wiring for system flow-selection and temperature control

An optional 3-speed blower fan is available to supplement airflow when ram air may be inadequate such as during ground operation.

28 VDC for Environmental System Control Panel operation is supplied through the 2-amp CABIN AIR CONTROL breaker on MAIN BUS 1.

The optional Blower Fan is powered by 28 VDC supplied through a 15-amp CABIN FAN breaker on A/C BUS 2.

### ***Serials with Optional Air Conditioning System:***

The optionally installed Air Conditioning System is designed to cool the cabin to desired temperature settings and maintain comfortable humidity levels. The system consists of an engine driven compressor, condenser assembly, evaporator assembly, exhaust heat exchanger, fresh air inlet, air-mixing chamber, blower fan, distribution manifold, ducting, windshield diffuser, vent outlets, associated plumbing, controls, actuators, wiring for system flow-selection and temperature control.

28 VDC for Air Conditioner Condenser operation is supplied through the 15-amp A/C COND breaker on A/C BUS 1.

28 VDC for Air Conditioner Compressor operation is supplied through the 5-amp A/C COMPR breaker on A/C BUS 2.

The airplane engine must be running for the air conditioner to operate.

## Distribution

Ventilation and cooling is provided by ducting fresh air from a NACA inlet on the RH cowl to the mixing chamber located on the lower RH portion of the firewall. Depending on operating mode and temperature

selection, the air in the mixing chamber is ducted directly into the distribution system or, if in optional air conditioning mode, is further cooled as it passes through the evaporator assembly located under the front passenger seat. Heating is accomplished by mixing ventilation air from the fresh air inlet with heated air provided by the heat exchanger in the mixing chamber on the firewall. From the mixing chamber - which also controls airflow into the cabin compartment - the conditioned air is forced by ram air pressure or by blower fan into a distribution manifold mounted to the center, aft side of the firewall. The distribution manifold uses butterfly valves to control airflow to the floor and defrost vents. Airflow is ducted directly to all panel air vents.

Crew panel air vents are located inboard on the RH and LH bolster panels and on the outboard section of the instrument panel. The crew floor air vents are mounted to the bottom of each kick plate. The passenger panel air vents are chest high outlets mounted in the armrests integral to the LH and RH cabin wall trim panels. The passenger floor air vents are mounted to the bottom portion of the LH and RH cabin wall trim panels. The windshield diffuser, located in the glareshield assembly, directs conditioned air to the base of the windshield.

## **Heating**

Ram air from the rear ports of the intercoolers is ducted to a heat exchanger surrounding the exhaust system crossover tube. The heated air is then routed to the hot air valve, mounted to the forward side of the firewall, which controls entry of hot air into the cabin distribution system. When the valve is open, the air flows into the cabin mixing chamber. When the valve is closed, the heated air exits into the engine compartment and is exhausted overboard with the engine cooling airflow.

Cabin heat is regulated by controlling the volume of hot air admitted into the distribution system's air mixing chamber. The proportion of heated air to fresh air is accomplished using the temperature selector mounted on the RH instrument panel. For over-temperature protection (the turbocharger bleed air is further heated, under some conditions the hot air source temperature may be in excess of 300°F), the controller monitors mixed air temperature through a sensor downstream of the mixing chamber. If mixed air temperature exceeds duct temperature limit, the hot air flow is reduced and fresh airflow increased until temperature is reduced. Valves are automatically

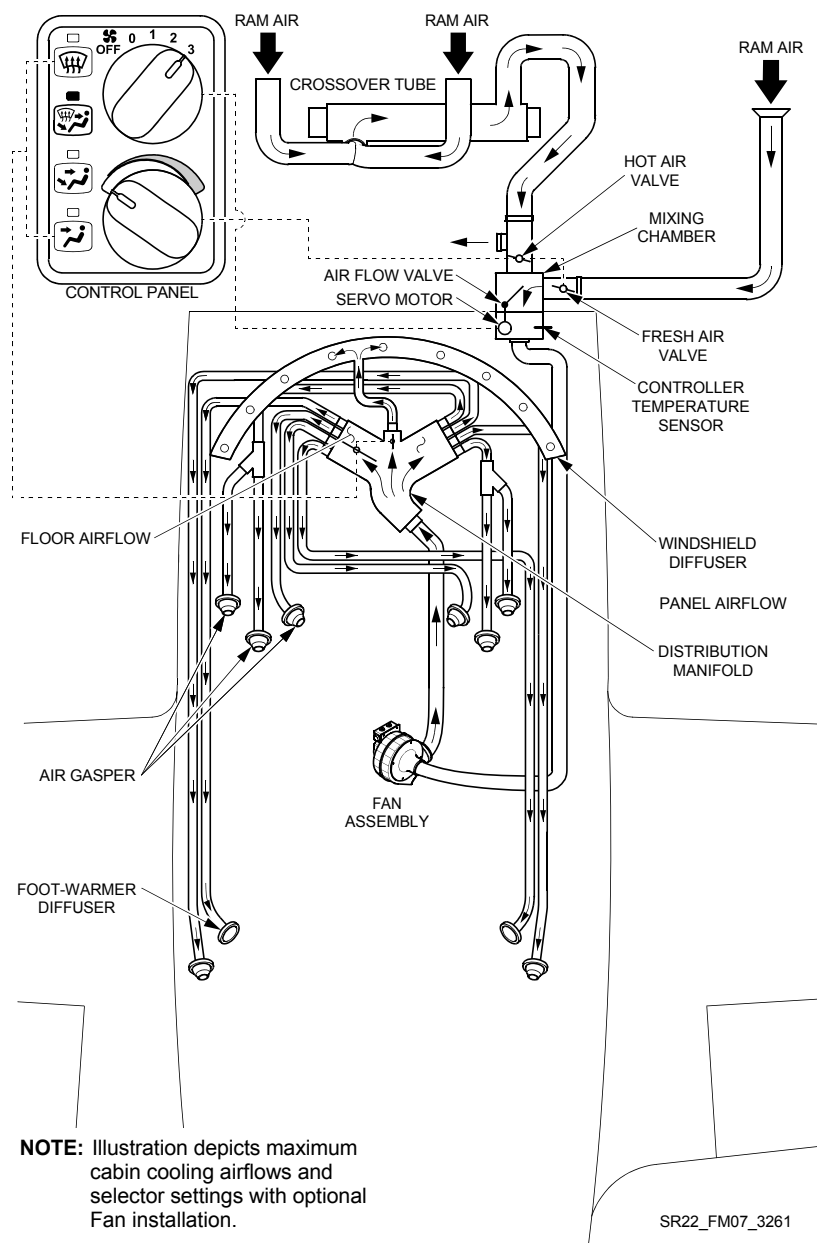
cycled to ensure supply temperature is maintained below duct temperature limits.

Conditioned air can be directed to passengers and/or the windshield diffuser by manipulating the cabin vent selector mounted on the RH instrument panel. The conditioned air enters the cabin through adjustable air vents located in each kick plate and through non-adjustable, floor level vents located in the rear cabin trim side panels. Conditioned air can also be distributed to the windshield diffuser in the glareshield.

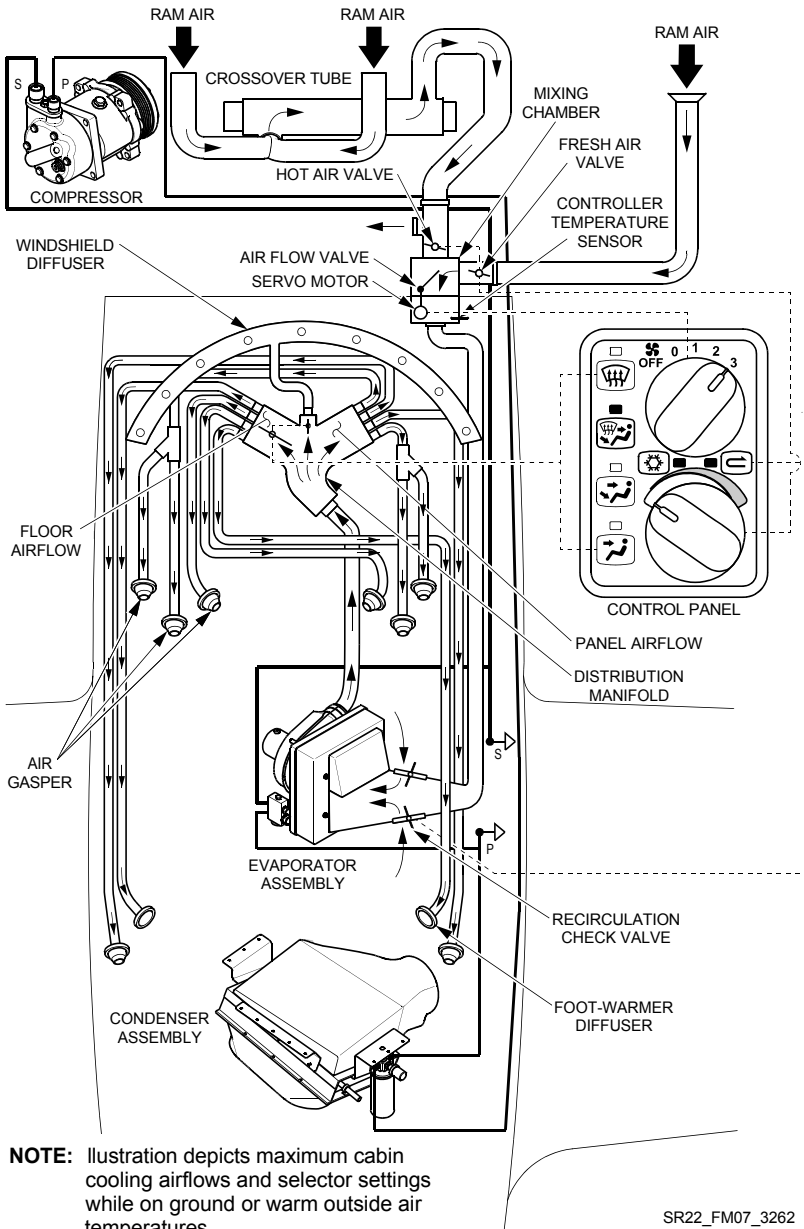
## **Cooling**

Standard cabin cooling is provided by ram air admitted through the NACA inlet on the RH cowl to the fresh air valve, mounted to the forward side of the firewall. When the fresh air valve is open, the air flows into the cabin mixing chamber. When the fresh air valve is closed, the cooled air exits into the engine compartment and is exhausted overboard with the engine cooling airflow.

For airplane with optionally equipped Air Conditioning System, R134A refrigerant enters the engine mounted compressor as a vapor and is pressurized until the heat-laden vapor reaches a point much hotter than the outside air. The compressor then pumps the vapor to the condenser where it cools, changes to a liquid, and passes to the receiver-drier. The receiver-drier's function is to filter, remove moisture, and ensure a steady flow of liquid refrigerant into the evaporator through the expansion valve - a temperature controlled metering valve which regulates the flow of liquid refrigerant to the evaporator. Inside the evaporator, the liquid refrigerant changes state to a gas and in doing so, absorbs heat. The evaporator then absorbs the heat from the air passing over the coils and the moisture from the air condenses and is drained overboard through the belly of the airplane. From the evaporator, the refrigerant vapor returns to the compressor where the cycle is repeated. During normal air conditioning operation, ram air from the fresh air intake flows into the evaporator assembly, is cooled as it passes through the evaporator coils, and is then ducted forward to the distribution manifold. During maximum air conditioning operation - or recirculation mode - the fresh air valve closes and valves in the evaporator assembly open allowing cabin air to be recirculated and further cooled as the air passes through the evaporator coils and ducted forward to the distribution manifold.



**Figure 7-13**  
**Standard Environmental System**



**Figure 7-14**  
**Optional Air Conditioning System**

## Airflow Selection

The airflow selector on the system control panel regulates the volume of airflow allowed into the cabin distribution system. When the airflow selector is moved past the OFF position an electro-mechanical linkage actuates a valve in the mixing chamber on the forward firewall to the full open position. The air is then distributed by either ram air or by an optional blower fan to the distribution manifold mounted to the center, aft side of the firewall. The optional blower fan system includes 0 (ram air), 1 (low fan), 2 (med fan), and 3 (high fan) airflow settings.

## Vent Selection

Air from the distribution manifold is proportioned and directed to passengers and/or the windshield by pressing the cabin vent selector buttons which electrically actuate butterfly valves at the entrances to the windshield diffuser and the cabin floor ducting.

When the Temperature Selector is in the blue "cool" zone, there is continuous ram airflow to the panel and armrest eyeball outlets. Each occupant can control the flow rate from 'off' to maximum by rotating the nozzle.

When the Panel selector button is pushed, both butterfly valves are closed providing maximum airflow to the instrument panel and armrest eyeball outlets.

Pressing the Panel-Foot selector button opens the cabin floor butterfly valve allowing airflow to the rear seat foot warmer diffusers and the front seat outlets mounted to the underside of each kickplate.

Selecting Panel-Foot-Windshield button opens the windshield diffuser butterfly valve which permits shared airflow to the defrosting mechanism and cabin floor outlets.

When the Windshield selector button is pushed the cabin floor butterfly valve is closed providing maximum airflow to the windshield diffuser.

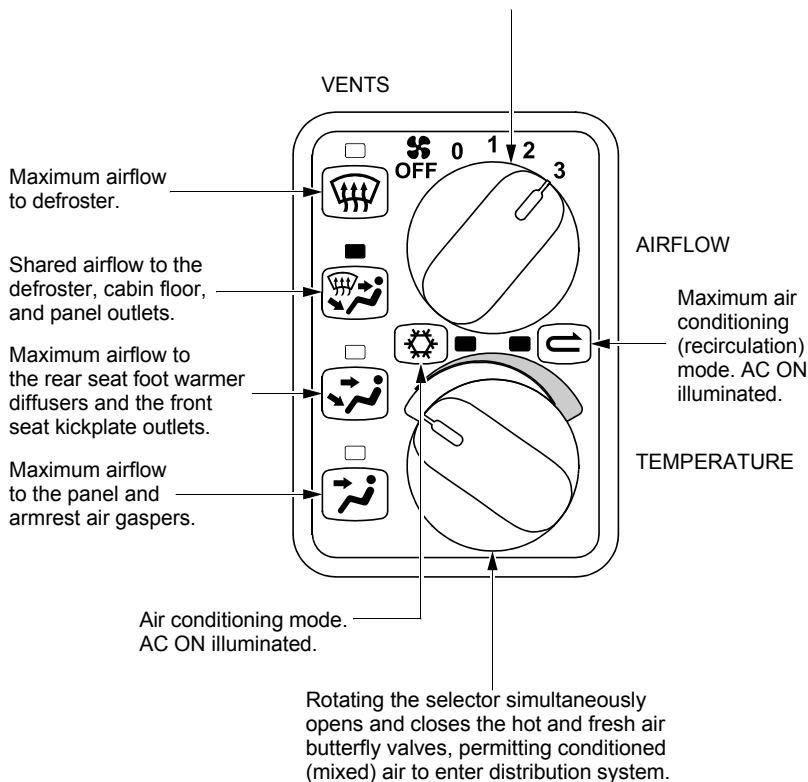
## **Temperature Selection**

The temperature selector is electrically linked to the hot and cold air valves. Rotating the selector simultaneously opens and closes the two valves, permitting hot and cold air to mix and enter the distribution system. Rotating the selector clockwise, permits warmer air to enter the system - counterclockwise, cooler air.

On airplane with the optional Air Conditioning System installed, when the air conditioning button (snowflake) is pushed, the valve on the firewall completely closes and the air-conditioner is activated. When recirculation button is pushed, the fresh air valve completely closes and cabin air is recirculated to provide for maximum air conditioning operation. When the air conditioning system is on and the temperature selector is rotated to the full cool position, recirculating mode can be activated to provide maximum cabin cooling. Air conditioning or recirculating mode is not available when the airflow fan selector is in the "0" position. Recirculating mode is not available unless the air conditioning system is operating.



Rotating the selector controls the volume of airflow allowed into the cabin distribution system through use of an electro-mechanical linkage to a butterfly (hot air) valve in the mixing chamber on the forward firewall. When the airflow selector fan speed is moved to the 1, 2, or 3 position the electro-mechanical linkage actuates the hot air valve to the full open position and the 3-speed blower fan is turned on.



**NOTE:** Illustration depicts settings for Emergency Procedures Smoke and Fume Elimination.  
If source of smoke and fume is firewall forward, turn Airflow Selector OFF.

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**Figure 7-15**  
**Environmental System Operation**

## **Stall Warning System**

The airplane is equipped with an electro-pneumatic stall warning system to provide audible warning of an approach to aerodynamic stall. The system consists of an inlet in the leading edge of the right wing, a pressure switch and associated plumbing, and the avionics system aural warning system.

As the airplane approaches a stall, the low pressure on the upper surface of the wings moves forward around the leading edge of the wings. As the low pressure area passes over the stall warning inlet, a slight negative pressure is sensed by the pressure switch. The pressure switch then provides a signal to cause the warning horn to sound, the red STALL warning CAS annunciation to illuminate, and, if engaged, the autopilot system to disconnect.

The warning sounds at approximately 5 knots above stall with full flaps and power off in wings level flight and at slightly greater margins in turning and accelerated flight.

The system operates on 28 VDC supplied through the 2-amp STALL WARNING circuit breaker on the ESS BUS 2.

## Pitot-Static System

The Pitot-Static system consists of a single heated Pitot tube mounted on the left wing and dual static ports mounted in the fuselage. The Pitot heat is pilot controlled through a panel-mounted switch. An internally mounted alternate static pressure source provides backup static pressure should that the primary static source becomes blocked. Water traps with drains, under the floor in the cabin, are installed at each Pitot and static line low point to collect any moisture that enters the system. The traps should be drained at the annual inspection and when water in the system is known or suspected.

### Pitot Heat Switch

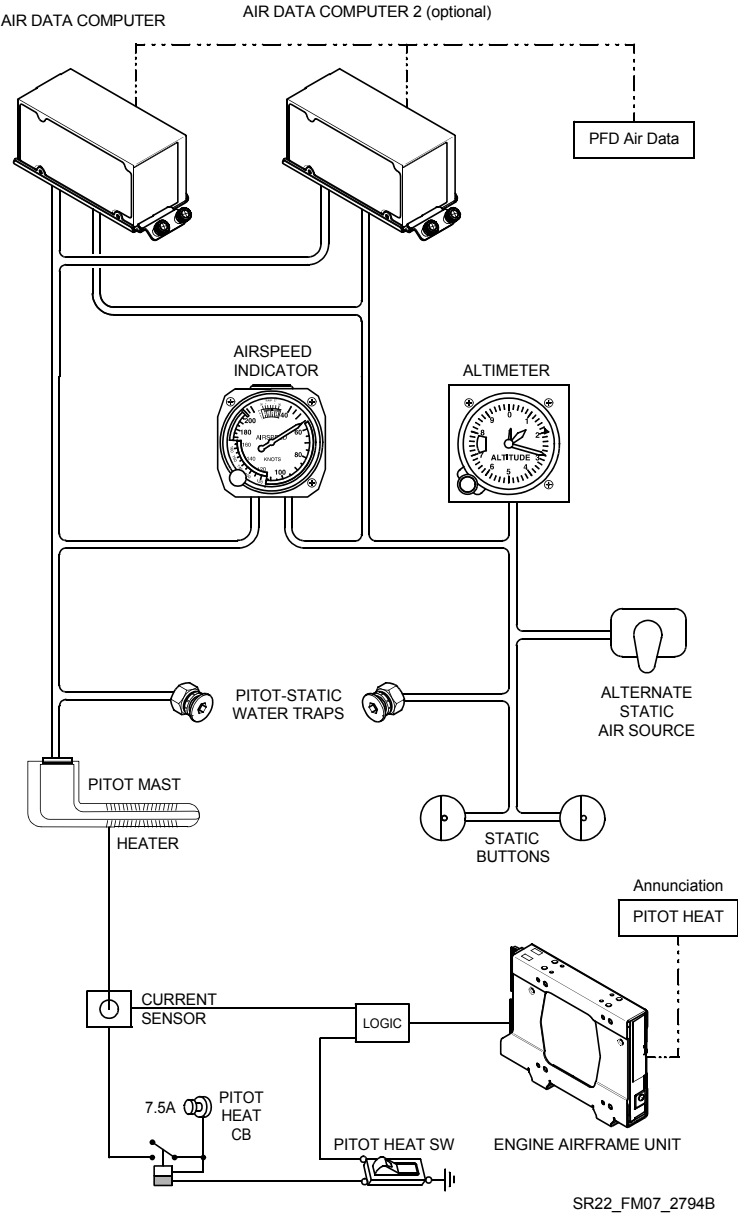
The heated Pitot system consists of a heating element in the Pitot tube, a rocker switch labeled PITOT HEAT, and associated wiring. The switch and circuit breaker are located on the left side of the switch and control panel. When the Pitot heat switch is turned on, the element in the Pitot tube is heated electrically to maintain proper operation in possible icing conditions. The Pitot heat system operates on 28 VDC supplied through the 7.5-amp PITOT HEAT circuit breaker on the NON-ESSENTIAL BUS.

### Pitot Heat Annunciation

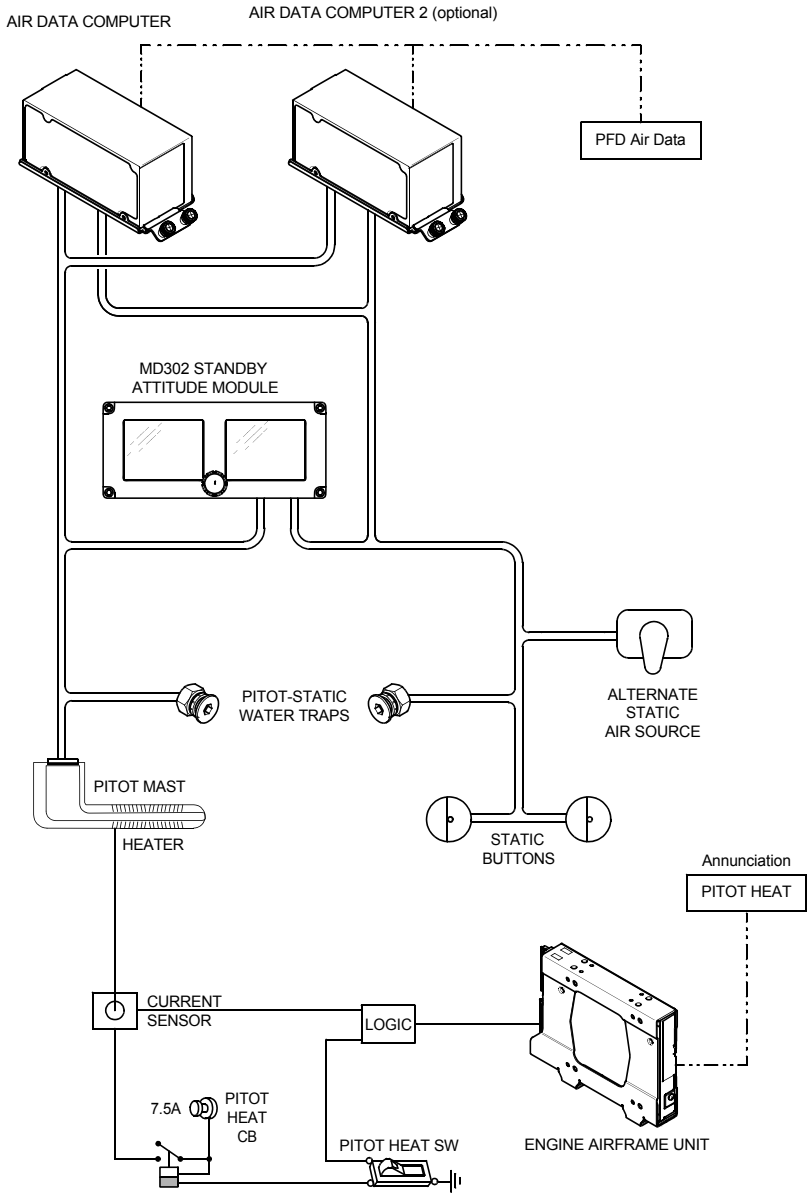
Illumination of the PITOT HEAT FAIL Caution indicates that the Pitot Heat switch is ON and the Pitot heater is not receiving electrical current. Illumination of PITOT HEAT REQD Caution indicates the system detects OAT is less than 41°F (5°C) and Pitot Heat Switch is OFF. A current sensor on the Pitot heater power supply wire provides current sensing.

### Alternate Static Source

An alternate static pressure source valve is installed on the switch and control panel to the right of the pilot's leg. This valve supplies static pressure from inside the cabin instead of the external static port. If erroneous instrument readings are suspected due to water or ice in the pressure line going to the standard external static pressure source, the alternate static source valve should be turned on. Pressures within the cabin will vary with open heater/vents. Whenever the alternate static pressure source is selected, refer to [Section 5: Performance Data](#) for airspeed calibration and altitude corrections to be applied.



**Figure 7-16**  
**Pitot-Static System - Serials w/o MD302 (1 of 2)**



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**Figure 7-16**  
**Pitot-Static System - Serials 0954, 0963 & subs w/ MD302 (2 of 2)**  
**P/N 13772-005**  
**Revision 1**

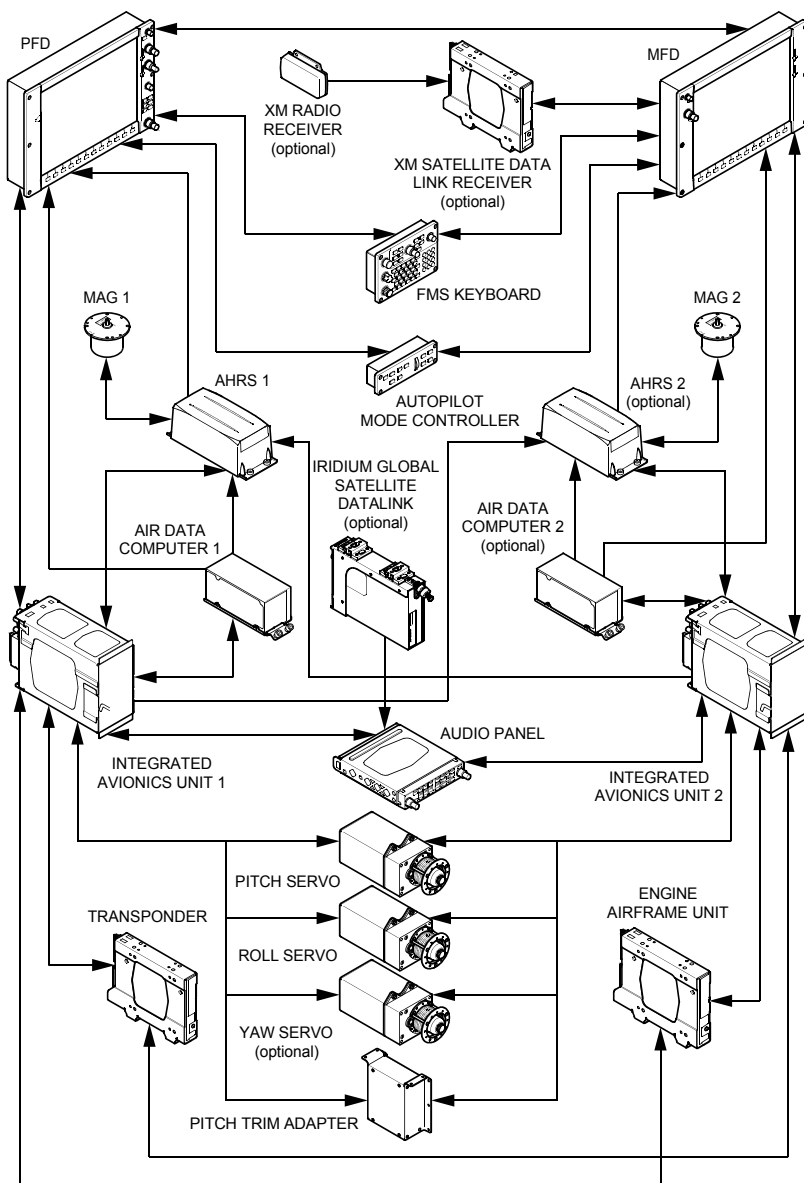
## Avionics

### Perspective Integrated Avionics System

The Perspective Integrated Avionics System provides advanced cockpit functionality and improved situational awareness through the use of fully integrated flight, engine, communication, navigation and monitoring equipment. The system consists of the following components:

- GDU Primary Flight Display (PFD)
- GDU Multifunction Display (MFD)
- GCU 478 Flight Management System Keyboard
- GRS 77 Attitude and Heading Reference System
- GDC 74A Air Data Computer
- GIA 63W Integrated Avionics Units
- GEA 71 Engine Airframe Unit
- GTX 32 Mode A, C, or GTX 33 Mode S, or GTX 33 ES Mode S with Extended Squitter (Optional)
- GMA 350 Audio Panel with Marker Beacon Receiver
- GFC 700 3-Axis Autopilot and GMC 705 Controller
- GSR 56 Iridium Global Satellite Datalink (Optional)
- GDL 69/69A XM Satellite Weather/Radio Receiver (Optional)
  - GRT 10 XM Radio Remote Transceiver (Optional)
  - GRC 10 XM Radio Remote Control (Optional)
- Traffic Advisory System (Optional)
- Weather Information System (Optional)
- Bendix/King KR 87 Automatic Direction Finder (Optional)
- Bendix/King KN 63 Distance Measuring Equipment (Optional)
- Synthetic Vision System (Optional)
- Max Viz Enhanced Vision System (Optional)
- MD302 Standby Attitude Module (Optional)

Refer to the Perspective Integrated Avionics System Pilot's Guide for a detailed description of the system and its operating modes.



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**Figure 7-17**  
**Perspective Integrated Avionics System Schematic**

### ***GDU Primary Flight Display***

The Primary Flight Display, located directly in front of the pilot, is intended to be the primary display of flight parameter information (attitude, airspeed, heading, and altitude) during normal operations. The PFD accepts data from a variety of sources, including the MFD and the Integrated Avionics Units through a high-speed data bus connection. In conjunction with Flight Management System Keyboard, the PFD also controls and displays all communication and navigation frequencies as well as displaying warning/status annunciations on airplane systems. During engine start, reversionary operation (MFD failure), or when the DISPLAY BACKUP switch is selected, engine system information is displayed on the PFD.

Redundant power sources provide 28 VDC for PFD operation. Power is supplied through the 5-amp PFD 1 circuit breaker on the ESS BUS 1 and the 5-amp PFD 2 circuit breaker on MAIN BUS 2. Either circuit is capable of powering the PFD. System start-up is automatic once power is applied. Power-on default brightness is determined by ambient lighting and is user adjustable. Typical alignment time is 60 seconds from battery turn on.

#### **Display Backup Mode**

In the event of a detected display failure, the Integrated Avionics System automatically switches to Display Backup Mode. In Display Backup Mode, all essential flight information from the PFD is presented on the remaining display in the same format as in normal operating mode with the addition of the Engine Indicating System. The change to backup is completely automated and no pilot action is required. However, if the system fails to detect a display problem, Display Backup Mode may be manually activated by pressing the red DISPLAY BACKUP Button. Pressing this button again deactivates Display Backup Mode.

### ***GDU Multifunction Display***

The Multifunction Display, located above the center console, depicts navigation, terrain, lightning, traffic data, NAV/COM frequencies, and annunciation information. All engine data is displayed on a dedicated ENGINE page. When the ENGINE page is not shown, all essential engine information is shown on an engine strip at the edge of the display.

Redundant power sources provide 28 VDC for MFD operation. Power is supplied through the 5-amp MFD 1 circuit breaker on the MAIN BUS



3 and the 5-amp MFD 2 circuit breaker on MAIN BUS 1. Either circuit is capable of powering the MFD. System start-up is automatic once power is applied. Power-on default brightness is determined by ambient lighting and is user adjustable.

### ***GCU 478 Flight Management System Keyboard***

The Flight Management System Keyboard is found on the upper section of the center console and is the primary interface for avionics system data entry, PFD/MFD operation, NAV/COM tuning, and heading, course and altitude selection.

28 VDC for Flight Management System Keyboard operation is supplied through the 5-amp KEYPADS / AP CTRL circuit breaker on MAIN BUS 1.

### ***GRS 77 Attitude and Heading Reference System (AHRS)***

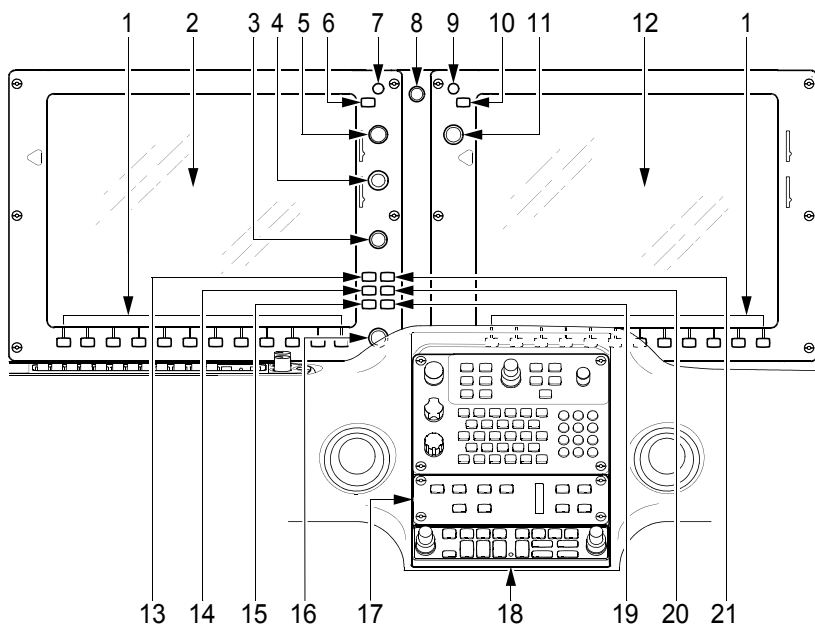
The Attitude and Heading Reference System (AHRS) unit(s), mounted behind the PFD, provide airplane attitude and heading information to both the PFD and the primary Air Data Computer. The AHRS units(s) contain advanced sensors (including accelerometers and rate sensors) and interfaces with the: primary Magnetometer to obtain magnetic field information, the Air Data Computer to obtain air data, and both Integrated Avionics Units to obtain GPS information.

28 VDC for AHRS 1 operation is supplied through the 5-amp AHRS 1 circuit breaker on the ESS BUS 1. If option installed, 28 VDC for AHRS 2 operation is supplied through the 5-amp AHRS 2 circuit breaker on the MAIN BUS 2.

### ***GDC 74A Air Data Computer (ADC)***

The Air Data Computer(s), mounted behind the instrument panel to the right of the MFD, process data from the Pitot/Static system and outside air temperature (OAT) sensor(s). This unit(s) provide pressure altitude, airspeed, vertical speed and OAT information to the Integrated Avionics System, and communicate with the primary PFD, Integrated Avionics Unit, and AHRS units. The Air Data Computer(s) is also connected directly to the Outside Air Temperature probe(s) and Pitot-Static System.

28 VDC for ADC 1 operation is supplied through the 5-amp ADC 1 circuit breaker on the ESS BUS 1. If option installed, 28VDC for ADC 2 operation is supplied through a 5-amp AHRS 2 / ADC 2 circuit breaker on the MAIN BUS 2.

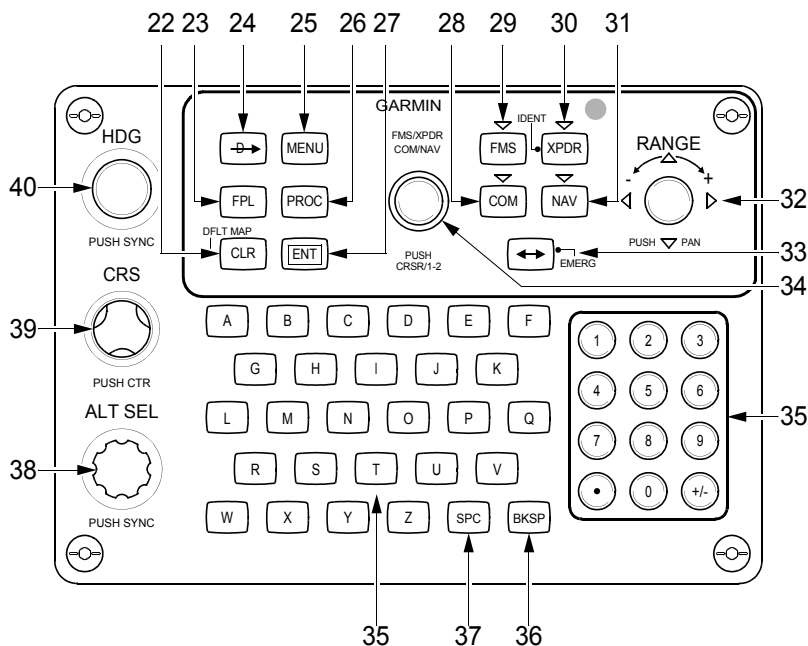


### Legend

- |  |                                      |
|--|--------------------------------------|
| 1. Soft Keys                                     | 11. NAV Transceiver Selection & Tune |
| 2. PFD   | 12. MFD                              |
| 3. PFD Range/Pan Joystick                        | 13. PFD Direct-to-Course             |
| 4. Barometric Pressure                           | 14. PFD Flight Plan Page             |
| 5. COM Transceiver Selection & Tune              | 15. PFD Clear/Cancel Information     |
| 6. COM Frequency Transfer<br>(& 121.5 Emer Tune) | 16. PFD Flight Management System     |
| 7. COM Volume and Squelch                        | 17. GFC 705 Mode Controller          |
| 8. Display Backup Selection                      | 18. Audio Panel                      |
| 9. NAV and ID Audio Volume                       | 19. PFD Enter Key                    |
| 10. NAV Frequency Transfer                       | 20. PFD Procedures                   |
|  | 21. PFD Menu Key                     |

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**Figure 7-18**  
**Perspective Integrated Avionics System (Sheet 1 of 2)**



Flight Management System Keyboard

### Legend

- |  |                                     |
|--|-------------------------------------|
| 22. MFD Clear/Cancel Information (Default Map) | 31. NAV Tuning Mode                 |
| 23. MFD Flight Plan Page                       | 32. MFD Range/Pan Joystick          |
| 24. MFD Direct-to-Course                       | 33. Frequency Transfer (121.5 Tune) |
| 25. MFD Menu                                   | 34. MFD FMS XPDR/NAV/COM Control    |
| 26. MFD Procedures                             | 35. Alphanumeric Keys               |
| 27. MFD Enter Key                              | 36. Backspace Key                   |
| 28. COM Tuning Mode                            | 37. Space Key                       |
| 29. FMS Mode                                   | 38. Altitude Selection (PFD)        |
| 30. Transponder Mode (Ident)                   | 39. Course Selection (HSI)          |
|  | 40. Heading Selection (PFD HSI)     |

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**Figure 7-18**  
**Perspective Integrated Avionics System (Sheet 2 of 2)**

### ***GIA 63W Integrated Avionics Units***

The Integrated Avionics Units, located behind the MFD and instrument panel, function as the main communication hub, linking all Integrated Avionics System components with the PFD. Each Integrated Avionics Unit contains a GPS WAAS receiver, VHF COM/NAV/GS receivers, system integration microprocessors, and flight director if the optional AFCS is installed. The Integrated Avionics Units are not paired together and do not communicate with each other directly.

28 VDC for Integrated Avionics Unit 1 operation is supplied through the 7.5-amp COM 1 and 5-amp GPS NAV GIA 1 circuit breakers on the ESS BUS 1. 28 VDC for Integrated Avionics Unit 2 operation is supplied through the 7.5-amp COM 2 and 5-amp GPS NAV GIA 2 circuit breakers on the MAIN BUS 2.

### ***GEA 71 Engine Airframe Unit***

The Engine Airframe Unit, mounted behind the MFD, receives and processes analog signals from the fuel gaging system, CHT, EGT, MAP, RPM and other sensors and transmits this data to the Integrated Avionics Unit.

28 VDC for Engine Airframe Unit operation is supplied through the 3-amp ENGINE INSTR circuit breaker on the ESS BUS 2.

### ***GTX 32 Transponder***

The GTX 32 solid-state transponder communicates with the primary Integrated Avionics Unit and provides Modes A and C interrogation/reply capabilities. The transponder is controlled via the PFD or Flight Management System Keyboard and is located in the empennage avionics compartment.

28 VDC for Transponder operation is supplied through the 2-amp XPONDER circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a complete description of the system, its operating modes, and additional detailed operating procedures.

### ***GMA 350 Audio Panel with Marker Beacon Receiver***

The 350 Audio Panel, installed on the center console below the Flight Management System Keyboard, integrates NAV/COM digital audio, intercom and marker beacon controls. The VHF communications portion of the unit interfaces with both Integrated Avionics Units to provide external radio communication, receive and demodulate VOR, Localizer, and Glide Slope signals.

28 VDC for Audio Panel operation is supplied through the 5-amp AUDIO PANEL circuit breaker on the AVIONICS bus.

• Note •

COM swap mode is not available in this installation.

For a detailed operating instructions, refer to the GMA 350 Audio Panel Pilot's Guide.

### ***Annunciation and Alert System***

Aircraft annunciations and alerts are displayed in the Crew Alerting System (CAS) window located to the right of the altimeter and VSI. Aircraft annunciations are grouped by criticality and sorted by order of appearance with the most recent message on top. The color of the message text is based on its urgency and required action:

- Warning (red) – Immediate crew awareness and action required.
- Caution (yellow) – Immediate crew awareness and future corrective action required.
- Advisory (white) – Crew awareness required and subsequent action may be required.

In combination with the CAS Window, the system issues an audio alert when specific system conditions are met and an expanded description of the condition is displayed in the Alerts Window located in the lower RH corner of the PFD.

• Note •

For specific pilot actions in response to System Annunciations, refer to [Section 3: Emergency Procedures](#) and [Section 3A: Abnormal Procedures](#).

For additional information on Engine Instrument Markings and Annunciations, refer to [Section 2: Limitations](#).

### ***GFC 700 3-Axis Autopilot and GMC 705 Autopilot Controller***

Refer to latest revision of AFM Supplement 13772-135 [GFC 700 Automatic Flight Control System](#) for operating information.

## Optional Avionics

### ***GTX 33 Mode S and GTX 33 ES Mode S Transponders***

The GTX 33 Mode S and GTX 33 ES Mode S solid-state transponders communicate with the primary Integrated Avionics Unit and provide Modes A, C, and S interrogation/reply capabilities. The GTX 33 ES Mode S transponder includes Extended Squitter ADS-B Out. The transponders are controlled via the PFD or Flight Management System Keyboard and are located in the empennage avionics compartment.

28 VDC for Mode S Transponder operation is supplied through the 2-amp XPONDER circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a complete description of the system, its operating modes, and additional detailed operating procedures.

### ***GSR 56 Iridium Satellite Network Transceiver***

The Iridium Satellite Network Transceiver, mounted in the empennage avionics compartment, communicates with the primary Integrated Avionics Unit and Audio Panel to provide near real-time weather, voice, and data services to the cabin audio system and integrated displays. The GSR 56 receives near real-time satellite weather information for display on the MFD and PFD and can also provide telephone/voice communications and text messaging (SMS) through the Iridium Satellite Network. The voice service is available through the audio panel via the TEL (telephone) input selection. SMS and weather products are displayed on the MFD.

28 VDC for Iridium Satellite Network Transceiver operation is supplied through the 5-amp DATA LINK/WEATHER circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a complete description of the system, its operating modes, and additional detailed operating procedures.

### ***GDL 69/69A XM Satellite Weather and Radio***

The Data Link Satellite Receiver, mounted in the empennage avionics compartment, receives and transmits real-time weather information to the MFD and PFD. If GDL 69A option is installed, this unit also provides digital XM audio entertainment to the cabin audio system via the GRT 10 XM Radio Remote Transceiver, mounted in the empennage avionics compartment and controlled by the GRC 10 Remote Control.

28 VDC for Satellite Datalink Receiver operation is supplied through the 5-amp WEATHER/DATA LINK circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a complete description of the system, its operating modes, and additional detailed operating procedures.

### ***Traffic Advisory System***

The Traffic Advisory System (TAS) advises the pilot of transponder-equipped airplane that may pose a collision threat. TAS information is displayed on the MFD and indicates the relative range, bearing, and altitude of intruder airplane. The Traffic Advisory System consists of a Transmitter Receiver Computer under the LH cockpit seat, and two directional antennas installed on the airplane exterior. The system utilizes inputs from the secondary Integrated Avionics Units via the primary Air Data Computer and is controlled via the MFD or Flight Management System Keyboard.

28 VDC for Traffic Advisory System operation is supplied through the 5-amp TRAFFIC circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a general description of the system and its operating modes.

### ***Stormscope WX-500 Weather Mapping Sensor***

The Stormscope WX-500 System detects electrical discharges associated with thunderstorms and displays the activity on the MFD. The system consists of an antenna located on top of the fuselage and a processor unit mounted under the aft baggage floor. The antenna detects the electrical and magnetic fields generated by intra-cloud, inter-cloud, or cloud to ground electrical discharges occurring within 200 nm of the airplane and sends the “discharge” data to the processor. The processor digitizes, analyzes, and converts the “discharge” signals into range and bearing data and communicates the data to the MFD every two seconds via the secondary Integrated Avionics Unit.

28 VDC for Weather System operation is supplied through the 5-amp WEATHER/DATA LINK circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a general description of the system and its operating modes. If applicable, refer to the L-3 Stormscope WX-500 Weather Mapping Sensor Pilot's Guide for a detailed discussion of the system.

### ***Bendix/King KR 87 Automatic Direction Finder (ADF)***

The KR 87 ADF System is used as a means of identifying positions, receiving low and medium frequency voice communications, homing, tracking, and for navigation on instrument approach procedures. The system consists of an antenna installed on the airplane exterior and the KR 87 receiver which communicates with the Integrated Avionics System via the secondary Integrated Avionics Unit. The HSI Bearing Needle may be configured to indicate ADF tracking and homing information. 28 VDC for ADF System operation is supplied through the 3-amp DME/ADF circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a general description of the system and its operating modes. Refer to the Bendix/King ADF System Pilot's Guide for a detailed discussion of the system.

### ***Bendix/King KN 63 Distance Measuring Equipment (DME)***

The KN 63 DME determines airplane distance to a land-based transponder by sending and receiving pulse pairs - two pulses of fixed duration and separation. The ground stations are typically collocated with VORs. The system consists of an antenna installed on the airplane exterior and the KN 63 receiver which communicates with the Integrated Avionics System via the secondary Integrated Avionics Unit. 28 VDC for ADF System operation is supplied through the 3-amp



DME/ADF circuit breaker on AVIONICS BUS. Refer to the Perspective Integrated Avionics System Pilot's Guide for a general description of the system and its operating modes. Refer to the Bendix/King DME System Pilot's Guide for a detailed discussion of the system.

### ***Synthetic Vision System***

The Synthetic Vision System (SVS) is intended to provide the pilot with enhanced situational awareness by placing a three dimensional depiction of terrain, obstacles, traffic and the desired flight path on the PFD so that proximity and location is more easily understood during instrument scanning. The SVS database is created from a digital elevation model with a 9 arc-sec (approx. 885 ft (270m)) horizontal resolution.

The synthetic vision system is not intended to be used independently of traditional attitude instrumentation. Consequently, SVS is disabled when traditional attitude instrumentation is not available. Otherwise, the traditional attitude instrumentation will always be visible in the foreground with SVS features in the background. The PFD with SVS installed includes:

- Perspective depiction of surrounding terrain,
- Zero pitch line,
- Perspective depiction of runways,
- Perspective depiction of large bodies of water,
- Perspective depiction of obstacles,
- Flight path marker,
- Terrain warning system,
- Field of view depiction on the MFD Navigation Page.

Refer to the Perspective Integrated Avionics System Pilot's Guide for a complete description of the system, its operating modes, and additional detailed operating procedures.

### ***Max Viz Enhanced Vision System***

The Enhanced Vision System is an electro-optical system that uses a Long-Wave Infrared (IR) camera. Infrared is particularly effective at night, smoke, haze, and smog in addition to a broad spectrum of rain, snow, and radiation-type fog. However, penetration is limited during certain environmental conditions associated with heavy rain, heavy snow, coastal fog and most cloud formations. Therefore the EVS is not intended for all atmospheric conditions and may only be used for acquisition of objects normally viewed through the cockpit windows. EVS is an aid to visual acquisitions of:

- Ground vehicles and other ground-based equipment/obstacles,
- Aircraft on taxi-ways and runways,
- Other traffic during takeoff, approach, and landing,
- Runway and taxi lights,
- Runway and terrain features during climb, descent, and low altitude maneuvering.

The EVS sensor, located on the underside of the LH wing, contains a long-wave infrared camera that produces a infrared image and a low-light CMOS camera that produces a visible image. The two images are then combined to produce a single fused image and transmitted directly to the MFD. Upon power-up the Sensor requires approximately 90 seconds to produce a usable image. The image generated is a monochrome image. The hotter an object is the whiter it appears on the display.

28 VDC Enhanced Vision System operation is supplied through the 5-amp EVS CAMERA circuit breaker on MAIN BUS 3. Refer to the Max Viz Enhanced Vision System Pilot's Guide for a detailed discussion of the system. For maintenance information and special precautions to be followed, refer to Section 8, *Enhanced Vision System Sensor Windows (Optional)*.

### ***Approach Baro-VNAV***

While executing an LNAV/VNAV approach with SBAS unavailable, Baro-VNAV is used for vertical approach guidance. This occurs due to any of the following conditions:

- SBAS fails or becomes unavailable prior to final approach fix (FAF)
- the aircraft is outside SBAS coverage
- SBAS is manually disabled on the GPS Status page (To simulate a Baro-VNAV approach, SBAS must be manually disabled prior to activation of the approach procedure.)

Baro-VNAV is also the source of vertical approach guidance if the LNAV/VNAV procedure does not support SBAS vertical guidance.

While Baro-VNAV is being utilized, the Glidepath Indicator appears as a magenta pentagon. If the approach type downgrades past the FAF, "NO GP" is displayed in place of the pentagon.

While executing an LNAV/VNAV approach, between FAF and missed approach point (MAP), excessive deviation indicators appear as vertical yellow lines to indicate an area where the vertical deviation exceeds  $\pm 75$  feet.

#### **Autopilot Interface**

The GFC 700 Automatic Flight Control System uses the GP mode via the APR button to follow Approach Baro-VNAV guidance, as opposed to the VNAV mode via the VNV button. When coupled in GP mode, the GFC 700 will not capture a preselected altitude while tracking a Baro-VNAV glidepath.

#### **Approach Downgrades**

For approaches with minimums that support both SBAS and baro altitude vertical guidance, downgrading or reverting to barometric altitude guidance is allowed prior to 60 seconds before the FAF. If SBAS becomes unavailable after the approach is active but prior to 60 seconds before the FAF, an approach downgrade may be performed (e.g. LPV to LNAV/VNAV) or a vertical source reversion to baro altitude may be performed (e.g. SBAS LNAV/VNAV to baro LNAV/VNAV).

If a loss of SBAS occurs prior to 60 seconds before the FAF, the system will determine whether or not the approach mode can be supported using Baro-VNAV. If Baro-VNAV can be supported, the "APR ADVISORY - SBAS VNAV not available. Using Baro VNAV."

message will be displayed on the PFDs and the vertical deviation indicator (VDI) will be flagged. If SBAS is required for the approach, the approach mode (e.g. LPV) will be shown in amber but the GPS/SBAS VDI will be displayed until 60 seconds prior to the FAF. If the SBAS integrity has not been restored at 60 seconds prior to the FAF, the system will display the “APR DWNGRADE - Apr downgraded. Baro VNAV.” message and flag the VDI.

Once the pilot acknowledges either message by viewing it on the PFD, the VDI will be restored using baro altitude vertical guidance instead of SBAS. There is no downgrade from SBAS to barometric altitude after the FAF or within 60 seconds of the FAF; “LNAV” is the only downgrade option in those cases. For approaches using barometric vertical guidance, downgrade is not allowed; if altitude or temperature data becomes invalid, the vertical deviation will be flagged.

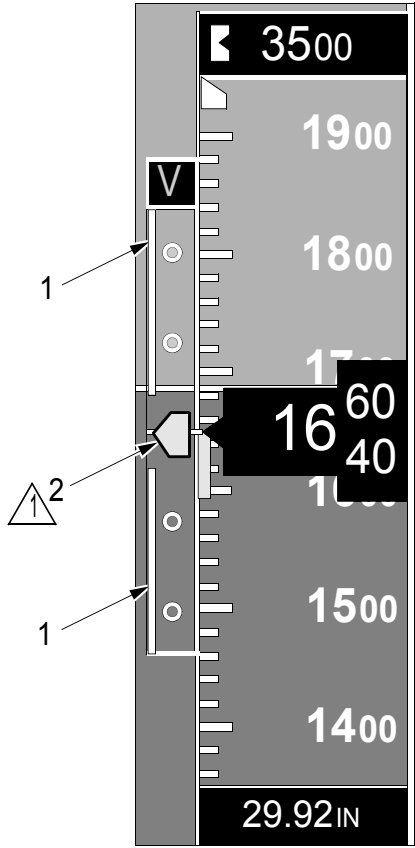
#### Sensor Failures

The Outside Air Temperature (OAT) from the selected-side Air Data Computer will be used. If the OAT becomes invalid the VDI on that side will be flagged as invalid. The crew must select the off-side Air Data Computer sensor and VDI will return regardless of if prior to or after the FAF.


#### Sensor Comparison Annunciation

The temperature compensated altitudes from the pilot and co-pilot side are continuously compared. If a miscompare of greater than 50 feet is detected, the text “VDI MISCOMP” is displayed in the sensor comparison annunciation area on the PFD in black text with an amber background.

When a temperature-compensated altitude is not available for comparison, a “VDI NO COMP” annunciation is posted in comparison annunciation area on the PFD in black text with a white background.



**NOTE**

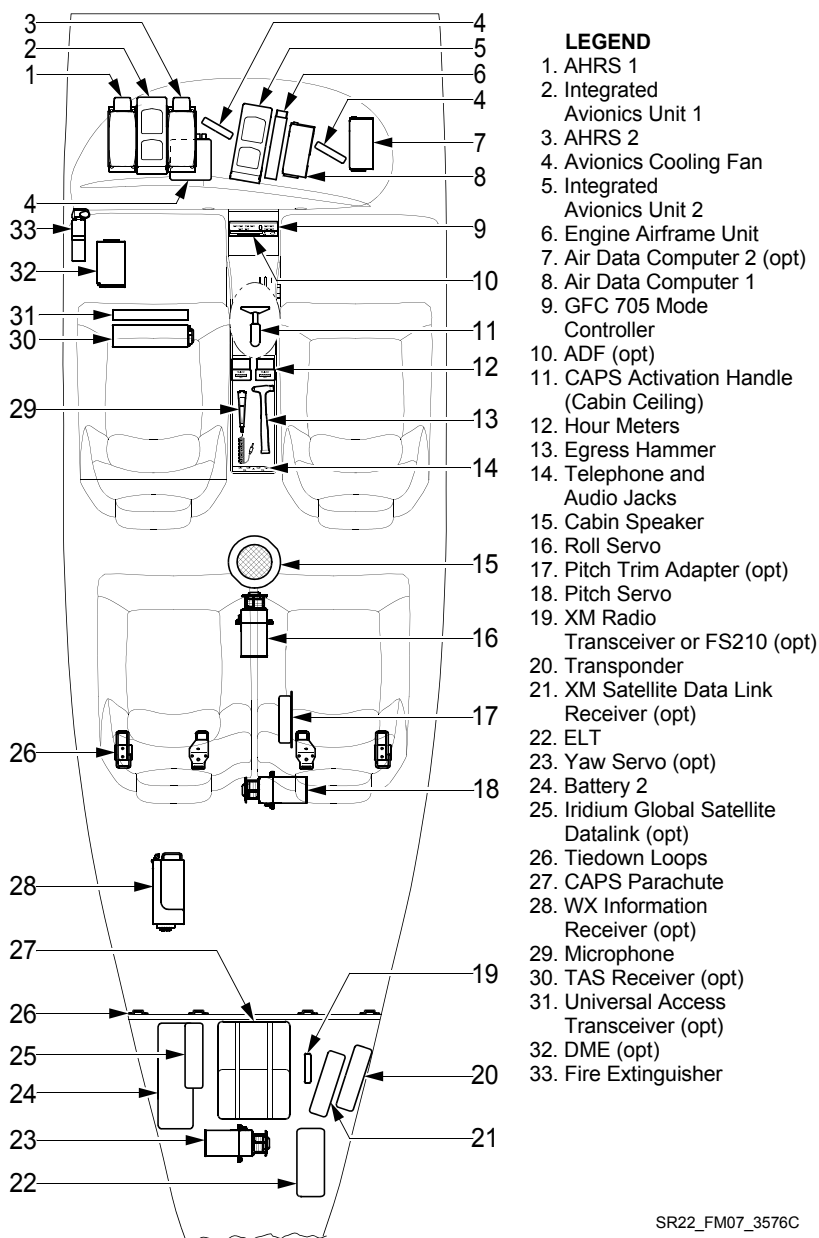
 While Baro-VNAV is being utilized, the Glidepath Indicator appears as a magenta pentagon.

**LEGEND**

- 1.Excessive Deviation Indicator
- 2.Glidepath Indicator

SR22\_FM02\_3686

**Figure 7-19**  
**Baro-VNAV Vertical Deviation Indicator**



SR22\_FM07\_3576C

**Figure 7-20**  
**Equipment Locations**

## Avionics Support Equipment

### *Antennas*

Two rod-type COM antennas are mounted to the airplane's exterior; COM 1 is mounted directly above the passenger compartment, COM 2 is mounted directly below the baggage compartment. These antennas are connected to the two VHF communication transceivers contained in the Integrated Avionics Units.

The optional blade-type DME antenna is mounted on the airplane underside just aft, right of the firewall.

The optional combined loop/sense ADF antenna is mounted to the underside of the airplane just aft of the main wing spar. The antenna combines antenna signals into a single signal input to the ADF receiver.

A sled-type marker beacon antenna is mounted below the baggage compartment floor and provides a signal to the marker beacon receiver located in the audio panel. If the optional air conditioning system is installed this antenna is located below the baggage floor inside of the airplane.

The transponder antenna is located on the bottom side of the airplane, just aft of the baggage compartment bulkhead on the RH side of the airplane.

GPS 1 antenna is mounted directly above the passenger compartment. If the optional XM system is installed, a combination GPS 1/ XM antenna is installed in this location. A combination GPS 2 / Iridium antenna is mounted just forward of the baggage compartment window. These antennas are connected to the two GPS receivers contained in the Integrated Avionics Units.

The optional Traffic System antenna is mounted just above the pilot/ copilot compartment.

If the Avidyne TAS or Garmin GTS 800 Series TAS is installed, a second blade-type antenna is located on the bottom RH side of the airplane just forward of the baggage compartment.

The optional Lightning Detection antenna is mounted directly above the passenger compartment.

The Navigation antenna is mounted to the top of the vertical fin. This antenna provides VOR and glidescope signals to the VHF navigation receivers contained in the Integrated Avionics Units.

### ***Headset and Microphone Installation***

The airplane is equipped with provisions for five Active Noise Reduction (ANR) and three conventional (MIC/HEADPHONES) headsets. Headset jacks for the pilot, front, and rear seat passenger are located in the map case.

The forward headset mics use the remote Push-To-Talk (PTT) switches located on the top of the associated control yoke grip. The rear headsets do not have COM transmit capabilities and do not require PTT switches. Audio to headsets is controlled by the individual audio selector switches on the audio control panel

### ***Audio Input Jack***

The aircraft contains multiple audio input jacks which can be used to connect personal entertainment devices into the cabin sound system. Two 3.5 mm audio input jacks (AUDIO INPUT) are provided on the center console. One jack is located near the convenience outlet for use by the pilot and forward passenger, and the other is located on the aft portion of the center console for the rear passengers.

Distribution of a device connected to the forward jack is through the MUS 1 selection on the audio panel. Distribution of the rear jack is by the MUS 2 selection on the audio panel. A third jack on the audio panel will also accept an entertainment input. A device connected to this jack is distributed by selecting the *Entertainment button* (shown as a phone and music-note symbol) on the audio panel. Audio volume can be controlled by the device itself and can be further refined by the audio panel distribution volume control.

### ***Cell Phone Input Jack***

One 2.5 mm cell phone jack is located on the front of the audio panel and is distributed by selecting the *Entertainment button* (shown as a phone and music-note symbol) on the audio panel. Volume is controlled by the volume selector on the audio panel

### ***Avionics Cooling Fans***

Electric fans provide forced ambient-air cooling for the Integrated Avionics System. A fan located forward of the instrument panel provides ambient air cooling directly to the Integrated Avionics Units. Two additional fans blow air directly onto the heat sinks located on the forward sides of the PFD and MFD.



28 VDC for MFD Fan operation is supplied through the 5-amp AVIONICS FAN 1 circuit breaker on NON-ESSENTIAL BUS. 28 VDC for PFD and Integrated Avionics Unit Fan operation is supplied through the 5-amp AVIONICS FAN 2 circuit breaker on MAIN BUS 2.

## Cabin Features

### Emergency Locator Transmitter

The airplane is equipped with a self-contained emergency locator transmitter (ELT). The transmitter and antenna are installed immediately behind the aft cabin bulkhead, slightly to the right of the airplane centerline. The main transmitter control switch, labeled ON-OFF-ARMED, on the transmitter is in the armed position for normal operations. A remote switch and indicator panel is installed on the left console near the pilot's right knee. If rapid deceleration is detected, the transmitter will repeatedly transmit VHF band audio sweeps at 121.5 MHz and 243.0 MHz approximately 0.5 seconds apart.

The transmitter and antenna are accessible through the avionics bay access panel along the aft portion of the RH fuselage or the lower aft center access panel of baggage compartment. The ELT can be removed from the airplane and used as a personal locating device if it is necessary to leave the airplane after an accident. Eight dated "D" cell alkaline batteries contained within the transmitter unit power the ELT transmitter. The batteries must be replaced at specified intervals based upon the date appearing on the battery (Refer to Airplane Maintenance Manual).

#### ***ELT Remote Switch and Indicator Panel***

The ELT remote switch and indicator panel, located on the left console near the pilot's right knee, provides test and monitoring functions for the ELT. The panel contains a button labeled ON, a button labeled RESET, and a red LED (light). The red light flashes when the ELT is transmitting. The ON button is used to test the unit in accordance with the maintenance manual procedures. The RESET button can be used to cancel an inadvertent transmission. A 6-volt Lithium battery mounted in the panel powers the LED. The battery must be replaced at regular intervals (Refer to Airplane Maintenance Manual).

In the event of an accident:

1. Verify ELT operation by noting that the ELT indicator light on the remote panel is flashing.
2. If possible, access the unit as described below and set the ELT main transmitter control switch ON.

Portable use of ELT:

- a. Remove access at lower aft center of baggage compartment.

- b. Disconnect fixed antenna lead from front of unit.
- c. Disconnect lead from remote switch and indicator unit.
- d. Loosen attach straps and remove transmitter unit and portable antenna.
- e. Attach portable antenna to antenna jack on front of unit.
- f. Set main control switch to ON.
- g. Hold antenna upright as much as possible.

## Fire Extinguisher

A liquefied-gas-type fire extinguisher, mounted on the forward outboard side of the pilot-side footwell, contains Halon 1211/1301 extinguishing agent (*Serials w/o gaged fire extinguisher*), or Halon 1211 extinguishing agent (*Serials w/ gaged fire extinguisher*).

The extinguisher is approved for use on class B (liquid, grease) and class C (electrical equipment) fires. A pin is installed through the discharge mechanism to prevent inadvertent discharge of extinguishing agent.

*Serials w/o gaged fire extinguisher:* The fire extinguisher must be replaced after each use.

*Serials w/ gaged fire extinguisher:* The fire extinguisher must be recharged or replaced after each use.

To operate the extinguisher:

1. Loosen retaining clamp and remove the extinguisher from its mounting bracket.
2. Hold the extinguisher upright and pull the pin.
3. Get back from the fire and aim nozzle at base of fire at the nearest edge.
4. Press red lever and sweep side to side.

### • WARNING •

Halon gas used in the fire extinguisher can be toxic, especially in a closed area. After discharging fire extinguisher, ventilate cabin by opening air vents and unlatching door. Close vents and door after fumes clear.

The extinguisher must be inspected before each flight to ensure that it is available, charged, and operable. The preflight inspection consists

of ensuring that the nozzle is unobstructed, the pin has not been pulled, and the canister has not been damaged.

**Serials w/o gaged fire extinguisher:** The unit should weigh approximately 1.5 lb (0.7 kg). For preflight, charge can be determined by 'hefting' the unit.

**Serials w/ gaged fire extinguisher:** The unit should weigh approximately 2.5 lb (1.1 kg). For preflight, charge can be determined by verifying the gage pressure is in the operable (green) range, or by 'hefting' the unit.

## Hour Meters

The airplane is equipped with two hour meters located inside the armrest storage compartment between the pilot and copilot seats. The #1 hour meter, labeled HOBBS begins recording when the BAT 1 switch is ON and either the ALT 1 or ALT 2 switch is ON. The #2 hour meter records flight time and is labeled FLIGHT. Recording begins when the airplane reaches a speed of approximately 35 KIAS and is controlled by the Integrated Avionics Unit.

28 VDC for hour meter operation is supplied through the 5-amp FUEL QTY circuit breaker on MAIN BUS 1.

## Emergency Egress Hammer

An eight-ounce ball-peen type hammer is located in the center armrest accessible to either front seat occupant. In the event of a mishap where the cabin doors are jammed or inoperable, the hammer may be used to break through the acrylic windows to provide an escape path for the cabin occupants.

## Convenience Outlet(s)

A 12-volt convenience outlet is installed in the center console. The receptacle accepts a standard cigarette-lighter plug. The outlet may be used to power portable entertainment equipment such as CD players and portable radios. Amperage draw through the outlet must not exceed 3.5 amps. Power for the convenience outlet is supplied through the 5-amp 12V DC OUTLET circuit breaker on the MAIN BUS 3.

**Serials 0954, 0963 & subs:** Four Universal Serial Bus-A (USB-A) high-power dedicated charging ports are installed in the center console. Two ports are located near the 12-volt convenience outlet for use by the pilot and forward passenger, and two ports are located on the aft portion of the center console for use by the rear passengers. The ports

comply with USB Battery Charging 1.2 Compliance Plan, and are intended for USB-compatible devices only. There is no data or audio access at the ports. Amperage draw through the each USB charging port must not exceed 2.1 amps. Power for the USB ports is supplied through the 5-amp 12V DC OUTLET circuit breaker on the MAIN BUS 3.

## Cirrus Airframe Parachute System

The airplane is equipped with a Cirrus Airframe Parachute System (CAPS) designed to bring the airplane and its occupants to the ground in the event of a life-threatening emergency. The system is intended to save the lives of the occupants but will most likely destroy the airplane and may, in adverse circumstances, cause serious injury or death to the occupants. Because of this it is important to carefully read Section 3 - Emergency Procedures, *CAPS Deployment* Checklist and Section 10 - Safety Information, *Cirrus Airframe Parachute System (CAPS)* to consider when and how you would use the system.

### • WARNING •

The parachute system can be activated at any time. The solid-propellant rocket flight path is upward from the parachute cover. Stay clear of parachute canister area when airplane is occupied. Do not allow children in the airplane unattended.

## System Description

The CAPS consists of a parachute, a solid-propellant rocket to deploy the parachute, a rocket activation handle, and a harness imbedded within the fuselage structure.

A composite box containing the parachute and solid-propellant rocket is mounted to the airplane structure immediately aft of the baggage compartment bulkhead. The box is covered and protected from the elements by a thin composite cover.

The parachute is enclosed within a deployment bag that stages the deployment and inflation sequence. The deployment bag creates an orderly deployment process by allowing the canopy to inflate only after the rocket motor has pulled the parachute lines taut.

The parachute itself is a round canopy equipped with a slider, an annular-shaped fabric panel with a diameter significantly less than the open diameter of the canopy. The slider has grommets spaced around its perimeter. The canopy suspension lines are routed through these grommets so that the slider is free to move along the suspension lines. Since the slider is positioned at the top of the suspension lines near the canopy, at the beginning of the deployment sequence the slider limits the initial diameter of the parachute and the rate at which the parachute inflates. As the slider moves down the suspension lines the canopy inflates.

A three-point harness connects the airplane fuselage structure to the parachute. The aft harness strap is stowed in the parachute canister and attached to the structure at the aft baggage compartment bulkhead. The forward harness straps are routed from the canister to firewall attach points just under the surface of the fuselage skin. When the parachute deploys, the forward harness straps pull through the fuselage skin covering from the canister to the forward attach points.

## Activation Handle

CAPS is initiated by pulling the CAPS Activation T-handle installed in the cabin ceiling on the airplane centerline just above the pilot's right shoulder. A placarded cover, held in place with hook and loop fasteners, covers the T-handle and prevents tampering with the control. The cover is removed by pulling the black tab at the forward edge of the cover.

Pulling the activation T-handle will activate the rocket and initiate the CAPS deployment sequence. To activate the rocket, two separate events must occur:

1. Pull the activation T-handle from its receptacle. Pulling the T-handle removes it from the o-ring seal that holds it in place and takes out the slack in the cable (approximately two inches (5 cm) of cable will be exposed). Once the slack is removed, the T-handle motion will stop and greater force will be required to activate the rocket.
2. Clasp both hands around activation T-handle and pull straight downward with a strong, steady, and continuous force until the rocket activates. A chin-up type pull works best. Up to 45.0 pounds (20.4 Kg) force, or greater, may be required to activate the rocket. The greater force required occurs as the cable arms and then releases the igniter switch plunger activating the electronic igniter.

• Note •

Jerking or rapidly pulling on the activation T-handle greatly increases the pull forces required to activate the rocket.

Attempting to activate the rocket by pushing the activation T-handle forward and down limits the force that can be applied. Pulling the activation T-handle straight down generates the greatest force.

A maintenance safety pin is provided to ensure that the activation handle is not pulled during maintenance. However, there may be some circumstances where an operator may wish to safety the CAPS system; for example, the presence of unattended children in the airplane, the presence of people who are not familiar with the CAPS activation system in the airplane, or during display of the airplane.

The pin is inserted through the handle retainer and barrel locking the handle in the “safe” position. A “Remove Before Flight” streamer is attached to the pin.

**• WARNING •**

After maintenance has been performed or any other time the system has been safetied, operators must verify that the pin has been removed before further flight.

## **Deployment Characteristics**

When the rocket launches, the parachute assembly is extracted outward due to rocket thrust and rearward due to relative wind. In approximately two seconds the parachute will begin to inflate.

When air begins to fill the canopy, forward motion of the airplane will dramatically be slowed. This deceleration increases with airspeed but in all cases within the parachute envelope should be less than 3 g's. During this deceleration a slight nose-up may be experienced, particularly at high speed; however, the rear riser is intentionally snubbed short to preclude excessive nose-up pitch. Following any nose-up pitching, the nose will gradually drop until the airplane is hanging nose-low beneath the canopy.

Eight seconds after deployment, the rear riser snub line will be cut and the airplane tail will drop down into its final approximately level attitude. Once stabilized in this attitude, the airplane may yaw slowly back and forth or oscillate slightly as it hangs from the parachute. Descent rate is expected to be less than 1700 feet per minute with a lateral speed equal to the velocity of the surface wind. In addition, surface winds may continue to drag the airplane after ground impact.



• Caution •

Ground impact is expected to be equivalent to touchdown from a height of approximately 13 feet. While the airframe, seats and landing gear are designed to accommodate this stress, occupants must prepare for it in accordance with Section 3 - *CAPS Deployment* Checklist.

• Note •

The CAPS is designed to work in a variety of airplane attitudes, including spins. However, deployment in an attitude other than level flight may yield deployment characteristics other than those described above.

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## ***Section 8: Handling and Servicing***

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## Introduction

This section provides general guidelines for handling, servicing and maintaining your aircraft. In order to ensure continued safe and efficient operation of your airplane, keep in contact with your Authorized Cirrus Service Center to obtain the latest information pertaining to your aircraft.

## Operator's Publications

The FAA Approved Airplane Flight Manual and Pilot's Operating Handbook (POH) is provided at delivery. Additional or replacement copies may be obtained from Cirrus Design.

## Service Publications

The following service publications are available for purchase from Cirrus Design:

- Airplane Maintenance Manual (AMM) – Maintenance Manual divided into chapters as specified by GAMA and ATA covering inspection, servicing, maintenance, troubleshooting, and repair of the airplane structure, systems, and wiring. Revision Service for this manual is also available. A current copy of the AMM is provided at delivery.
- Engine Operators and Maintenance Manual – Cirrus Design provides a Teledyne Continental Engine Operator's and Maintenance Manual at the time of delivery. Engine and engine accessory overhaul manuals can be obtained from the original equipment manufacturer.
- Avionics Component Operator and Maintenance Manuals — Cirrus Design provides all available operator's manuals at the time of delivery. Maintenance manuals, if available, may be obtained from the original equipment manufacturer.

Cirrus Design publishes Service Bulletins, Service Advisories, and Service Information Letters. Copies can be obtained from Cirrus Design at [www.cirrusaircraft.com](http://www.cirrusaircraft.com).

- Service Bulletins – are of special importance. When a Service Bulletin is published affecting your airplane, comply with it promptly.
- Service Advisories – are used to notify you of optional Service Bulletins, supplier Service Bulletins or Service Information

Letters affecting your airplane, and maintenance data or corrections not requiring a Service Bulletin. Give careful attention to the Service Advisory information.

## **Obtaining Publications**

Pilot's Operating Handbooks and aircraft service publications can be obtained from Cirrus Design at [www.cirrusaircraft.com](http://www.cirrusaircraft.com), or the Cirrus Connection at [www.cirrusconnection.com](http://www.cirrusconnection.com).

## Towing

The airplane may be moved on the ground by the use of the nose wheel steering bar that is stowed in the rear baggage compartment or by power equipment that will not damage or excessively strain the nose gear assembly. The steering bar is engaged by inserting it into lugs just forward of the nose wheel axle.

### • Caution •

While pushing the aircraft backward, the tow bar must be installed to keep the nose wheel from turning abruptly.

Do not use the vertical or horizontal control surfaces or stabilizers to move the airplane. If a tow bar is not available, use the wing roots as push points.

Do not push or pull on control surfaces or propeller to maneuver the airplane.

Do not tow the airplane when the main gear is obstructed with mud or snow.

If the airplane is to be towed by vehicle, do not turn the nose wheel more than 90 degrees either side of center or structural damage to the nose gear could result.

1. Refer to Section 1, *Airplane Three View* for turning radius clearances. Be especially cognizant of hangar door clearances.
2. Insert tow bar into the lugs just forward of the nose wheel axle.
3. Release parking brake and remove chocks.
4. Move airplane to desired location.
5. Install chocks.
6. Remove tow bar.

To obtain a minimum radius turn during ground handling, the airplane may be rotated around either main landing gear by pressing down on the fuselage just forward of the horizontal stabilizer to raise the nose wheel off the ground.

## Taxiing

Before attempting to taxi the airplane, ground personnel should be instructed and authorized by the owner to taxi the airplane. Instruction should include engine starting and shutdown procedures in addition to taxi and steering techniques.

### • Caution •

Verify that taxi and propeller wash areas are clear before beginning taxi.

Do not operate the engine at high RPM when running up or taxiing over ground containing loose stones, gravel, or any loose material that may cause damage to the propeller blades.

Taxi with minimum power needed for forward movement. Excessive braking may result in overheated or damaged brakes.

1. Remove chocks.
2. Start engine in accordance with Starting Engine procedure.
3. Release parking brake.
4. Advance throttle to initiate taxi. Immediately after initiating taxi, apply the brakes to determine their effectiveness. During taxiing, use differential braking to make slight turns to ascertain steering effectiveness.

### • Caution •

Observe wing clearance when taxiing near buildings or other stationary objects. If possible, station an observer outside the airplane.

Avoid holes and ruts when taxiing over uneven ground.

5. Taxi airplane to desired location.
6. Shut down airplane and install chocks and tie-downs in accordance with Shutdown procedure.



## ***Raise Airplane***

### **• Caution •**

Do not jack the aircraft outside or in open hangar with winds in excess of 10 mph.

The empty CG is forward of the wing jacking points. To prevent airplane from tipping forward during maintenance or jacking, use a weighted tailstand (300-lb minimum) attached to the tail tiedown.

Jacks must be used in pairs. Do not attempt to jack only one side of aircraft. Keep the airplane as level as possible when jacking.

1. Position airplane on a hard, flat, level surface.
2. Remove main gear fairings. (Refer to AMM 32-10)
3. Remove and stow tie-down rings from wings.
4. Attach a weighted tailstand to tail tiedown ring.
5. Position jacks and jack points for jacking. Insert jack point into wing tiedown receptacle. Holding the jack point in place, position the jack under the point and raise the jack to firmly contact the jack point. Repeat for opposite jacking point.
6. Raise airplane no more than required for maintenance being performed.
7. Raise the airplane keeping the airplane as level as possible.
8. Secure jack locks.

## ***Lower Airplane***

1. Release pressure on all jacks simultaneously to keep airplane as level as possible.
2. Remove jacks, jack points, and tailstand. Stow points in baggage compartment.
3. Install tiedown rings.
4. Install main gear fairings. (Refer to AMM 32-10)

## Servicing

### Landing Gear Servicing

*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21:* The main landing gear wheel assemblies use 15 x 6.00 x 6 tires and tubes. The nose wheel assembly uses a 5.00 x 5 tire and tube.

*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs:* The main landing gear wheel assemblies use 15 x 6.00 x 6 tubeless tires. The nose wheel assembly uses a 5.00 x 5 tubeless tire.

*All Serials:* Always keep tires inflated to the rated pressure to obtain optimum performance and maximum service. The landing gear struts do not require servicing. With the exception of replenishing brake fluid, wheel and brake servicing must be accomplished in accordance with AMM procedures.

### Brake Servicing

#### ***Brake Replenishing***

*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21:* The brake system is filled with MIL-H-5606 hydraulic brake fluid.

*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs:* The brake system is filled with MIL-PRF-87257 hydraulic brake fluid.

*All Serials:* The fluid level should be checked at every oil change and at the annual/100-hour inspection, replenishing the system when necessary. The brake reservoir is located on the right side of the battery support frame. If the entire system must be refilled, refer to the AMM.

To replenish brake fluid:

1. Chock tires and release parking brake.
2. Remove top engine cowling to gain access to hydraulic fluid reservoir.
3. Clean reservoir cap and area around cap before opening reservoir cap.
4. Remove cap and add appropriate hydraulic fluid as necessary to fill reservoir.
5. Install cap, inspect area for leaks, and then install and secure engine cowling.

## Tire Inflation

For maximum service from the tires, keep them inflated to the proper pressure. When checking tire pressure, examine the tires for wear, cuts, nicks, bruises and excessive wear.

To inflate tires:

1. Remove inspection buttons on wheel pants to gain access to valve stems. It may be necessary to move airplane to get valve stem aligned with the access hole.
2. Remove valve stem cap and verify tire pressure with a dial-type tire pressure gage.
3. *Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21:* Inflate nose tire to 40+/-2 psi (276 kPa) and main wheel tires to 62+2/-0 psi (427 kPa).

### • Caution •

*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs:* The LH and RH main wheel tire pressures must be within 20 psi of each other to ensure the load is evenly distributed between the main wheels.

4. *Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs:* Inflate nose tire to 40 - 90 psi (276 - 621 kPa) and main wheel tires to 62 - 112 psi (427 - 772 kPa).
5. Replace valve stem cap and inspection buttons.

All wheels and tires are balanced before original installation and the relationship of tire, tube, and wheel should be maintained upon reinstallation. In the installation of new components, it may be necessary to rebalance the wheels with the tires mounted. Unbalanced wheels can cause extreme vibration in the landing gear.

## Propeller Servicing

The spinner and backing plate should be cleaned and inspected for cracks frequently. Before each flight the propeller should be inspected for nicks, scratches, and gouges. If found, they should be repaired as soon as possible by a rated mechanic, since a nick or scratch causes an area of increased stress which can lead to serious cracks or the loss of a propeller tip.

Propeller blades are painted with a durable specialized coating that is resistant to abrasion. If this coating becomes eroded, it is necessary to

repaint the blades to provide proper erosion protection. Painting should be performed by an authorized propeller repair station.

It is permissible to perform a blade touch-up with aerosol paint in accordance with the appropriate revision of the Hartzell Propeller Owner's Manual (p/n 145).

## Oil Servicing

The oil capacity of the Teledyne Continental IO-550-K engine is 8 quarts. It is recommended that the oil be changed every 50 hours and sooner under unfavorable operating conditions. The following grades are recommended for the specified temperatures at sea level (SL):

Ambient Air Temperature (SL)	Single Viscosity	Multi-Viscosity
All Temperatures	—	20W-60 20W-50 15W-50
Below 40°F	SAE 30	10W-30 20W-60 20W-50 15W-50
Above 40°F	SAE 50	20W-60 20W-50 15W-50

An oil filler cap and dipstick are located at the left rear of the engine and are accessible through an access door on the top left side of the engine cowl.

### • Caution •

The engine should not be operated with less than six quarts of oil. Seven quarts (dipstick indication) is recommended for extended flights.

To check and add oil:

1. Open access door on upper left-hand side of cowl. Pull dipstick and verify oil level.
2. If oil level is below 6 quarts (5.7 liters), remove filler cap and add oil through filler as required to reach 6-8 quarts (5.7-7.6 liters).
3. Verify oil level and install dipstick and filler cap.
4. Close and secure access panel.

### Approved Oils

*Engine Break-In:* For first 25 hours of operation or until oil consumption stabilizes use straight mineral oil conforming to MIL-C-6529. If engine oil must be added to the factory installed oil, add only MIL-C-6529 straight mineral oil.

*After Engine Break-In:* Use only oils conforming to Teledyne Continental Specification SAE J 1899 (Ashless Dispersant Lubrication Oil).

<b>Product</b>	<b>Supplier</b>
Aeroshell (R) W	Shell Australia
Aeroshell Oil W Aeroshell Oil W 15W-50 Anti-Wear Formulation Aeroshell 15W50	Shell Canada Ltd.
Aeroshell Oil W Aeroshell Oil W 15W-50 Anti-Wear Formulation Aeroshell 15W50	Shell Oil Company
Aviation Oil Type A	Phillips 66 Company
BP Aero Oil	BP Oil Corporation
Castrolaero AD Oil	Castrol Ltd. (Australia)
Chevron Aero Oil	Chevron U.S.A. Inc.
Conoco Aero S	Continental Oil
Delta Avoil	Delta Petroleum Co.
Exxon Aviation Oil EE	Exxon Company, U.S.A.
Mobil Aero Oil	Mobil Oil Company
Pennzoil Aircraft Engine Oil	Pennzoil Company
Quaker State AD Aviation Engine Oil	Quaker State Oil & Refining Co.
Red Ram Aviation Oil 20W-50	Red Ram Ltd. (Canada)
Sinclair Avoil	Sinclair Oil Company
Texaco Aircraft Engine Oil – Premium AD	Texaco Inc.
Total Aero DW 15W50	Total France
Turbonycoil 3570	NYCO S.A.
Union Aircraft Engine Oil HD	Union Oil Company of California

**Figure 8-1**  
**Approved Oils**

If sampling reveals contamination, the gascolator and tank drains must be sampled again repeatedly until all contamination is removed. It is helpful to gently rock the wings and lower the tail slightly to move contaminants to the drain points for sampling. If after repeated samplings (three or more), evidence of significant contamination remains, do not fly the airplane until a mechanic is consulted, the fuel system is drained and purged, and the source of contamination is determined and corrected.

If sampling reveals the airplane has been serviced with an improper fuel grade, do not fly the airplane until the fuel system is drained and refueled with an approved fuel grade.

To help reduce the occurrence of contaminated fuel coming from the supplier or fixed based operator, pilots should assure that the fuel supply has been checked for contamination and that the fuel is properly filtered. Also, between flights, the fuel tanks should be kept as full as operational conditions permit to reduce condensation on the inside of fuel tanks.

### ***Draining Fuel System***

The bulk of the fuel may be drained from the wing fuel tanks by the use of a siphon hose placed in the cell or tank through the filler neck. The remainder of the fuel may be drained by opening the drain valves. Use the same precautions as when refueling airplane. Refer to the AMM for specific procedures.

## **Battery Service**

The aircraft is delivered with a maintenance-free, rechargeable, sealed, lead acid primary battery. Battery #1 is mounted to the forward right side of the firewall and access is gained by removing the upper cowl. The battery vent is connected to an acid resistant plastic tube that vents gases and electrolyte overflow overboard.

A capacity check must be performed at initial 24 months or 1200 hours in service and then every 12 months or 200 hours thereafter. Refer to the AMM for additional information on Battery #1 Overhaul and Replacement Schedule and Scheduled Maintenance Checks.

### **• Note •**

*For aircraft equipped with conventional lead acid battery requiring periodic electrolyte level check:* Refer to the AMM for information on Battery Overhaul and Replacement Schedule and Scheduled Maintenance Checks.

Battery #2 is a maintenance-free, rechargeable, sealed, lead acid battery. Mounted in the empennage just aft of bulkhead 222, there is no need to check the specific gravity of the electrolyte or add water to these batteries during their service life. Refer to the AMM for Overhaul and Replacement Schedule.

The external power receptacle is located on the left side of the fuselage just aft of the firewall. Refer to the AMM for battery servicing procedures.

## Oxygen System Servicing

### • Caution •

To preclude the possibility of fire by spontaneous combustion, oil, grease, paint, hydraulic fluid, and other flammable material should be kept away from oxygen equipment.

Service the oxygen system per the appropriate revision of the Precise Flight Instructions for Continued Airworthiness for the Cirrus SR22/SR22T Built-In Oxygen System, STC number SA01708SE, document number 102NPMAN0003.

## Key Fob Battery Replacement

*Serials 1233 & subs w/ Convenience Lighting:*

If the key fob does not function properly at normal range, the battery should be replaced. To replace the key fob battery:

1. Using a thin flat object, pry the top and bottom halves of the key fob apart.
2. Remove and replace the battery with a new CR2032, or equivalent, 3-volt battery. Install the new battery with the positive side (+) facing up, away from the circuit board.
3. Press the top and bottom halves of the key fob back together.



## Care of Graphics

Graphics require care similar to any fine paint finish. Use high quality products designed specifically for use on automobile finishes. Use products in accordance with the manufacturer's instructions.

### ***Exposure to Environmental Conditions***

Graphics, like paint, are degraded by prolonged exposure to sun and atmospheric pollutants. Store the aircraft in a hangar, under a cloth cover, or in shaded area whenever possible. Protect the aircraft from dew and rain which may contain acidic pollutants (commonly found in large metropolitan areas).

#### **• Caution •**

If graphics start to discolor or turn brown as a result of exposure to acidic pollution, immediately have a professional remove the graphic from the aircraft to avoid staining the underlying paint.

### ***Regular Washing***

Wash graphics whenever the aircraft appears dirty. Contaminants allowed to remain on the exterior may be more difficult to remove.

1. Rinse off as much dirt and grit as possible with a spray of water.
2. Clean graphic with a wet, non-abrasive detergent such as 3M™ Car Wash Soap 39000, Meguiar's NXT Generation® Car Wash, or Deep Crystal® Car Wash, and a soft, clean cloth or sponge.
3. Rinse thoroughly with clean water.
4. To reduce water spotting, immediately use a silicone squeegee to remove water.
5. Dry with a clean microfiber cloth.

### ***Pressure Washing***

Although hand washing is preferred, pressure washing may be used when necessary to remove dirt and contaminants. Pressure washing must be performed in accordance with the following procedure:

1. Ensure the water pressure is less than 2000 psi (14 MPa).
2. Ensure water temperature is less than 180 °F (82 °C).
3. Use a spray nozzle with a 40 degree wide angle spray pattern.

• **Caution** •

Holding the nozzle of a pressure washer at an angle less than 90 degrees to the graphic may lift the edges of the graphic.

4. Keep the spray nozzle perpendicular to the graphic, and at a distance of at least 1 foot (30 cm).
5. To reduce water spotting, immediately use a silicone squeegee to remove water.
6. Dry with a clean microfiber cloth.

***Removing Difficult Contaminants***

Difficult contaminants such as bugs, bird droppings, or tree sap may require spot cleaning.

• **Caution** •

To prevent scratching the graphic, refrain from rough scrubbing and the use of abrasive tools.

1. Soften contaminants by soaking with hot, soapy water for several minutes.
2. Rinse thoroughly with clean water.
3. To reduce water spotting, immediately use a silicone squeegee to remove water.
4. Dry with a clean microfiber cloth.

• **Caution** •

Initially test cleaning products on an inconspicuous area of the graphic to verify they will not cause damage.

5. If further cleaning is needed, one of the following products may be used: Meguiar's Gold Class™ Bug and Tar Remover, 3M™ Citrus Base Cleaner, a mixture of two parts isopropyl alcohol to one part water (mix ratio 2:1), or denatured alcohol.
6. Immediately rinse off all residue with clean water.
7. To reduce water spotting, immediately use a silicone squeegee to remove water.
8. Dry with a clean microfiber cloth.

### ***Cleaning Fuel Spills***

**• Caution •**

Immediately clean fuel spills to avoid degrading the vinyl and adhesive used in the graphic.

1. Wipe off spilled fuel.
2. Clean graphic with a wet, non-abrasive detergent such as 3M™ Car Wash Soap 39000, Meguiar's NXT Generation® Car Wash, or Deep Crystal® Car Wash, and a soft, clean cloth or sponge.
3. Rinse thoroughly with clean water.
4. To reduce water spotting, immediately use a silicone squeegee to remove water.
5. Dry with a clean microfiber cloth.

## Graphic Restoration

If typical cleaning methods fail to produce satisfactory results, refer to the recommended restoration products and mixtures below to help preserve the condition of the graphics on your aircraft.

### • Caution •

Do not use abrasive polishes or cutting compounds.

Do not use polish or wax on graphics with a matte or texture finish.

Initially test restoration products and mixtures on an inconspicuous area of the graphic to verify they will not cause damage.

### • Note •

Use an all-purpose cleaner to remove wax or wax residue.

Film or Finish Type	Product or Mixture
Smooth Gloss	3M™ Perfect-it™ Show Car Paste Wax 39526; Meguiar's Gold Class™ Carnuaba Plus Premium Liquid Wax
Matte or Satin Texture	Mixture of two parts isopropyl alcohol to one part water (mix ratio 2:1)
Matte White (1080-M10) Carbon Fiber White Texture (1080-CF10)	Depending on the type and degree of contamination to be removed, use one or more of the following solutions in the order shown: 1. Hot, soapy water solution 2. Mixture of two parts isopropyl alcohol to one part water (mix ratio 2:1) 3. Simple Green® All-Purpose Cleaner 4. Household chlorine bleach, followed by a mixture of two parts isopropyl alcohol to one part water (mix ratio 2:1) 5. Mineral spirits, followed by a mixture of two parts isopropyl alcohol to one part water (mix ratio 2:1)
Carbon Fiber or Brushed Metal Texture	3M™ Tire Restorer or Meguiar's Natural Shine Protectant
Carbon Fiber Black Texture (1080-CF12)	Meguiar's Ultimate Black Plastic Restorer

**Figure 8-3**  
**Recommended Graphic Restoration Products and Mixtures**

### ***Windscreen and Windows***

Before cleaning an acrylic window, rinse away all dirt particles before applying cloth or chamois. Never rub dry acrylic. Dull or scratched window coverings may be polished using a special acrylic polishing paste.

#### **• Caution •**

Clean acrylic windows with a solvent-free, nonabrasive, antistatic acrylic cleaner. Do not use gasoline, alcohol, benzene, carbon tetrachloride, thinner, acetone, or glass window cleaning sprays.

Use only a nonabrasive cotton cloth or genuine chamois to clean acrylic windows. Paper towel or newspaper are highly abrasive and will cause hairline scratches.

1. Remove grease or oil using a soft cloth saturated with kerosene then rinse with clean, fresh water.

#### **• Note •**

Wiping with a circular motion can cause glare rings. Use an up and down wiping motion to prevent this.

To prevent scratching from dirt that has accumulated on the cloth, fold cloth to expose a clean area after each pass.

2. Using a moist cloth or chamois, gently wipe the windows clean of all contaminants.
3. Apply acrylic cleaner to one area at a time, then wipe away with a soft, cotton cloth.
4. Dry the windows using a dry nonabrasive cotton cloth or chamois.

### ***Enhanced Vision System Sensor Windows (Optional)***

The Enhanced Vision System Sensor is located on the underside of the LH wing. The three sensor windows are made of Germanium. In contrast to visible light energy, infrared energy typically passes through dirt on the window. As such, the Sensor windows requires only occasional cleaning with mild liquid soap and water or isopropyl alcohol, and a soft cloth.

#### **• Caution •**

If a EVS Sensor Window breaks, use gloves and masks when handling broken germanium window material.

Do not use abrasive cleansers or cleaning pads on the germanium window. Abrasive cleaning can damage the sensor window coating.

Do not use any cleansers containing ammonia. Ammonia will remove the sensor window coating.

### ***Engine Compartment***

Before cleaning the engine compartment, place a strip of tape on the magneto vents to prevent any solvent from entering these units.

1. Place a large pan under the engine to catch waste.
2. Remove induction air filter and seal off induction system inlet.
3. With the engine cowling removed, spray or brush the engine with solvent or a mixture of solvent and degreaser. In order to remove especially heavy dirt and grease deposits, it may be necessary to brush areas that were sprayed.

#### **• Caution •**

Do not spray solvent into the alternator, vacuum pump, starter, or induction air intakes.

4. Allow the solvent to remain on the engine from 5 to 10 minutes. Then rinse engine clean with additional solvent and allow it to dry.

#### **• Caution •**

Do not operate the engine until excess solvent has evaporated or otherwise been removed.

5. Remove the protective tape from the magnetos.
6. Open induction system air inlet and install filter.
7. Lubricate in accordance with the Airplane Maintenance Manual (AMM), Chapter 12, Servicing.

### ***Landing Gear***

Before cleaning the landing gear, place a plastic cover or similar material over the wheel and brake assembly.

1. Place a pan under the gear to catch waste.
2. Spray or brush the gear area with solvent or a mixture of solvent and degreaser, as desired. Where heavy grease and dirt deposits have collected, it may be necessary to brush areas that were sprayed, in order to clean them.
3. Allow the solvent to remain on the gear from five to ten minutes. Then rinse the gear with additional solvent and allow to dry.
4. Remove the cover from the wheel and remove the catch pan.
5. Lubricate the gear in accordance with the Airplane Maintenance Manual (AMM), Chapter 12, Servicing.

## Cleaning Interior Surfaces

Seats, carpet, upholstery panels, and headliners should be vacuumed at regular intervals to remove surface dirt and dust. While vacuuming, use a fine bristle nylon brush to help loosen particles.

### • Caution •

Remove any sharp objects from pockets or clothing to avoid damaging interior panels or upholstery.

### ***Windshield and Windows***

Never rub dry acrylic. Dull or scratched window coverings may be polished using a special acrylic polishing paste.

### • Caution •

Clean acrylic windows with a solvent free, none abrasive, antistatic acrylic cleaner. Do not use gasoline, alcohol, benzene, carbon tetrachloride, thinner, acetone, or glass window cleaning sprays.

Use only a nonabrasive cotton cloth or genuine chamois to clean acrylic windows. Paper towel or newspaper are highly abrasive and will cause hairline scratches.

### • Note •

Wiping with a circular motion can cause glare rings. Use an up and down wiping motion to prevent this.

To prevent scratching from dirt that has accumulated on the cloth, fold cloth to expose a clean area after each pass.

1. Using a moist cloth or chamois, gently wipe the windows clean of all contaminants.
2. Apply acrylic cleaner to one area at a time, then wipe away with a soft, cotton cloth.

Dry the windows using a dry nonabrasive cotton cloth or chamois.



<b>Cleaning Product</b>	<b>Cleaning Application</b>	<b>Supplier</b>
Prist	Interior Windscreen and Windows	Prist Aerospace
Optimax	Display Screens	PhotoDon
Mild Dishwasher Soap (abrasive free)	Cabin Interior	Any Source
Leather Care Kit 50689-001	Leather Upholstery	Cirrus Design
Leather Cleaner 50684-001	Leather Upholstery	Cirrus Design
Ink Remover 50685-001	Leather Upholstery	Cirrus Design
Leather Conditioner 50686-001	Leather Upholstery	Cirrus Design
Spot and Stain Remover 50687-001	Leather Upholstery	Cirrus Design
Vinyl Finish Cleaner 50688-001	Vinyl Panels	Cirrus Design
Vinyl & Leather Cleaner 51479-001	Vinyl and Leather Upholstery	Cirrus Design

**Figure 8-4  
Recommended Interior Cleaning Products**

### ***Instrument Panel and Electronic Display Screens***

The instrument panel, control knobs, and plastic trim need only to be wiped clean with a soft damp cloth. The multifunction display, primary flight display, and other electronic display screens should be cleaned with Optimax - LCD Screen Cleaning Solution as follows:

#### **• Caution •**

To avoid solution dripping onto display and possibly migrating into component, apply the cleaning solution to cloth first, not directly to the display screen.

Use only a lens cloth or nonabrasive cotton cloth to clean display screens. Paper towels, tissue, or camera lens paper may scratch the display screen.

Clean display screen with power OFF.

1. Gently wipe the display with a clean, dry, cotton cloth.
2. Moisten clean, cotton cloth with cleaning solution.
3. Wipe the soft cotton cloth across the display in one direction, moving from the top of the display to the bottom. Do not rub harshly.
4. Gently wipe the display with a clean, dry, cotton cloth.

### ***Headliner and Trim Panels***

The airplane interior can be cleaned with a mild detergent or soap and water. Harsh abrasives or alkaline soaps or detergents should be avoided. Solvents and alcohols may damage or discolor vinyl or urethane parts. Cover areas where cleaning solution could cause damage. Use the following procedure:

#### **• Caution •**

Solvent cleaners and alcohol should not be used on interior parts. If cleaning solvents are used on cloth, cover areas where cleaning solvents could cause damage.

1. Clean headliner, and side panels, with a stiff bristle brush, and vacuum where necessary.
2. Soiled upholstery, may be cleaned with a good upholstery cleaner suitable for the material. Carefully follow the manufacturer's instructions. Avoid soaking or harsh rubbing.

### ***Leather Upholstery and Seats***

For routine maintenance, occasionally wipe leather upholstery with a soft, damp cloth. For deeper cleaning, start with mix of mild detergent and water then, if necessary, work your way up to the products available from Cirrus for more stubborn marks and stains. Do not use soaps as they contain alkaline which will alter the leather's pH balance and cause the leather to age prematurely. Cover areas where cleaning solution could cause damage. Use the following procedure:

#### **• Caution •**

Solvent cleaners and alcohol should not be used on leather upholstery.

1. Clean leather upholstery with a soft bristle brush, and vacuum where necessary.
2. Wipe leather upholstery with a soft, damp cloth.
3. Soiled upholstery, may be cleaned with the approved products available from Cirrus Design. Avoid soaking or harsh rubbing.

### ***Carpets***

To clean carpets, first remove loose dirt with a whiskbroom or vacuum. For soiled spots and stubborn stains use a non-flammable, dry cleaning fluid. Floor carpets may be cleaned like any household carpet.

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## ***Section 9: Log of Supplements***

<b>Inst</b>	<b>Part Number</b>	<b>Title</b>	<b>Rev Date</b>
___	13772-109 R2	Approved Oxygen Systems	01-06-10
___	13772-114 R2	SR22 / SR22T Airplanes Registered in Canada	09-24-13
___	13772-122 R1	SR22 / SR22T Airplanes Registered in European Union	07-07-10
___	13772-131 R2	Artex ME406 406 MHz ELT System	01-06-10
___	13772-135 R4	GFC 700 Automatic Flight Control System	09-08-14
___	13772-136 R1	Garmin Terrain Awareness/Warning System	01-06-10
___	13772-143 R2	Part 135 Electrical Loading Shedding Procedure	01-06-10
___	13772-146 R1	SR22T Airplanes Registered in South Africa	02-25-13
___	13772-147	SR22 / SR22T Airplanes Registered in Colombia	10-07-10
___	13772-148 R1	SR22 / SR22T Airplanes Registered in Chile	03-11-13
___	13772-149 R1	SR22 / SR22T Airplanes Registered in Mexico	09-24-13
___	13772-151	TKS Anti-Ice System	02-01-13
___	13772-155	SR22T Airplanes Registered in Egypt	04-07-14
___	13772-156	Artex ELT 1000 406 MHz ELT System	11-20-14

FAA Approved POH Supplements must be in the airplane for flight operations when the subject optional equipment is installed or the special operations are to be performed.

This Log of Supplements shows all Cirrus Design Supplements available for the aircraft at the corresponding date of the revision level shown in the lower left corner. A check mark in the Part Number column indicates that the supplement is applicable to the POH. Any installed supplements not applicable to the POH are provided for reference only.

If the pilot elects to touchdown with a door opened, there are several additional factors the pilot must consider: loss of door, possibility of head injury, or injury from an object coming through the open door.

- If a door is open prior to touchdown in a CAPS landing, the door will most likely break away from the airplane at impact.
- If the door is open and the airplane contacts the ground in a rolled condition, an occupant could be thrown forward and strike their head on the exposed door pillar. Contacting the ground in a rolled condition could be caused by terrain that is not level, contacting an obstacle such as a tree, or by transient aircraft attitude.
- With a door open, it is possible for an object such as a tree limb or flying debris to come through the opening and strike an occupant.

**• WARNING •**

If it is decided to unlatch a door, unlatch one door only. Opening only one door will provide for emergency egress as well as reduce risks associated with ground contact. Typically, this would be the copilot's door as this allows the other occupants to exit first after the airplane comes to rest.

***Water Landings***

The ability of the airplane to float after a water landing has not been tested and is unknown. However, since there is the possibility that one or both doors could jam and use of the emergency egress hammer to break out a window could take some time, the pilot may wish to consider unlatching a door prior to assuming the emergency landing body position in order to provide a ready escape path should the airplane begin to sink.

### ***Post Impact Fire***

If there is no fire prior to touchdown and the pilot is able to shut down the engine, fuel, and electrical systems, there is less chance of a post impact fire. If the pilot suspects a fire could result from impact, unlatching a door immediately prior to assuming the emergency landing body position should be considered to assure rapid egress.

### ***Ground Gusts***

If it is known or suspected that ground gusts are present in the landing zone, there is a possibility that the parachute could drag the airplane after touchdown, especially if the terrain is flat and without obstacles. In order to assure that the occupants can escape the airplane in the timeliest manner after the airplane comes to rest, the pilot may elect to unlatch the copilot's door for the CAPS landing. Occupants must be in the Emergency Landing Body Position for touchdown. Occupants must not loosen seat belts until the airplane comes to rest. When the airplane comes to rest, the occupants should exit the airplane and immediately move upwind to prevent a sudden gust from dragging the airplane in their direction.



## Taxiing, Steering, and Braking Practices

Cirrus aircraft use a castering nose wheel and rely on aerodynamic forces and differential braking for directional control while taxiing. Proper braking practices are therefore critical to avoid potential damage to the brakes.

The most common cause of brake damage and/or failure is the creation of excessive heat through improper braking practices. Pilots unaccustomed to free castering nose wheel steering may be inclined to “ride” the brakes to maintain constant taxi speeds and use the brakes excessively for steering.

### • Caution •

When brake temperatures are between 270-293°F (132-145°C), the Crew Alerting System will display a BRAKE TEMP Caution annunciation. A BRAKE TEMP Warning annunciation occurs when brake temperature exceeds 293°F (145°C). If either annunciation occurs, the pilot should stop the aircraft and allow the brakes to cool to avoid damaging the brake system.

## Operating Practices

When taxiing, directional control is accomplished with rudder deflection and intermittent braking (toe taps) as necessary. Use only as much power as is necessary to achieve forward movement. Deceleration or taxi speed control using brakes but without a reduction in power will result in increased brake temperature.

On flat, smooth, hard surfaces, do not exceed 1000 RPM maximum continuous engine speed for taxi. Power settings slightly above 1000 RPM are permissible to start motion, for turf, soft surfaces, and on inclines. Use minimum power to maintain constant taxi speed.

“Riding the brakes” while taxiing is similar to driving a car with one foot on the brake and one foot on the gas. This causes a continuous build up of energy that would otherwise be moving the airplane.

Observe the following operating practices:

- Verify that the parking brake is completely disengaged before taxi.
- The rudder is effective for steering on the ground and should be used.

- Use only as much power (throttle) as is necessary to achieve forward movement. Keep in mind, any additional power added with the throttle will be absorbed in the brakes to maintain constant speed.
- Use rudder deflection and the minimum necessary inputs of differential braking to achieve directional control.
- Do not “ride the brakes”. Pilots should consciously remove pressure from the brakes while taxiing. Failure to do so results in excessive heat buildup, premature brake wear, and increased possibility of brake failure or fire.
- Avoid unnecessary high-speed taxiing. High-speed taxiing may result in excessive demands on the brakes, increased brake wear, and the possibility of brake failure or fire.
- Brakes have a large energy absorbing capacity; therefore, cooling time should be considered. Energy absorbed during a few seconds of deceleration can take up to an hour to dissipate (*Serials 0442 thru 0656, 0658 thru 0689 before SB2X-32-21*), or several minutes to dissipate (*Serials 0442 thru 0656, 0658 thru 0689 after SB2X-32-21, 0657, 0690 & subs*). Always allow adequate cooling time after brake use.
- Allow a cooling period following a high-energy braking event. High-energy braking can include an aborted takeoff or the equivalent energy required for a Maximum Gross Weight full-stop from 70 knots in less than 1000 feet.

## Brake Maintenance

The brake assemblies and linings should be checked at every oil change (50 hours) for general condition, evidence of overheating, and deterioration.

The aircraft should not be operated with overheated, damaged, or leaking brakes. Conditions include, but are not limited to:

- Leaking brake fluid at the caliper. This can be observed by checking for evidence of fluid on the ground or deposited on the underside of the wheel fairing. Wipe the underside of the fairing with a clean, white cloth and inspect for red colored fluid residue.
- Overheated components, indicated by discoloration or warping of the disk rotor. Excessive heat can cause the caliper

components to discolor or cause yellowing of the part identification label.

Refer to Section 8, *Landing Gear Servicing* for specific servicing information on the Brake System.

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